

# TYPE CERTIFICATE DATA SHEET

No. EASA.IM.R.131

**for** 269

## **Type Certificate Holder**

Schweizer RSG LLC

3901 N Main St. Fort Worth, Texas 76106 USA

For Model: 269A 269B 269C, 269C-1 269D



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#### SECTION 1: Model 269A

SECTION	1: Model 269A			
I. General				
1. Тур	e/ Model/ Variant			
1.1	Туре	269		
1.2	Model	269A		
1.3	Variant			
2. Airv	vorthiness Category	Small Rotor	craft, Normal Category	
3. Mar	nufacturer	Schweizer R 3901 N Main Fort Worth, U.S.A.		
4. Тур	e Certification Application Date	to FAA: 23 J	anuary 1956	
5. Stat	e of Design Authority	Federal Avia	ation Administration (FAA), USA	
6. Тур	e Certificate Date	by FAA: by LBA:	9 April 1959 15 June 1962	
7. Тур	e Certificate n°	by FAA: by LBA:	4H12 3018/RC	
8. Тур	e Certificate Data Sheet n°	by FAA: by LBA:	4H12 3018/RC	
9. EAS	A Type Certification Date		rer 2003, ce with CR (EU) 1702/2003, Article 2, 3., (a), t, 2 <sup>nd</sup> indented bullet.	
II. Certific	II. Certification Basis			

# 1. Reference Date for determining the 1 April 1957 applicable requirements

2. Airworthiness Requirements

CAR Part 6, dated 15 January 1951, including Amdts. 6-1 through 6-7 and 6-8, except for CAR 6.604(c). In addition, compliance with CAR 6.401(b) effective 17 May 1958 and CAR 6.637 effective 1 April 1957 has been required, based on the conditions of Director, Bureau of Flight Standards letter dated 27 March 1959, granting extension of effectiveness of Application for Type Certificate until 1 July 1959.

3.	Special Conditions	none
4.	Exemptions	none
5.	Deviations	none
6.	Equivalent Safety Findings	none
7.	Requirements elected to comply	none
8.	Environmental Protection Requirements	
	8.1 Noise Requirements	See TCDSN EASA.IM.R.131
	8.2 Emission Requirements	n/a
9.	Operational Suitability Data (OSD)	Not required for rotorcraft that are no longer in production. CR (EU) 748/2012, as amended by CR (EU) 69/2014 does not require OSD elements for this model (see Article 7a, 1.).



#### III. Technical Characteristics and Operational Limitations

<u>III.</u>	echn	ical Characteristics and Operat	lonal Limitatio	<u>115</u>		
1.	Туре	e Design Definition	D	rawing 269A0045-	BCS/-1	
2.	Des	cription	aı sk	ght single reciproc rticulated main rot kid type standard la assenger (see appr	or, twin blade te anding gear, one	pilot and one
3.	Equi	ipment	Cu th Ru	ertification Basis) s ne Airworthiness C	hall be installed ertificate release	worthiness rules (see on the helicopter for e. A Helicopter" Report
4.	Dim	ensions				
	4.1	Fuselage	W	ength: /idth: eight:	6.81 m 1.30 m 2.52 m	
	4.2	Main Rotor	(it bl D	lade assembly) iameter:	7.71 m	A1145 main rotor otor blade assembly)
	4.3	Tail Rotor	D	iameter:	1.30 m	
5.	Engi	ine				
	5.1	Model	1 1	/coming Engines x Model HO-360-E x Model O-360-C2 x Model HIO 360-I	D, or,	
	5.2	Type Certificate		AA TC/TCDS n°: ASA TC/TCDS n°:	E-286, 1E10 EASA.IM.E.032	
	5.3	Limitations	In	istalled Engine Lim	itations	
		For: HO-360-B1A, HO-360-B1B	Power [hp]	Rpm [min <sup>-1</sup> ]	Man. Press. [in Hg]	Altitude [ft]
		Max Continuous	160	2 900	26.0	MSL
		Max Continuous	160	2 900	24.8	4 000
		ТКОҒ	160	2 900	25.0	to 300 above GND
		Max PWR (5 min)	180	2 900	full throttle	more than 300 above GND
		For: 0-360-C2D	Power [hp]	Rpm [min <sup>-1</sup> ]	Man. Press. [in Hg]	Altitude [ft]
		Max Continuous	160	2 700	26.0	MSL
		Max Continuous	160	2 700	24.8	4 000
		TKOF (5 min)	165	2 900	26.0	MSL
			Power	Rpm	Man. Press.	Altitude
		For: HIO-360-B1A, HIO-360-B1B	[hp]	[min <sup>-1</sup> ]	[in Hg]	[ft]
		-		-	[in Hg] 26.2	[ft] MSL
		HIO-360-B1B	[hp]	[min <sup>-1</sup> ]		



6.	Fluids (Fuel/ Oil/ Additives)		
	6.1 Fuel	MIL-G-5572, Grade gasoline	91/96 minimum grade aviation
	6.2 Oil	<ul> <li>MIL-L-6082 or SAE J</li> <li>For detailed inform No. 1014.</li> <li>Main and tail rotor f</li> <li>MIL-L-2105E or SAE</li> </ul>	
	6.3 Additives	n/a	
7.	Fluid capacities		
	7.1 Fuel	Fuel tank capacity:	94.6 litres STA 107 (25 US gal), 113.6 litres STA 107 (30 US gal) with optional tank
	7.2 Oil	Engine: Main transmission: Tail rotor transm.:	7.6 litres STA 91 (2 US gal) 2.84 litres (0.75 US gal) 0.24 litres (0.063 US gal)
	7.3 Coolant System Capacity	n/a	
8.	Air Speed Limitations		MSL ∈ with altitude see approved 269A I and related Supplements.
9.	Rotor Speed Limitations	main rotor blades:Power on:EnMaximum2 5Minimum2 7Power off:RoMaximum53Minimum40	0
		or 269A1190 main r Power on: En Maximum 2.9	9B1145, 269B1145-1, 269B1145-25, otor blades: gine [rpm] 900 500
		Power off:RoMaximum53Minimum40	
10.	Maximum Operating Altitude and Temperature		
	10.1 Altitude		DA with altitude see approved Pilot's elated Supplements.
	10.2 Temperature	none given	
11.	Operating Limitations	the airworthiness a	nstruments and equipment, required by and/or operating rules, are approved, operable condition. See approved Pilot's



### 12. Maximum Mass

s/n 0011 through 0314: 703 kg (1 550 lb)

Max. mass may be increased to 726 kg (1 600 lb) if all the following components are installed:

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Component	p/n
Blade Assembly - Main Rotor	269A1131, 269A1131-1, 269B1145, 269B1145-25, or 269B1145-1
Blade Dampers - Main Rotor	269A1222, 269A1927 or 269A1927-3
Engine	HO-360-B1A, HO-360-B1B, HIO-360-B1A, or HIO-360-B1B
Landing Gear Assembly	269A3240

s/n 0315 and up: 757 kg (1 670 lb), see Note 2

13.	Centre of Gravity Range	Longitudinal: STA 95 to 100. For limits with accessories installed, see approved Pilot's Flight Manual. Lateral: See Loading Instructions in approved Pilot's Flight Manual.		
14.	Datum	Longitudinal: the datum line (STA 0) is located at 2 540 mm (100.0 in) forward of the main rotor hub centreline. Lateral: the datum line (B.L. 0) is at helicopter centreline.		
15.	Levelling Means	Top of main rotor hub		
16.	Minimum Flight Crew	1 pilot, operating from the left seat at STA 84.9		
17.	Maximum Passenger Seating Capacity	1, at STA 84.9		
18.	Passenger Emergency Exit	2, one on each side of the cockpit		
19.	Maximum Baggage/ Cargo Loads	See Loading Instructions and Limitations in approved Pilot's Flight Manual.		
20.	Rotor Blade Control Movement			
	Main rotor (relative to rigging position):			
	Collective pitch (up and down):	12°±1°		
	Cyclic pitch (longitudinal):	Forward 7.5° to 9.4° Aft 6.0° to 7.5°		
	Cyclic pitch (lateral):	Left 6.5° to 7.5° Right 5.3° to 6.3°		
	With tail rotor assembly 269A6004 or 269A6003 installed (relative to rigging position):			
	Collective pitch:	Full-left pedal (thrust to right) +19.0° to +21.0° Full-right pedal (thrust to left) -9.0° to -11.0°		
	With tail rotor assembly 269A6034 or 269ASK16 installed (relative to rigging position):			
	Collective pitch:	Full-left pedal (thrust to right) +24.0° to +26.0° Full-right pedal (thrust to left) -11.0° to -13.0°		
	For rigging information of main rotor and tail A-1, B & C Helicopters Handbook of Maintena	rotor refer to latest issue of Sikorsky Models 269A, TH-55A, ince Instructions.		
21.	Auxiliary Power Unit (APU)	n/a		





Refer to Publication No. CSP-C-2 Sikorsky Models 269A,

22. Life-limited Parts

Maintenance Instructions, Appendix B, Periodic Inspection Overhaul and Retirement Schedule, and Weight and Balance Procedures.

#### IV. Operating and Service Instructions

1.	Flight Manual	Refer to Publication No. CSP-AA-1 approved Rotorcraft Flight Manual Schweizer Model 269A Helicopter
2.	Maintenance Manual	Refer to Publication No. CSP-C-2 Sikorsky Models 269A, TH-55A, A-1, B & C Helicopters Basic Handbook of Maintenance Instructions.
3.	Structural Repair Manual	n/a
4.	Weight and Balance Manual	Refer to Publication No. CSP-AA-1 approved Rotorcraft Flight Manual Schweizer Model 269A Helicopter Section IV
5.	Illustrated Parts Catalogue	Refer to Publication No. CSP-C-7 Model 269A, 200 Model 269A-1, 300 Model 269B, 300C Model 269C, U.S. Army Model TH-55A Illustrated Parts Catalog
6.	Service Letters and Service Bulletins	As published by Schweizer RSG.
		For information published by previous Type Certificate holders see Note 4 in 'Section: Notes (data pertinent to all Models [])'.
7.	Required Equipment	Refer to "Equipment List Model 269A Helicopter" Report No. JW-00-1.

#### V. Notes (Model 269A only)

- Manufacturer's eligible serial numbers: s/n --0001 through --0008, --0011 and subsequent. s/n --0650 through --1109 were manufactured under the Delegation Option provisions of FAR 21.
- 2. Current weight and balance report, including list of equipment including certificated empty weight and loading instructions, must be provided for each helicopter at the time of original airworthiness certification and at all times thereafter (except in the case of operators having an appropriate weight control system). Ballast, when necessary, must be carried in accordance with the loading instructions in the approved Pilot's Flight Manual.



SEC	TION 2: Model 269B		
<u>I. G</u>	eneral		
1.	Type/ Model/ Variant		
	1.1 Type	269	
	1.2 Model	269B	
	1.3 Variant		
2.	Airworthiness Category	Small Rotorc	craft, Normal Category
3.	Manufacturer	Schweizer RS 3901 N Main Fort Worth, U.S.A.	n St.
4.	Type Certification Application Date	to FAA: 28 A	ugust 1963
5.	State of Design Authority	Federal Avia	tion Administration (FAA), USA
6.	Type Certificate Date	by FAA: by LBA:	30 December 1963 1 April 1965
7.	Type Certificate n°	by FAA: by LBA:	4H12 3018/RC
8.	Type Certificate Data Sheet n°	by FAA: by LBA:	4H12 3018/RC
9.	EASA Type Certification Date		er 2003, ce with CR (EU) 1702/2003, Article 2, 3., (a), c, 2 <sup>nd</sup> indented bullet.
<u>II. C</u>	Certification Basis		
1.	Reference Date for determining the applicable requirements	1 April 1957	
2.	Airworthiness Requirements		
	CAR Part 6, dated 15 January 1951, including a addition, compliance with CAR 6.401(b) effect 27.1323 of Amendment 27-2 effective 25 Feb	tive 17 May 19	958, CAR 6.637 effective 1 April 1957 and FAR
3.	Special Conditions	none	
4.	Exemptions	none	
5.	Deviations	none	
6.	Equivalent Safety Findings	none	
7.	Requirements elected to comply	none	
8.	Environmental Protection Requirements		
	8.1 Noise Requirements	See TCDSN E	ASA.IM.R.131
	8.2 Emission Requirements	n/a	
9.	Operational Suitability Data (OSD)	production.	d for rotorcraft that are no longer in CR (EU) 748/2012, as amended by CR (EU) es not require OSD elements for this model 7a, 1.).

#### III. Technical Characteristics and Operational Limitations

1.	Type Design Definition	Drawing 269A0046-BCS/-1
2.	Description	Light single reciprocating engine rotorcraft, three blades



TCDS No.: EASA.IM.R.131 Issue: 2	269 Date: 4 July 2019
	articulated main rotor, twin blade teetering tail rotor, skid type standard landing gear, one pilot and two passengers (see approved Pilot's Flight Manual).
3. Equipment	Basic equipment required by the airworthiness rules (see Certification Basis) shall be installed on the helicopter for the Airworthiness Certificate release. Refer to "Equipment List for FAA Certification 269B Helicopter" Report No. 269B-X-8001.
4. Dimensions	
4.1 Fuselage	Length:         6.81 m           Width:         1.30 m           Height:         2.52 m
4.2 Main Rotor	Diameter: 7.62 m (if equipped with p/n 1125, 1131, or A1145 main rotor blade assembly) Diameter: 7.71 m (if equipped with p/n B1145 main rotor blade assembly)
4.3 Tail Rotor	Diameter: 1.30 m
5. Engine	
5.1 Model	Lycoming Engines 1 x Model HIO-360-A1A
5.2 Type Certificate	FAA TC/TCDS n°: 1E10

5.3 Limitations

Installed Engine Limitations

EASA.IM.E.032

EASA TC/TCDS n°:

	Power [hp]	Rpm [min <sup>-1</sup> ]	Man. Press. [in Hg]	Altitude [ft]
Max Continuous	160	2 900	23.5	MSL
Max Continuous	160	2 900	22.0	7 200
TKOF	180	2 900	26.1	MSL
Max PWR (5 min)	180	2 900	25.0	3 900

- 6. Fluids (Fuel/ Oil/ Additives)
- 6.1 Fuel Grade 100/130 (green) 6.2 Oil Engine: MIL-L-22851 or SAE J1899 (ashless dispersant type)\* MIL-L-6082 or SAE J1966 (straight mineral type)\* \* For detailed information see Lycoming Service Instruction No. 1014. Main and tail rotor transmission: MIL-L-2105E or SAE J2360\*\* \*\* For detailed information see S-300C Basic HMI. 6.3 Additives n/a 7. Fluid capacities 7.1 Fuel Fuel tank capacity: 94.6 litres STA 107 (25 US gal) 113.6 litres STA 107 (30 US gal) with optional tank 7.2 Oil Engine: 7.6 litres STA 91 (2 US gal) Main transmission: 2.84 litres (0.75 US gal) Tail rotor transm.: 0.24 litres (0.063 US gal) 7.3 Coolant System Capacity n/a



Date: 4 July 2019

8.	Air Speed Limitations	V <sub>NE</sub> : 87 KIAS	at MSL	
		For reduction on $V_{NE}$ with altitude see approved Pilot's Flight Manual and related Supplements.		
9.	Rotor Speed Limitations	Power on: Maximum Minimum	Engine [rpm] 2 900 2 700	
		Power off: Maximum Minimum	Rotor [rpm] 530 400	
10.	Maximum Operating Altitude and Temperature			
	10.1 Altitude	10 000 ft (3 048 m) DA For reduction of V <sub>NE</sub> with altitude see approved Pilot's Flight Manual and related Supplements.		
	10.2 Temperature	none given		
11.	Operating Limitations	<ul> <li>VFR day and night*</li> <li>Non-icing conditions</li> <li>* With appropriate instruments and equipment, required by the airworthiness and/or operating rules, are approved, installed and are in operable condition. See approved Pilot's Flight Manual for further limitations.</li> </ul>		
12.	Maximum Mass	757 kg (1 670 lb) Normal Category, see SECTION NOTES, Note 1		
13.	Centre of Gravity Range	Longitudinal: STA 95 to 101. For limits with accessories installed, see approved Pilot's Flight Manual. Lateral: See Loading Instructions in approved Pilot's Flight Manual.		
14.	Datum	Longitudinal: The datum line (STA 0) is located at 2 540 mm (100.0 in) forward of the main rotor hub centreline. Lateral: The datum line (B.L. 0) is at helicopter centreline.		
15.	Levelling Means	Top of main rotor hub		
16.	Minimum Flight Crew	1 pilot, operating	g from the left seat at STA 84.9	
17.	Maximum Passenger Seating Capacity	2, 1 at STA 78.5	and 1 at STA 84.9	
18.	Passenger Emergency Exit	2, one on each s	ide of the cockpit	
19.	Maximum Baggage/ Cargo Loads	See Loading Inst Pilot's Flight Ma	ructions and Limitations in approved nual.	
20.	Rotor Blade Control Movement			
	Main rotor (relative to rigging position):			
	Collective pitch (up and down):	12°±1°		
	Cyclic pitch (longitudinal):	Forward 7.5° to Aft 6.0° to 7.5°	9.4°	
	Cyclic pitch (lateral):	Left 6.5° to 7.5° Right 5.3° to 6.3		
	With tail rotor assembly 269A6004 or 269A600	03 installed (relati	ive to rigging position):	
	Collective pitch:	Full-left pedal (t +19.0° to +21.0° Full-right pedal -9.0° to -11.0°		



With tail rotor assembly 269A6034 or 269ASK16 installed (relative to rigging position):

Collective pitch:	Full-left pedal (thrust to right)
	+24.0° to +26.0°
	Full-right pedal (thrust to left)
	-11.0° to -13.0°

For rigging information of main rotor and tail rotor refer to latest issue of Sikorsky Models 269A, TH-55A, A-1, B & C Helicopters Handbook of Maintenance Instructions.

21.	Auxiliary Power Unit (APU)	n/a
22.	Life-limited Parts	Refer to latest issue of Sikorsky Models 269A, TH-55A, A-1, B & C Helicopters Handbook of Maintenance Instructions, Appendix B – Periodic Inspection Overhaul and Retirement Schedule, and Weight and Balance Procedures.
<u>IV. (</u>	Operating and Service Instructions	
1.	Flight Manual	Refer to latest issue of approved Pilot's Flight Manual.
2.	Maintenance Manual	Refer to latest issue of Sikorsky Models 269A, TH-55A, A- 1, B & C Helicopters Handbook of Maintenance Instructions.
3.	Structural Repair Manual	n/a
4.	Weight and Balance Manual	Refer to Publication No. CSP-BA-1 approved Rotorcraft Flight Manual Schweizer Model 269B Helicopter Section IV.
5.	Illustrated Parts Catalogue	Refer to Publication No. CSP-C-7 Model 269A, 200 Model 269A-1, 300 Model 269B, 300C Model 269C, U.S. Army Model TH-55A Illustrated Parts Catalog
6.	Service Letters and Service Bulletins	As published by Schweizer RSG. For information published by previous Type Certificate holders see Note 4 in 'Section: Notes (data pertinent to all Models [])'.
7.	Required Equipment	Refer to "Equipment List for FAA Certification 269B Helicopter", Report No. 269B-X-8001.

#### V. Notes (Model 269B only)

 Manufacturer's eligible serial numbers: s/n --0001 and up. s/n --0236 through --0475 were manufactured under the Delegation Option provisions of FAR 21.



I

#### SECTION 3: Model 269C

9	SECTION 3: Model 269C	
<u> </u>	. General	
-	1. Type/ Model/ Variant	
	1.1 Type	269
	1.2 Model	269C
	1.3 Variant	
2	2. Airworthiness Category	Small Rotorcraft, Normal Category
3	3. Manufacturer	Schweizer RSG LLC 3901 N Main St. Fort Worth, Texas 76106 U.S.A.
4	4. Type Certification Application Date	to FAA: 13 August 1968
ļ	5. State of Design Authority	Federal Aviation Administration (FAA), USA
(	5. Type Certificate Date	by FAA: 15 May 1970 by LBA: 3 September 1970
		by DGAC FR: 3 November 1988
-	7. Type Certificate n°	by FAA: 4H12 by LBA: 3018/RC
		by DGAC FR: IM 90
8	<ol><li>Type Certificate Data Sheet n°</li></ol>	by FAA: 4H12 by LBA: 3018/RC
		by DGAC FR: IM 90
ġ	9. EASA Type Certification Date	28 September 2003, in accordance with CR (EU) 1702/2003, Article 2, 3., (a), (i), 2 <sup>nd</sup> bullet, 2 <sup>nd</sup> indented bullet.
I	I. Certification Basis	
	<ol> <li>Reference Date for determining the applicable requirements</li> </ol>	25 February 1968
2	2. Airworthiness Requirements	
	addition, compliance with CAR 6.401(b) effec 27.1323 of Amdt. 27-2 effective 25 February	Amdts. 6-1 through 6-7 and 6-8, except CAR 6.604(c). In tive 17 May 1958, CAR 6.637 effective 1 April 1957 and FAR 1968 in lieu of CAR 6.612(a) has been required. ion Option Authorization Provisions of FAR 21.
	3. Special Conditions	none
4	4. Exemptions	none
ļ	5. Deviations	none
(	6. Equivalent Safety Findings	none
7	7. Requirements elected to comply	none
8	8. Environmental Protection Requirements	
	8.1 Noise Requirements	See TCDSN EASA.IM.R.131 (see also Note 2)
	8.2 Emission Requirements	n/a

9. Operational Suitability Data (OSD) See Section 7 below



#### III. Technical Characteristics and Operational Limitations

<u>III.</u> <sup>-</sup>	Technical Characteristics and Oper	ational Limitations	<u>5</u>		
1.	Type Design Definition	Dra	wing 269A0050-	BCS/-003	
2.	Description	arti skic	Light single reciprocating engine rotorcraft, three blades articulated main rotor, twin bladed teetering tail rotor, skid type standard landing gear, one pilot and two passengers (see approved Pilot's Flight Manual).		
3.	Equipment	Cer the "Eq	tification Basis) s Airworthiness Co uipment List Mo		ter S/N 1796 –
4.	Dimensions				
	4.1 Fuselage	Wic	gth: Jth: ght:	6.81 m 1.30 m 2.52 m	
	4.2 Main Rotor	(if e blao Dia	de assembly) meter:	7.71 m	A1145 main rotor tor blade assembly)
	4.3 Tail Rotor	Dia	meter:	1.30 m	
5.	Engine				
	5.1 Model		oming Engines Model HIO-360-I	D1A	
	5.2 Type Certificate		A TC/TCDS n°: SA TC/TCDS n°:	1E10 EASA.IM.E.032	
	5.3 Limitations	Inst	alled Engine Lim	itations	
		Power [kW (hp)]	Rpm [min⁻¹]	Man. Press. [in Hg]	Altitude [ft]
	Max Continuous	141.7 (190)	3 200	26.0	MSL
	Max Continuous	141.7 (190)	3 200	24.7	4 200
6.	Fluids (Fuel/ Oil/ Additives)				
	6.1 Fuel	AST	M D910A, Grade	e 100/130 (green)	)
	6.2 Oil	MIL MIL * F Ma MIL	-L-6082 or SAE J For detailed inform No. 1014. in and tail rotor t -L-2105E or SAE	ransmission:	neral type)* g Service Instruction
	6.3 Additives	n/a			
7.	Fluid capacities				
	7.1 Fuel	Fue	l tank capacity:		A 107 (30 US gal) A 107 (49 US gal) tank
		Usa	ble fuel:	112.8 litres (29	
	7.2 Oil	-	ine: in transmission:	7.6 litres STA 9 2.84 litres (0.7	· - ·



		Tail rotor transm	.: 0.24 litres (0.06	3 US gal)
	7.3 Coolant System Capacity	n/a		
8.	Air Speed Limitations	VNE:95 KIAS at MSLVDoors 'OFF':89 KIAS at MSLFor reduction on VNE with altitude see approved PilFlight Manual and related Supplements.		
9.	Rotor Speed Limitations	Power on: Maximum Minimum	Engine [rpm] 3 200 3 000	
		Power off: Maximum Minimum	Rotor [rpm] 504 390	
10.	Maximum Operating Altitude and Temperature			
	10.1 Altitude	12 000 ft (3 657	up to 771 kg (1 700 lb)	
	10.2 Temperature	none given		
11.	Operating Limitations	<ul> <li>VFR day and night*</li> <li>Non-icing conditions</li> <li>* With appropriate instruments and equipment, required by the airworthiness and/or operating rules, are approved, installed and are in operable condition. See approved Pilot Flight Manual for further limitations.</li> </ul>		
12.	Maximum Mass	930 kg (2 050 lb) Normal Category, see SECTION NOTES, Note 1		
		Maximum mass	may be increased to:	
		975 kg (2 150 lb) for take-off, with agricultural kit (p/n 269A4153-1001) installed, in accordance with specific limitations shown on Supplement C of appr Pilot's Flight Manual.		
		installed, in acco	with Cargo Hook kit ( rdance with specific lin approved Pilot's Fligh	mitations shown in
13.	Centre of Gravity Range		Longitudinal	
		Forward STA [in (mm)]		STA mm)]
		95.0 (2 413)	101.0	(2 565)
			Lateral	
		STA [in (mm)]	LH [in (mm)]	RH [in (mm)]
		95.0 (2 413)	-1.0 (-25)	+3 (+76)
		99.5 (2 527)	-2.12 (-54)	+4.0 (+102)
		101.0 (2 565)	-2.5 (-63)	+2.0 (+51)
		Note: Looking forv	vard, "+" indicates right o	of helicopter

<u>Note:</u> Looking forward, "+" indicates right of helicopter centreline, and "- " indicates left of helicopter centreline. For limits with accessories installed, see approved Pilot's Flight Manual.

Longitudinal: The datum line (STA 0) is located at 2 540 mm (100.0 in) forward of the main rotor hub centreline.

14. Datum



		Lateral: The datum line (B.L. 0) is at helicopter centreline.
15.	Levelling Means	Top of main rotor hub
16.	Minimum Flight Crew	1 pilot, operating from the left seat at STA 83.2
17.	Maximum Passenger Seating Capacity	2, 1 at STA 80.0 and 1 at STA 83.2
18.	Passenger Emergency Exit	2, one on each side of the cockpit
19.	Maximum Baggage/ Cargo Loads	See Loading Instructions and Limitations in approved Pilot's Flight Manual.
20.	Rotor Blade Control Movement	
	Main rotor (relative to rigging position):	
	Collective pitch (up and down):	12°±1°
	Cyclic pitch (longitudinal):	Forward 8.5° to 9.75° Aft 6.5° to 7.5°
	Cyclic pitch (lateral):	Left 6.5° to 7.5° Right 4.5° to 6.5°
	Tail rotor (relative to rigging position):	
	Collective pitch:	Full-left pedal (thrust to right) +25.0° to +27.0° Full-right pedal (thrust to left) -11.0° to -13.0°

For rigging information of main rotor and tail rotor refer to latest issue of Sikorsky Models 269A, TH-55A, A-1, B & C Helicopters Handbook of Maintenance Instructions, or,

to latest issue of Sikorsky S-300C Model 269C Helicopter Basic Handbook of Maintenance Instructions (Effective S/N S1809 and Subsequent) as applicable.

21.	Auxiliary Power Unit (APU)	n/a
22.	Life-limited Parts	Refer to latest issue of Sikorsky Models 269A, TH-55A, A- 1, B & C Helicopters Handbook of Maintenance Instructions, Appendix B – Periodic Inspection Overhaul and Retirement Schedule, and Weight and Balance Procedures, or, to latest issue of Sikorsky S-300C Model 269C Helicopter Basic Handbook of Maintenance Instructions (Effective S/N S1809 and Subsequent), Appendix B - Periodic Inspection Overhaul and Retirement Schedule, and Weight and Balance Procedures, as applicable.
<u>IV. (</u>	Operating and Service Instructions	

n/a

Flight Manual 1.

2. Maintenance Manual Refer to latest issue of S-300C Pilot's Flight Manual.

Refer to latest issue of Sikorsky Models 269A, TH-55A, A-1, B & C Helicopters Handbook of Maintenance Instructions, or,

to latest issue of Sikorsky S-300C Model 269C Helicopter Basic Handbook of Maintenance Instructions (Effective S/N S1809 and Subsequent), as applicable.

- Structural Repair Manual
- 4. Weight and Balance Manual
- 5. Illustrated Parts Catalogue

Refer to Publication No. CSP-C-1 Pilot's Flight Manual containing the approved Rotorcraft Flight Manual for Sikorsky S-300C Helicopter Model 269C Section VI.22.

Refer to Publication No. CSP-C-9 Schweizer Model 269C Helicopter Illustrated Parts Catalog (IPC) Serial Numbers 1166 and subsequent



3.

6.	Service Letters and Service Bulletins	As published by Schweizer RSG. For information published by previous Type Certificate holders see Note 4 in 'Section: Notes (data pertinent to all Models [])'.
7.	Required Equipment	Refer to "Equipment List Model 269C Helicopter S/N 1796 – Subsequent", Report No. SA-269C-22-4.

V. Notes (Model 269C only)

 Manufacturer's eligible serial numbers: s/n --0004 and subsequent, except --1246, --1643 and --1660. s/n --0004 through --0082 were manufactured under the Delegation Option provisions of FAR 21.

 Noise Substantiation: Although not part of the Certification Basis, the Model 269C Helicopter is compliant with the requirements of FAR Part 36 Appendix J, Amendment 20.



# SECTION A. Model 260C 1

SEC	TION 4: Model 269C-1			
<u>I. G</u>	eneral			
1.	Type/ Model/ Variant			
	1.1 Туре	269		
	1.2 Model	269C-1		
	1.3 Variant			
2.	Airworthiness Category	Small Rotor	craft, Normal Category	
3.	Manufacturer	Schweizer R 3901 N Maiı Fort Worth, U.S.A.		
4.	Type Certification Application Date	to FAA: 9 Fe	bruary 1995	
5.	State of Design Authority	Federal Avia	ation Administration (FAA), USA	
6.	Type Certificate Date	by FAA: by LBA:	31 July 1995 25 March 1996	
7.	Type Certificate n°	by FAA: by LBA:	4H12 3018/RC	
8.	Type Certificate Data Sheet n°	by FAA: by LBA:	4H12 3018/RC	
9.	EASA Type Certification Date		er 2003, ce with CR (EU) 1702/2003, Article 2, 3., (a), t, 2 <sup>nd</sup> indented bullet.	
<u>II. C</u>	Certification Basis			
1.	Reference Date for determining the applicable requirements	25 February	1968	
2.	Airworthiness Requirements			
	CAR Part 6, dated 15 January 1951, including Amdts. 6-1 through 6-7 and 6-8, except CAR 6.604(c). In addition, compliance with CAR 6.401(b) effective 17 May 1958, CAR 6.637 effective 1 April 1957 and FAR 27.1323 of Amdt. 27-2 effective 25 February 1968 in lieu of CAR 6.612(a).			
3.	Special Conditions	none		
4.	Exemptions	none		
5.	Deviations	none		
6.	Equivalent Safety Findings	none		
_				

**Operational Suitability Data (OSD)** See SECTION 7 below **III.** Technical Characteristics and Operational Limitations

Requirements elected to comply

8.1 Noise Requirements 8.2 Emission Requirements

**Environmental Protection Requirements** 

1. Type Design Definition Drawing 269A0051--001/-003/-005/-007. 2. Description Light single reciprocating engine rotorcraft, three blades articulated main rotor, twin bladed teetering tail rotor, skid type standard landing gear, one pilot and two passengers (see approved Pilot's Flight Manual).

none

n/a

See TCDSN EASA.IM.R.131



7.

8.

9.

3.	Equipment	Basic equipment required by the airworthiness rules (see Certification Basis) shall be installed on the helicopter for the Airworthiness Certificate release. Refer to "Equipment List/ Weight and Balance Model 269C-1",			
		Report No. SA-269C-22-5.			
4.	Dimensions 4.1 Fuselage	Length: Width: Height:	6.81 m 1.30 m 2.52 m		
	4.2 Main Rotor	Diameter: (if equipped with blade assembly) Diameter: (if equipped with	7.620 m p/n 1125, 11 7.708 m		
	4.3 Tail Rotor	Diameter:	1.295 m		
5.	Engine				
	5.1 Model	Lycoming Engines 1 x Model HIO-360-D1A, or, 1 x Model HIO-360-G1A			
	5.2 Type Certificate	FAA TC/TCDS n°: EASA TC/TCDS n°	1E10 : EASA.IM.I	E.032	
	5.3 Limitations	Installed Engine L	imitations		
		Power [kW (hp)]	Rpm [min <sup>-1</sup> ]	Man. Press. [in Hg]	Altitude [ft]
	Max Continuous	134.2 (180)	2 700	full throttle	MSL
6.	Fluids (Fuel/ Oil/ Additives)				
	6.1 Fuel	ASTM D910A, Gra (purple) MIL-F-55			
	6.2 Oil	<ul> <li>(purple) MIL-F-5572, or Grade100LL (blue) ASTM-D910</li> <li>Engine:</li> <li>MIL-L-22851 or SAE J1899 (ashless dispersant type)*</li> <li>MIL-L-6082 or SAE J1966 (straight mineral type)*</li> <li>* For detailed information see Lycoming Service Instruction No. 1014.</li> <li>Main and tail rotor transmission:</li> <li>MIL-L-2105E or SAE J2360**</li> <li>** For detailed information see S-300C Basic HMI.</li> </ul>			e)*
	6.3 Additives	n/a			
7.	Fluid capacities				
	7.1 Fuel	For s/n 0001 thro Standard at STA 1 Fuel tank capacit Usable fuel: Standard + Auxili Fuel tank capacit Usable fuel: For s/n 0106 and Standard at STA 1 Fuel tank capacit Usable fuel: Standard + Auxili Fuel tank capacit	108.5: y: 133.2 lit 132.5 lit ary (optional) y: 246.8 lit 238.5 lit subsequent 108.5: y: 124.9 lit 123.0 lit ary (optional)	tres (65.2 US gal tres (63.0 US gal tres (33.0 US gal tres (32.5 US gal	) ) )



		Usable fuel:		242.2 litr	es (64.0 US gal)
	7.2 Oil	Engine:			STA 91 (2 US gal)
		Main transmissio	n: :	2.84 litre	es (0.75 US gal) es (0.063 US gal)
	7.3 Coolant System Capacity	n/a		0.24 1110	5 (0.005 05 gal)
8.	Air Speed Limitations	•	<ias at<="" td=""><td>t MSL</td><td></td></ias>	t MSL	
		VDoors 'OFF': 90 k	<ias at<="" td=""><td>t MSL</td><td></td></ias>	t MSL	
		For reduction on Flight Manual and			de see approved Pilot's lements
9.	Rotor Speed Limitations	Power on:		e [rpm]	iements.
0.		Maximum	2 700		
		Minimum	2 534		
		Power off: Maximum	Rotor 504	[rpm]	
		Minimum	390		
10.	Maximum Operating Altitude and Temperature				
	10.1 Altitude	Enroute:		000 ft	(3 048 m) DA
		Take-off/Landing	: 8	000 ft	(2 438 m) DA
11	10.2 Temperature	none given	<b>*</b> *		
11.	Operating Limitations	VFR day and nigh Non-icing conditi			
		* With appropriate instruments and equipment, required by the airworthiness and/or operating rules, are approved,			
					ndition. See approved Pilot's
		Flight Manual fo	or furth	ner limitat	ions.
12.	Maximum Mass	794 kg (1 750 lb) see SECTION NO			pry,
	Maximum Mass Centre of Gravity Range		TES, N		
		see SECTION NOT	TES, N	ote 1	inal Aft STA
		see SECTION NOT Forward STA [in (mm)]	TES, N	ote 1	inal Aft STA [in (mm)]
		see SECTION NOT	TES, N	ote 1 Longitud	inal Aft STA [in (mm)] 101.0 (2 565)
		see SECTION NOT Forward STA [in (mm)] 95.0 (2 413)	TES, N	ote 1 Longitud Latera	inal Aft STA [in (mm)] 101.0 (2 565)
		see SECTION NOT Forward STA [in (mm)] 95.0 (2 413)	ading	ote 1 Longitud Latera	inal Aft STA [in (mm)] 101.0 (2 565) al ons in approved
		see SECTION NOT Forward STA [in (mm)] 95.0 (2 413) See Lo Longitudinal: The	ading Pilot	ote 1 Longitud Latera instructio 's Flight I m line (S	inal Aft STA [in (mm)] 101.0 (2 565) al ons in approved Manual. TA 0) is located at
13.	Centre of Gravity Range	see SECTION NOT Forward STA [in (mm)] 95.0 (2 413) See Lo Longitudinal: The	ading Pilot	ote 1 Longitud Latera instructio 's Flight I m line (S	inal Aft STA [in (mm)] 101.0 (2 565) al ons in approved Manual.
13.	Centre of Gravity Range	see SECTION NOT Forward STA [in (mm)] 95.0 (2 413) See Lo Longitudinal: The 2 540 mm (100.0 centreline.	ading Pilot e datur	ote 1 Longitud Latera instructio 's Flight I m line (S' rward of	inal Aft STA [in (mm)] 101.0 (2 565) al ons in approved Manual. TA 0) is located at
13.	Centre of Gravity Range	see SECTION NOT Forward STA [in (mm)] 95.0 (2 413) See Lo Longitudinal: The 2 540 mm (100.0 centreline.	ading Pilot datur in) foi	ote 1 Longitud Latera instructio 's Flight I m line (S' rward of	inal Aft STA [in (mm)] 101.0 (2 565) al ons in approved Manual. TA 0) is located at the main rotor hub
13.	Centre of Gravity Range Datum	see SECTION NOT Forward STA [in (mm)] 95.0 (2 413) See Lo Longitudinal: The 2 540 mm (100.0 centreline. Lateral: The datu	ading Pilot datur in) for m line r hub	ote 1 Longitud Latera instructi 's Flight I m line (S <sup>-</sup> rward of e (B.L. 0) i	inal Aft STA [in (mm)] 101.0 (2 565) al ons in approved Manual. TA 0) is located at the main rotor hub is at helicopter centreline.
13. 14. 15.	Centre of Gravity Range Datum Levelling Means	see SECTION NOT Forward STA [in (mm)] 95.0 (2 413) See Lo Longitudinal: The 2 540 mm (100.0 centreline. Lateral: The datu Top of main roto	ading Pilot e datur in) for m line r hub g from	ote 1 Longitud Latera instruction 's Flight I m line (S <sup>-</sup> rward of e (B.L. 0) i the left s	inal Aft STA [in (mm)] 101.0 (2 565) al ons in approved Manual. TA 0) is located at the main rotor hub is at helicopter centreline.
<ol> <li>13.</li> <li>14.</li> <li>15.</li> <li>16.</li> <li>17.</li> <li>18.</li> </ol>	Centre of Gravity Range Datum Levelling Means Minimum Flight Crew Maximum Passenger Seating Capacity Passenger Emergency Exit	see SECTION NOT Forward STA [in (mm)] 95.0 (2 413) See Lo Longitudinal: The 2 540 mm (100.0 centreline. Lateral: The datu Top of main roto 1 pilot, operating 2, 1 at STA 80.0 a 2, one on each sid	ading Pilot e datur in) for m line r hub g from and 1 a de of t	ote 1 Longitud Latera instruction 's Flight I m line (S' rward of e (B.L. 0) i the left s at STA 83 the cockp	inal Aft STA [in (mm)] 101.0 (2 565) al ons in approved Manual. TA 0) is located at the main rotor hub is at helicopter centreline. Seat at STA 83.2 .2 bit
<ol> <li>13.</li> <li>14.</li> <li>15.</li> <li>16.</li> <li>17.</li> </ol>	Centre of Gravity Range Datum Levelling Means Minimum Flight Crew Maximum Passenger Seating Capacity	see SECTION NOT Forward STA [in (mm)] 95.0 (2 413) See Lo Longitudinal: The 2 540 mm (100.0 centreline. Lateral: The datu Top of main roto 1 pilot, operating 2, 1 at STA 80.0 a 2, one on each sid	ading Pilot e datur in) for m line g from and 1 a de of t ructior	ote 1 Longitud Latera instruction 's Flight I m line (S' rward of e (B.L. 0) i the left s at STA 83 the cockp	inal Aft STA [in (mm)] 101.0 (2 565) al ons in approved Manual. TA 0) is located at the main rotor hub is at helicopter centreline.
<ol> <li>13.</li> <li>14.</li> <li>15.</li> <li>16.</li> <li>17.</li> <li>18.</li> </ol>	Centre of Gravity Range Datum Levelling Means Minimum Flight Crew Maximum Passenger Seating Capacity Passenger Emergency Exit	see SECTION NOT Forward STA [in (mm)] 95.0 (2 413) See Lo Longitudinal: The 2 540 mm (100.0 centreline. Lateral: The datu Top of main roto 1 pilot, operating 2, 1 at STA 80.0 a 2, one on each sin See Loading Instr	ading Pilot e datur in) for m line g from and 1 a de of t ructior	ote 1 Longitud Latera instruction 's Flight I m line (S' rward of e (B.L. 0) i the left s at STA 83 the cockp	inal Aft STA [in (mm)] 101.0 (2 565) al ons in approved Manual. TA 0) is located at the main rotor hub is at helicopter centreline. Seat at STA 83.2 .2 bit
<ol> <li>13.</li> <li>14.</li> <li>15.</li> <li>16.</li> <li>17.</li> <li>18.</li> <li>19.</li> </ol>	Centre of Gravity Range Datum Levelling Means Minimum Flight Crew Maximum Passenger Seating Capacity Passenger Emergency Exit Maximum Baggage/ Cargo Loads	see SECTION NOT Forward STA [in (mm)] 95.0 (2 413) See Lo Longitudinal: The 2 540 mm (100.0 centreline. Lateral: The datu Top of main roto 1 pilot, operating 2, 1 at STA 80.0 a 2, one on each sin See Loading Instr	ading Pilot e datur in) for m line g from and 1 a de of t ructior	ote 1 Longitud Latera instruction 's Flight I m line (S' rward of e (B.L. 0) i the left s at STA 83 the cockp	inal Aft STA [in (mm)] 101.0 (2 565) al ons in approved Manual. TA 0) is located at the main rotor hub is at helicopter centreline. Seat at STA 83.2 .2 bit
<ol> <li>13.</li> <li>14.</li> <li>15.</li> <li>16.</li> <li>17.</li> <li>18.</li> <li>19.</li> </ol>	Centre of Gravity Range Datum Levelling Means Minimum Flight Crew Maximum Passenger Seating Capacity Passenger Emergency Exit Maximum Baggage/ Cargo Loads Rotor Blade Control Movement	see SECTION NOT Forward STA [in (mm)] 95.0 (2 413) See Lo Longitudinal: The 2 540 mm (100.0 centreline. Lateral: The datu Top of main roto 1 pilot, operating 2, 1 at STA 80.0 a 2, one on each sin See Loading Instr	ading Pilot e datur in) for m line g from and 1 a de of t ructior	ote 1 Longitud Latera instruction 's Flight I m line (S' rward of e (B.L. 0) i the left s at STA 83 the cockp	inal Aft STA [in (mm)] 101.0 (2 565) al ons in approved Manual. TA 0) is located at the main rotor hub is at helicopter centreline. Seat at STA 83.2 .2 bit



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		Aft 6.5° to 7.5°
	Cyclic pitch (lateral):	Left 6.5° to 7.5°
		Right 4.5° to 6.5°
	Tail rotor (relative to rigging position):	
	Collective pitch:	Full-left pedal (thrust to right) +25.0° to +27.0° Full-right pedal (thrust to left) -11.0° to -13.0°
	For rigging information of main rotor and tail Basic Handbook of Maintenance Instructions.	rotor refer to Sikorsky S-300CB Model 269C-1 Helicopter
21.	Auxiliary Power Unit (APU)	n/a
22.	Life-limited Parts	Refer to latest issue of Sikorsky S-300CB Model 269C-1 Helicopter Basic Handbook of Maintenance Instructions, Appendix B - Periodic Inspection, Overhaul and Retirement Schedule, and Weight and Balance Procedures.
<u>IV. (</u>	Operating and Service Instructions	
1.	Flight Manual	Refer to latest issue of S-300CB Pilot's Flight Manual.
2.	Maintenance Manual	Refer to latest issue of Sikorsky S-300CB Model 269C-1 Helicopter Basic Handbook of Maintenance Instructions.
3.	Structural Repair Manual	n/a
4.	Weight and Balance Manual	Refer to Publication No. CSP-C1-1 Pilot's Flight Manual containing the approved Rotorcraft Flight Manual for Schweizer 300CB Helicopter Model 269C-1 Section VI 22.
5.	Illustrated Parts Catalogue	Refer to Publication No. CSP-C1-6 Schweizer Model 269C-1 Helicopter Illustrated Parts Catalog (IPC)
6.	Service Letters and Service Bulletins	As published by Schweizer RSG. For information published by previous Type Certificate holders see Note 4 in 'Section: Notes (data pertinent to all Models [])'.
7	Required Equipment	

7. Required Equipment

Refer to "Equipment List/ Weight and Balance Model 269C-1", Report No. SA-269C-22-5.

#### V. Notes (Model 269C-1 only)

 Manufacturer's eligible serial numbers: s/n --0001 and subsequent, except --0002, --0013 and --0255.



#### SECTION 5: Model 269D

JEC	Section 5. Model 2050			
<u>I. G</u>	I. General			
1.	Type/ Model/ Variant			
	1.1 Type	269		
	1.2 Model	269D		
	1.3 Variant			
2.	Airworthiness Category	Small Rotor	craft, Normal Category	
3.	Manufacturer	Schweizer R 3901 N Mair Fort Worth, U.S.A.		
4.	Type Certification Application Date	to FAA: 21 N	lovember 1987	
5.	State of Design Authority	Federal Avia	ition Administration (FAA), USA	
6.	Type Certificate Date	by FAA: by CAA SE:	14 September 1992 28 February 1994	
		by RLD:	29 May 1995	
7.	Type Certificate n°	by FAA: by CAA SE:	4H12 4/94	
		by RLD:	R-088-95	
8.	Type Certificate Data Sheet n°	by FAA: by CAA SE:	4H12 see Note 2	
		by RLD:	none	
9.	EASA Type Certification Date		er 2003, ce with CR (EU) 1702/2003, Article 2, 3., (a), t, 2 <sup>nd</sup> indented bullet.	

#### II. Certification Basis

1.	Reference Date for determining the	3 November 1987	
	applicable requirements		
2.	Airworthiness Requirements		

2

The certification basis for the Model 269D includes that of the 269C CAR Part 6, dated 15 January 1951, including Amdt. 6-1 through 6-7, and 6-8 except CAR 6.604(c). Compliance with CAR 6.401(b) effective 17 May 1958, CAR 6.637 effective 1 April 1957 and FAR 27.1323 Amdt. 27-2 effective 25 February 1968 in lieu of CAR 6.612(a) has been shown. Applicable FAR requirements covering the turbine engine installation per FAR 27 through Amdts. 27-21 in effect at time of application (3 November 1987) and noise standards per FAR 36 at time of certification are:

FAR 21.35(b)(2); 27.73(a)(2)(ii); 27.337; 27.339; 27.341; 27.361(a); 27.395; 27.397; 27.399; 27.547; 27.671; 27.901(b)(4)(c); 27.903(c); 27.907; 27.923; 27.927; 27.931; 27.939; 27.951(c); 27.955; 27.959; 27.961; 27.963; 27.965; 27.969; 27.971; 27.973; 27.975; 27.977(a)(2)(b)(c)(d); 27.993; 27.995; 27.997; 27.999; 27.1013(c); 27.1015; 27.1019; 27.1091(d)(e); 27.1093(b); 27.1121; 27.1141(d); 27.1143(d); 27.1145(b); 27.1191(a); 27.1194; 27.1195; 27.1305(f)(g)(n) through (s); 27.1323; 27.1353(f)(g); 27.1461; 27.1521(b)(5), (c)(3)(d thru f); 27.1529; 27.1557(c)(i)(iii) and 27.1583(b)(1); FAR 36 Appendix J, Amdt. 20.

3. **Special Conditions** none 4. Exemptions none 5. Deviations none 6. **Equivalent Safety Findings** none 7. Requirements elected to comply none



9.

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#### 8. Environmental Protection Requirements

8.1 Noise Requirements	See TCDSN EASA.IM.R.131
8.2 Emission Requirements	n/a
Operational Suitability Data (OSD)	See SECTION 7 below

#### III. Technical Characteristics and Operational Limitations

1.	Type Design Definition	Drawing 269D0	000-1/-5.
2.	Description	articulated main skid type standa	nine power rotorcraft, three blades n rotor, twin blade teetering tail rotor, ard landing gear, one pilot and three approved Pilot's Flight Manual)
3.	Equipment	Certification Bat the Airworthine	t required by the airworthiness rules (see sis) shall be installed on the helicopter for ess Certificate release. Refer to latest issue List 330 Helicopter" Report No. SA-269D-
4.	Dimensions		
	4.1 Fuselage	Length: Width: Height:	9.42 m 2.08 m 2.61 m
	4.2 Main Rotor	Diameter:	8.18 m
	4.3 Tail Rotor	Diameter:	1.30 m
5.	Engine		
	5.1 Model	Rolls-Royce	

		1 x Model 250-C20V	V
5.2	Type Certificate	FAA TC/TCDS n°:	E4CE
		EASA TC/TCDS n°:	EASA.IM.E.052

5.3 Limitations

5.3.1 Installed Engine Limitations

	Power [kW (hp)]	Torque [psi]	N₁ [% rpm]	ТОТ [°С]
TKOF (5 min)	175 (235)	61.7	105	810
Max Continuous	164 (220)	57.8		738
Start up/Shut down (10 sec)				810 - 927
Idle speed			59 - 65	

<u>Note:</u> 100% N<sub>1</sub> = 50 970 rpm

5.3.2 Output shaft (N<sub>2</sub>)

	Normal Operating Range N <sub>2</sub>	90% - 91%
	Installed PWR Turbine Limit 91% N <sub>2</sub>	30 294 rpm
	Installed PWR Output Shaft Limit 90% N <sub>2</sub>	5 475 rpm
	Engine torque	100% = 491.5 Nm(362.5 lb·ft)
5.3.3	Transmission Torque Limits	61.7 psi maximum 57.8 to 61.7 psi (5 min limit) 0 to 57.8 psi normal operating range



6.	Fluids (Fuel/ Oil/ Additives)		
	6.1 Fuel	-	<sup>r</sup> MIL-T-5624, Jet A, A-1, or B per ade JP-8 per MIL-T-83133
	6.2 Oil	Main and tail rotor tra MIL-L-2105E, or SAE J	e Maintenance Manual 10W2. ansmission:
	6.3 Additives	n/a	
7.	Fluid capacities		
	7.1 Fuel	Standard at STA 104.2 Fuel tank capacity: Usable fuel: Extended Range Capa Fuel tank capacity: Usable fuel:	230.1 litres (60.8 US gal) 227.1 litres (60.0 US gal)
	7.2 Oil	Engine: Main transmission: Tail rotor transm.:	4.26 litres STA 114.4 (1.125 US gal) 2.84 litres (0.75 US gal) 0.24 litres (0.063 US gal)
	7.3 Coolant System Capacity	n/a	
8.	Air Speed Limitations	For reduction on V <sub>NE</sub> V Flight Manual.	5 at MSL 5 at MSL with altitude see approved Pilot's any combination of cabin doors
9.	Rotor Speed Limitations	Power on:Nr [rMaximum471	(at 91% N <sub>2</sub> ) (at 90% N <sub>2</sub> )
10.	Maximum Operating Altitude and Temperature	Avoid operational are Flight Manual.	as shown in the approved Pilot's
	10.1 Altitude	10 000 ft (3 048 m) PA 12 800 ft (3 901 m) PA if equipped with 269A 269D7100-3 "ext. heig	A, \1002-11 main rotor inst. and
	10.2 Temperature	-17.8°C (0°F) minimun	n operating temperature
11.	Operating Limitations	the airworthiness and	truments and equipment, required by d/or operating rules, are approved, perable condition. See approved Pilot's ther limitations.
12.	Maximum Mass		rmal Category equipped with 269A1002-11 main 100-3 "ext. height" landing gear



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13.	Centre of Gravity Range		Longitudinal
		Fwd	94.0 in at 1 157 kg (2 550 lb) varying linearly to 92.0 in at 907 kg (2 000 lb) and below.
		Aft	96.0 in at 1 157 kg (2 550 lb) varying linearly to 101.0 in at 907 kg (2 000 lb and below.
			Lateral
		Right	B.L. +2.0 in at 1 157 kg (2 550 lb) varying linearly to +4.0 in at 907 kg 2 000 lb and below.
		Left	B.L. –1.0 in at 1 157 kg (2 550 lb) varying linearly to -3.0 in at 907 kg (2 000 lb) and below.
			ateral "+" CG is right of aircraft centreline, "-" is left of centreline when looking forward.
14.	Datum	2 540 centre	-
			I: The datum line (B.L. 0) is at helicopter centreline.
15.	Levelling Means	-	main rotor hub
16.	Minimum Flight Crew	1 pilot	, operating from the left seat at STA 68.6
17.	Maximum Passenger Seating Capacity	-	e configuration: (1 at STA 68.6, 1 at STA 78.6) e configuration (1 at STA 68.6, 2 at STA 78.6)
18.	Passenger Emergency Exit	2, one	on each side of the cockpit
19.	Maximum Baggage/ Cargo Loads		ading Instructions and Limitations in approved Flight Manual.
20.	Rotor Blade Control Movement		
	Main rotor (relative to rigging position):		
	Collective pitch (up and down):	12°±1	•
	Cyclic pitch (longitudinal):		ard 8.5° to 9.5° 5° to 10.0°
	Cyclic pitch (lateral):		.5° to 7.5° 6.0° to 7.0°
	Tail rotor (relative to rigging position):		
	Collective pitch:		ft pedal (thrust to right) +27.0° to +29.0° ght pedal (thrust to left) -11.0° to -13.0°
	For rigging information of main rotor and tail Handbook of Maintenance Instructions	rotor re	fer to Sikorsky S-330 Model 269D Helicopter Basic
21.	Auxiliary Power Unit (APU)	n/a	

22. Life-limited Parts

Refer to latest issue of Sikorsky S-330 Model 269D Helicopter Basic Handbook of Maintenance Instructions Appendix B - "Periodic Inspection, Overhaul and Retirement Schedule, and Weight and Balance Procedures"



#### IV. Operating and Service Instructions

1.	Flight Manual	Refer to latest issue of S-330 Pilot's Flight Manual.
2.	Maintenance Manual	Refer to latest issue of Sikorsky S-330 Model 269D Helicopter Basic Handbook of Maintenance Instructions.
3.	Structural Repair Manual	n/a
4.	Weight and Balance Manual	Publication No. CSP-D-1 Pilot's Flight Manual containing the approved Rotorcraft Flight Manual for Schweizer 330 Helicopter Model 269D Section VI.
5.	Illustrated Parts Catalogue	Publication No. CSP-D-6 Sikorsky 330 & 333 Models 269D/269D Config. "A" Helicopters Illustrated Parts Catalog (IPC)
6.	Service Letters and Service Bulletins	As published by Schweizer RSG. For information published by previous Type Certificate holders see Note 4 in 'Section: Notes (data pertinent to all Models [])'.
-	Described Frankright	

### 7. Required Equipment

Refer to latest issue of "Equipment List 330 Helicopter" Report No. SA-269D-22-2.

#### V. Notes (Model 269D only)

- Manufacturer's eligible serial numbers: s/n --0001 and subsequent, except --0007, --0011, --0013, --0017 and --0030 and all s/n containing the suffix "M" or "MB".
- 2. For the Swedish type acceptance (No 4/94) no Swedish TCDS was issued since it was a type acceptance process of the US TC 4H12. The validation is documented in the "Import Evaluation Report Nr 4/94", dated 28 February 1994.



### SECTION 6: Model 269D, variant: Configuration 'A'

## I. General

1. Type/ Model/ Variant	
-------------------------	--

	1.1 Type	269
	1.2 Model	269D
	1.3 Variant	269D Configuration 'A'
2.	Airworthiness Category	Small Rotorcraft, Normal Category
3.	Manufacturer	Schweizer RSG LLC 3901 N Main St. Fort Worth, Texas 76106 U.S.A.
4.	Type Certification Application Date	to FAA: 6 July 1999
5.	State of Design Authority	Federal Aviation Administration (FAA), USA
6.	Type Certificate Date	by FAA: 28 September 2000 by ENAC IT: 9 April 2002
7.	Type Certificate n°	by FAA: 4H12 by ENAC IT: A 386
8.	Type Certificate Data Sheet n°	by FAA: 4H12 by ENAC IT: SO/A 386
9.	EASA Type Certification Date	28 September 2003, in accordance with CR (EU) 1702/2003, Article 2, 3., (a), (i), 2 <sup>nd</sup> bullet, 2 <sup>nd</sup> indented bullet.

#### II. Certification Basis

1.	Reference Date for determining the applicable requirements	3 November 1987
2.	Airworthiness Requirements	
	The contification have fourthe Mandel 2000 Con	Constinue (A) in the

The certification basis for the Model 269D Configuration 'A' is the same as the Model 269D along with the following FAR 27 compliance upgrades as of 1 January 1999: FAR 27.337; 27.339; 27.341; 27.547; 27.923 and 27.927.

3.	Special Conditions	none
4.	Exemptions	none
5.	Deviations	none
6.	Equivalent Safety Findings	none
7.	Requirements elected to comply	none
8.	Environmental Protection Requirements	
	8.1 Noise Requirements	See TCDSN EASA.IM.R.131
	8.2 Emission Requirements	n/a
9.	Operational Suitability Data (OSD)	See SECTION 7 below

#### III. Technical Characteristics and Operational Limitations

1.	Type Design Definition	Drawing 269D0000-1/-5.
2.	Description	Light single turbine power rotorcraft, three blades articulated main rotor, twin blade teetering tail rotor, skid type standard landing gear, one pilot and three passengers (see approved Pilot's Flight Manual)



Equipment

Basic equipment required by the airworthiness rules (see Certification Basis) shall be installed on the helicopter for the Airworthiness Certificate release. Refer to latest issue of "Equipment List 330 Helicopter" Report No. SA-269D-22-2

#### 4. Dimensions

3.

5.

4.1 Fuselage	Length:	9.42 m
	Width:	2.08 m
	Height:	2.61 m
4.2 Main Rotor	Diameter:	8.18 m
4.3 Tail Rotor	Diameter:	1.30 m
Engine		
5.1 Model	Rolls-Royce	
	1 x Model 250-C20	W
5.2 Type Certificate	FAA TC/TCDS n°:	E4CE
	EASA TC/TCDS n°:	EASA.IM.E.052

#### 5.3 Limitations

5.3.1 Installed Engine Limitations

	Power [kW (hp)]	Torque [psi]	N1 [% rpm]	тот [°С]
TKOF (5 min)	188.7 (253)	67.6	105	810
Max Continuous	173 (232)	62.2		738
Start up/Shut down (10 sec)				810 – 927
Idle speed			59 - 65	

<u>Note:</u> 100% N<sub>1</sub> = 50 970 rpm

5.3.2 Output shaft (N<sub>2</sub>)

	Normal Operating Range N <sub>2</sub>	89% - 90%
	Installed PWR Turbine Limit 90% N2	29 961 rpm
	Installed PWR Output Shaft Limit 90% N2	5 414 rpm
	Engine torque	100% = 491.5 Nm (362.5 lb·ft)
5.3.3	Transmission Torque Limits	67.6 psi Maximum 62.2 to 67.6 psi (5 min limit) 0 to 62.2 psi normal operating range
Fluids (Fu	el/ Oil/ Additives)	
6.1 Fuel		Grade JP-4 or JP-5 per MIL-T-5624, Jet A, A-1, or B per ASTM D-1655, and Grade JP-8 per MIL-T-83133
6.2 Oil		Engine: MIL-L-7808*, or MIL-L-23699 * Reference Rolls-Royce Maintenance Manual 10W2. Main and tail rotor transmission: MIL-L-2105E, or SAE J2360** ** For detailed information see S-333 Basic HMI.
6.3 Addi	tives	n/a



6.

7.	Fluid capacities			
	7.1 Fuel	Fuel ta Usable Extend	ed Range Capao nk capacity:	: 230.1 litres (60.8 US gal) 227.1 litres (60.0 US gal) city at STA 104.2: 280.5 litres (74.1 US gal) 276.3 litres (73.0 US gal)
	7.2 Oil		: ransmission: cor transm.:	4.26 litres STA 114.4 (1.125 US gal) 2.84 litres (0.75 US gal) 0.24 litres (0.063 US gal)
	7.3 Coolant System Capacity	n/a		
8.	Air Speed Limitations	VNE: VNE PWR VNE Door	120 KIAS (Max. ma off: 94 KIAS	ass 2 301-2 550 lb) at MSL ass 2 300 lb and below) at MSL for any combination of
				vith altitude see approved Pilot's ited Supplements.
9.	Rotor Speed Limitations	Norma Power Maxim Minim	on: N <sub>r</sub> [r um 471	ange [rpm]: 466 – 471 pm] (at 90% N2) (at 89% N2)
		Power Maxim Minim	um 500	pm]
10.	Maximum Operating Altitude and Temperature		operational area Manual.	as shown in the approved Pilot's
	10.1 Altitude	13 000	ft (3 962 m) PA	
	10.2 Temperature	-17.8°C	C (0°F) minimum	n operating temperature
11.	Operating Limitations	Non-ici * With the a insta	airworthiness and	ruments and equipment, required by I/or operating rules, are approved, perable condition. See approved Pilot's her limitations.
12.	Maximum Mass		g (2 550 lb) No CTION NOTES,	
13.	Centre of Gravity Range			Longitudinal
		Fwd		7 kg (2 550 lb) varying linearly to kg (2 000 lb) and below.
		Aft		7 kg (2 550 lb) varying linearly to 7 kg (2 000 lb and below.
		L		Lateral
		Right		1 157 kg (2 550 lb) varying linearly 07 kg 2 000 lb and below.
		Left		1 157 kg (2 550 lb) varying linearly 17 kg (2 000 lb) and below.
		N		abt of allowedt accetualized "" " is laft of

<u>Note:</u> Lateral "+" CG is right of aircraft centreline, "-" is left of aircraft centreline when looking forward.



14.	Datum	Longitudinal: The datum line (STA 0) is located at 2 540 mm (100.0 in) forward of the main rotor hub centreline.
45		Lateral: The datum line (B.L. 0) is at helicopter centreline.
15.	Levelling Means	Top of main rotor hub
16.	Minimum Flight Crew	1 pilot, operating from the left seat at STA 68.6
17.	Maximum Passenger Seating Capacity	3 place configuration: (1 at STA 68.6, 1 at STA 78.6) 4 place configuration (1 at STA 68.6, 2 at STA 78.6)
18.	Passenger Emergency Exit	2, one on each side of the cockpit
19.	Maximum Baggage/ Cargo Loads	See Loading Instructions and Limitations in approved Pilot's Flight Manual.
20.	Rotor Blade Control Movement	
	Main rotor (relative to rigging position):	
	Collective pitch (up and down):	12°±1°
	Cyclic pitch (longitudinal):	Forward 8.5° to 9.5° Aft 9.5° to 10.0°
	Cyclic pitch (lateral):	Left 6.5° to 7.5° Right 6.0° to 7.0°
	Tail rotor (relative to rigging position):	
	Collective pitch:	Full-left pedal (thrust to right) +27.0° to +29.0° Full-right pedal (thrust to left) -11.0° to -13.0°
	For rigging information of main rotor and tail	rotor refer to S-333 Basic HMI.
~ ~		
21.	Auxiliary Power Unit (APU)	n/a
21. 22.	Life-limited Parts	n/a Refer to latest issue of Sikorsky S-333 Model 269D Configuration "A" Helicopter Basic Handbook of Maintenance Instructions Appendix B - "Periodic Inspection, Overhaul and Retirement Schedule, and Weight and Balance Procedures".
22.		Refer to latest issue of Sikorsky S-333 Model 269D Configuration "A" Helicopter Basic Handbook of Maintenance Instructions Appendix B - "Periodic Inspection, Overhaul and Retirement Schedule, and
22.	Life-limited Parts	Refer to latest issue of Sikorsky S-333 Model 269D Configuration "A" Helicopter Basic Handbook of Maintenance Instructions Appendix B - "Periodic Inspection, Overhaul and Retirement Schedule, and
22. <u>IV. (</u>	Life-limited Parts Dperating and Service Instructions	Refer to latest issue of Sikorsky S-333 Model 269D Configuration "A" Helicopter Basic Handbook of Maintenance Instructions Appendix B - "Periodic Inspection, Overhaul and Retirement Schedule, and Weight and Balance Procedures".
22. <u>IV. (</u> 1.	Life-limited Parts Operating and Service Instructions Flight Manual	Refer to latest issue of Sikorsky S-333 Model 269D Configuration "A" Helicopter Basic Handbook of Maintenance Instructions Appendix B - "Periodic Inspection, Overhaul and Retirement Schedule, and Weight and Balance Procedures". Refer to latest issue of S-333 Pilot's Flight Manual. Refer to latest issue of Sikorsky S-333 Model 269D Config. 'A' Helicopter Basic Handbook of Maintenance
22. <u>IV. (</u> 1. 2.	Life-limited Parts Operating and Service Instructions Flight Manual Maintenance Manual	Refer to latest issue of Sikorsky S-333 Model 269D Configuration "A" Helicopter Basic Handbook of Maintenance Instructions Appendix B - "Periodic Inspection, Overhaul and Retirement Schedule, and Weight and Balance Procedures". Refer to latest issue of S-333 Pilot's Flight Manual. Refer to latest issue of Sikorsky S-333 Model 269D Config. 'A' Helicopter Basic Handbook of Maintenance Instructions.
22. <u>IV. (</u> 1. 2. 3.	Life-limited Parts Deperating and Service Instructions Flight Manual Maintenance Manual Structural Repair Manual	Refer to latest issue of Sikorsky S-333 Model 269D Configuration "A" Helicopter Basic Handbook of Maintenance Instructions Appendix B - "Periodic Inspection, Overhaul and Retirement Schedule, and Weight and Balance Procedures". Refer to latest issue of S-333 Pilot's Flight Manual. Refer to latest issue of Sikorsky S-333 Model 269D Config. 'A' Helicopter Basic Handbook of Maintenance Instructions. n/a Publication No. CSP-D-8 Pilot's Flight Manual containing the approved Rotorcraft Flight Manual for Schweizer 333
22. <u>IV. (</u> 1. 2. 3. 4.	Life-limited Parts Deperating and Service Instructions Flight Manual Maintenance Manual Structural Repair Manual Weight and Balance Manual	Refer to latest issue of Sikorsky S-333 Model 269D Configuration "A" Helicopter Basic Handbook of Maintenance Instructions Appendix B - "Periodic Inspection, Overhaul and Retirement Schedule, and Weight and Balance Procedures". Refer to latest issue of S-333 Pilot's Flight Manual. Refer to latest issue of Sikorsky S-333 Model 269D Config. 'A' Helicopter Basic Handbook of Maintenance Instructions. n/a Publication No. CSP-D-8 Pilot's Flight Manual containing the approved Rotorcraft Flight Manual for Schweizer 333 Helicopter Model 269D Configuration 'A' Section VI Publication No. CSP-D-6 Sikorsky 330 & 333 Models 269D/269D Config. 'A' Helicopters Illustrated Parts



V. Notes (Model 269D, variant Configuration 'A' only)

 Manufacturer's eligible serial numbers: Optional configuration for production helicopters s/n --0026 and subsequent and for all other helicopters incorporating Retrofit Kit no. SA-269D-K-20. Production Configuration A helicopters have 'A' at the end of s/n. Retrofit Configuration 'A' helicopters have no '-A' at the end of s/n. Both production and retrofit helicopters have an additional 'Configuration A' Data Plate affixed next to standard data plate.



SECTION: NOTES (data pertinent to all Models except when specifically indicated)

- 1. Aircraft serial numbers are coded to show the month and year of manufacture sequence. Example: 1130103
  - 11 month of manufacture was November
  - 3 year of manufacture was 1963

0103 Serial number in consecutive order from 0001 for each model

Model 269C Helicopters, s/n 1065, s/n 1075 and subsequent will be delivered without the manufacturing date coding as part of the serial number. Serial numbers are prefixed by the letter "S" starting with s/n S1166 and up.

- 2. Current weight and balance report, including list of equipment including certificated empty weight and loading instructions, must be provided for each helicopter at the time of original airworthiness certification and at all times thereafter (except in the case of operators having an appropriate weight control system). Ballast, when necessary, must be carried in accordance with the loading instructions in the Rotorcraft Flight Manual.
- 3. The following placard must be installed in clear view of the pilot:

"This Helicopter must be operated in compliance with the operating limitations specified in the pertinent Rotorcraft Flight Manual."

For additional placards, see the pertinent Rotorcraft Flight Manual.

4. Service Bulletin information is organised by document prefix.

Please see the following breakdown:

- 'N-' = Hughes Aircraft (model effectivity noted inside document)
- 'B-' = old Schweizer Company (model effectivity 269A, 269B, 269C)
- 'C1B-' = old Schweizer Company (model effectivity 269C-1)
- 'DB-' = old Schweizer Company (model effectivity 269D)
- 'ASB B-' = Sikorsky Aircraft Company (model effectivity 269A, 269B, 269C)
- 'ASB C1B' = Sikorsky Aircraft Company (model effectivity 269C-1)
- 'ASB DB-' = Sikorsky Aircraft Company (model effectivity 269D)

SECTION: NOTES (data pertinent to all Models, except 269C-1, 269D and variant 269D Configuration 'A')

1. (a) The retirement times of critical parts are listed in the following table. These values of retirement or service life cannot be increased without EASA approval by. (See NOTE 3 for Model 269D and Note 4 for Model 269C-1)):

Description	p/n	Model 269A s/n 0001 thru 0008 [h]	Model 269A s/n 0011 & subs. Model 269B s/n 0001 & subs. [h]	Model 269C s/n 0004& subs. [h]
 Blade Assembly - M/R	269-1100	1 366		
	269A1125		1 366	
	269A1131	1 366	1 366	
	269A1131-1	1 366	1 366	
	269A1160			5 500
	269A1185-1,-7			5 500
	269A1185-9			3 050
	269A1190		5 500	
	269A1190-1		5 500	
	269B1145		1 366	
	269B1145-1		1 366	
	269B1145-25		1 366	
Pitch Brg. Shaft - M/R	269A1240-7			3600
Dampers-Elastomeric - M/R See NOTE 1(e)	269A1290-1, -3		6000	6000



		Model 269A s/n 0001 thru 0008	Model 269A s/n 0011 & subs. Model 269B s/n 0001 & subs.	Model 269C s/n 0004& subs.
Description	p/n	[h]	[h]	5003. [h]
Mast - M/R	269-2165	1 900		
	269A2010-5, -15			13 590
Thrust Bearing - M/R	269A5050-73		3 000	
	269A5050-63, -95 269A5050-50, -51	300	300	3 000
Tail Boom Assy (when 269ASK16 or	269A2320 with 269A2324		17 370	
269A6034 T/R is installed)	–13, -11 centre attach fitting installed			
	269A2320 with 269A2324		4 100	
	Basic, -7 centre attach fitting installed		4 100	
Tail Boom Assy	269A2320-7			2 100
	with269A2324-11 centre attach fitting installed			
	269A2320-7 with 269A2324-7 centre attach fitting installed			500
	269A2320-9		17 370	
	269A2320-11			2 100
	269A2320-17,-21			4 200
	269A2320-19			2 100
Tail Boom Struts (see NOTE 1 (f))	269A2015-5 269A2015-11, -13, -15, - 17, -113, -213, -215			500 10 700
Stab. Assy - Vert.	269A2419-3			20 540
Stab. Assy - Horiz.	269-2500	2 500		
	269A2511		2 500	
(when 269A2516 zero time Stab. is installed with 269ASK16 or 269A6034	269A2516		2500	
T/R)	269A2516		3 070	
.,,	269A2516-9			2 500
	269A2516-21			4 200
Main Gear Box Pinion Assy	269-5103	2 250		
	269A5103		6 000	6 000
	269A5103-9		6 000	6 000
	269A5103-21		6 000 6 000	6 000 6 000
Main Rotor Drive Shaft	269A5103-31, 41, -51, -55 269-5301	1 195		
	269A5305-3, -103		3 000	
	269A5305-11, -111			1 900
Main Rotor Drive Shaft (splined)	269A5326-1, -5			3 200
Main Rotor Hub (splined)	269A5325-1			8 000
Carrier Assembly-Ring Gear, see NOTE 1(h)	269A5194	6 000	6 000	6 000
Lower Pulley Coupling Shaft	269-5412	1 500		
Lower Pulley Coupling Shaft (269A5504-5 Assy)	269A5504-3		1 500	1 500
Lower Pulley Coupling Shaft (269A5559 Assy)	269A5559-3		6 000	6 000
Idler Pulley Bearings	269A5050-58		200	
	269A5050-62			600



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Description         p/n         [h]         [h] <th< th=""><th>Description</th><th>p/n</th><th>Model 269A s/n 0001 thru 0008 [h]</th><th>Model 269A s/n 0011 &amp; subs. Model 269B s/n 0001 &amp; subs.</th><th>Model 269C s/n 0004&amp; subs.</th></th<>	Description	p/n	Model 269A s/n 0001 thru 0008 [h]	Model 269A s/n 0011 & subs. Model 269B s/n 0001 & subs.	Model 269C s/n 0004& subs.
269A5609          3 000            see NOTE 1(d)         369A5406          unlimited         8 600           269A5626-3, -5          unlimited         8 600           269A5626-3, -5           8 600           269A5626-3, -5          3 000            T/R Drive Shaft         269A5607          3 000            Shaft Assy - T/R Drive         269A5701, -3          3 000            Shaft Assy - T/R Drive         269A6040, -85 M          3 000            Shaft Assy - T/R Drive         269A6040, -5, -5M          3 000            Shaft Assy - T/R Drive         269A6040, -5, -5M          3 000            269A6040, -7, -9, 9M          3 000            Spline Adapter Fitting         269A6035, -17, -21          5 000            Spline Adapter Fitting         269A6035, -17, -21          5 000            269A6035, 9, -19, -23          5 000          269A602           269A605          2 80					
see NOTE 1(d) 369A5406 unlimited 8 600 see NOTE 1(d) 269A5425, -3, -5 unlimited 8 600 269A5626-3, -5 8 000 Drive Spline - Aft End 269-5607 1 800 3 000 T/R Drive Shaft 269A5607 3 000 (includes end fittings) 269A5701, -3 3 000 (includes end fittings) 269A5701, -3 3 000 269A6040-5, -5M 3 000 269A6040-5, -5M 3 000 269A6040-5, -5M 3 000 269A6040-7, -9, 9M 6 000 269A5K09 3 000 269A5K09 3 000 269A5K04 3 000 269A5K04 9 000 269A5K04 9 000 269A5K04 9 000 269A6035, -17, -21 5 000 269A6035 5 000 269A6124 9 000 269A6124 9 000 269A6124 9 60 269A6124 9 60 269A6124 9 60 269A6055 2 800 3 540 269A6065 -507 2 800 3 540 269A6065 -507 2 800 3 540 269A6065 -507 2 800 3 540 269A6065 2 800 5 100 369A1706-505, -507 2 800 5 100 369A17	Shaft - Input I/R GB				
see NOTE 1(d)         369A5425, -3, -5          unlimited         8 600           269A5626-3, -5          8 600           Drive Spline - Aft End         269-5607         1 800          8 600           T/R Drive Shaft         269A5607          3 000            Shaft Assy - T/R Drive         269A5701, -3          3 000            (includes end fittings)         269A6040,-BSC M          3 000            269A6040-7, -9, 9M          3 000            269A6040-7, -9, 9M          6 000         269A6040-7, -9, 9M            269A6040-7, -9, 9M          3 000            269A6040-7, -9, 9M          6 000         269A6035           269A6035, -17, -21          5 000            269A6035, -17, -21          5 000            269A6035, -19, -23          9 000         269-6100         960            269A602124-9          2 69A6035          2 69A6035          2 800         3 540           2					
269A5626-3, -5           8 600           Drive Spline - Aft End T/R Drive Shaft         269-5607         1 800             Shaft Assy - T/R Drive         269-5701         3 000             Shaft Assy - T/R Drive         269-5701, -3          3 000            Shaft Assy - T/R Drive         269A6040, BSC M          3 000            Sbaft Assy - T/R Drive         269A6040-5, -5M          3 000            269A6040-7, -9, 9M           6 000         269A5K09          6 000           269A6035, -17, -21          5 000          269A6035          269A6035          269A6035          269A6035          269A6035          269A6035          269A6035          269A6035          9000         269-6100         960          269A6035          269A6035          269A6035          269A6035          269A6035          269A604          269A604          269A604        <					
Drive Spline - Aft End         269-5607         1 800          3 000            T/R Drive Shaft         269A5607          3 000             Shaft Assy - T/R Drive         269A5701, -3         3 000          3 000            Shaft Assy - T/R Drive         269A6040,-BSC M          3 000            Shaft Assy - T/R Drive         269A6040,-S, SM          3 000            269A6040-5, -SM          3 000          6 000           269A6040-7, -9, 9M          5 000          269A6040          6 000         269A500          6 000         269A500          6 000         269A500          269A6035          269A6035          269A6035          269A6035          269A6035          269A6035          269A6035          269A6035          269A6035          269A604          269A604          269A604          269A604          269A604          269A604	see NOTE 1(d)				
T/R Drive Shaft       269A5607        3 000          Shaft Assy - T/R Drive       269-5701       3 000        3 000          Shaft Assy - T/R Drive       269A5701, -3        3 000          Shaft Assy - T/R Drive       269A6040, -BSC M        3 000          269A6040-7, -9, 9M        3 000          269A6040-7, -9, 9M        6 000       269A500          269A6040-7, -9, 9M        3 000          Spline Adapter Fitting       269A6035, -17, -21        5 000          Blade Assy - T/R       269A6035, -17, -21        5 000          269A6035, -17, -21        5 000        269A6035, -17, -21        9 000         269A6035, -9, -19, -23        5 000        269A6035        9 000          269A6035, -9, -19, -23        9 600        269A605        269A605        269A605        269A605        269A605        269A6065       2800       3 540 <td></td> <td>269A5626-3, -5</td> <td></td> <td></td> <td>8 600</td>		269A5626-3, -5			8 600
Shaft Assy - T/R Drive (includes end fittings)       269-5701, -3        3 000          Shaft Assy - T/R Drive       269A6040,-BSC M        3 000          269A6040-5, -5M        3 000          269A6040-7, -9, 9M        3 000          269A6040-7, -9, 9M        6 000         269A5X09        3 000          Spline Adapter Fitting       269A5035, -17, -21        5 000          Blade Assy - T/R       269A6035, -17, -21        5 000          269A6035, -17, -21        5 000        269A6035         269A6035, -17, -21        5 000        269A6035        9 000         269A6035, -17, -21        5 000        269A6035        9 000       269-6100       960        269A6035        269A6035        269A603       269-6100       960        269A604       269-6100       269A6065        269A6065        269A6065       269A6065       269A6065       269A6065       269A6065       269A6065       269A6065 <td>Drive Spline - Aft End</td> <td>269-5607</td> <td>1 800</td> <td></td> <td></td>	Drive Spline - Aft End	269-5607	1 800		
(includes end fittings)         269A5701, -3          3 000            Shaft Assy - T/R Drive         269A6040, -BSC M          3 000            269A6040-5, -5M          3 000            269A6040-7, -9, 9M          6 000           269A6040-7, -9, 9M          6 000           269A6035, -17, -9, 9M          5 000            Spline Adapter Fitting         269A6035, -17, -21          5 000            269A6035, -17, -21          5 000          269A6035         5 000            269A6035, -17, -21          5 000          269A6035          9 000           269A6035, -17, -23          5 000          269A6035          9 000           269A6124          960          269A6035          269A6035            269A6035-90, -19, -23          960          269A6035          269A6035          269A6035          2800         3540         269A605 <t< td=""><td>T/R Drive Shaft</td><td>269A5607</td><td></td><td>3 000</td><td></td></t<>	T/R Drive Shaft	269A5607		3 000	
Shaft Assy - T/R Drive       269A6040,-BSC M        3 000          269A6040-5, -5M        3 000          269A6040-7, -9, 9M        6 000         269A6040-7, -9, 9M        6 000         269A6040-7, -9, 9M        6 000         269A6040-7, -9, 9M        6 000         269A6040-7, -9, 9M        6 000         Spline Adapter Fitting       269A6035, -17, -21        5 000          Blade Assy - T/R       269A6035, -17, -21        5 000          269A6035M        5 000        269A6035         269A6124        9 000       269-6100       960          269A6124        960           269A6124-9        960           269A6055        2 800       3 540       269A6055        2 800       5 100         369A1706        2 800       5 100       369A1706-505, -507        2 800       5 100         369A6176-507, -507        2 800       5 100 </td <td>Shaft Assy - T/R Drive</td> <td>269-5701</td> <td>3 000</td> <td></td> <td></td>	Shaft Assy - T/R Drive	269-5701	3 000		
269A6040-5, -5M          3 000            269A6040-7, -9, 9M          6 000           269A5K09          3 000            Spline Adapter Fitting         269A5K04          5 000            Blade Assy - T/R         269A6035, -17, -21          5 000            269A6035M          5 000          269A6035-9, -19, -23          9 000           269-6100         960          269A6124          9 000            269A6124-9          960           269A6055          269A6124           269A6124          960           269A6124          269A6124          269A6124          269A6105	(includes end fittings)	269A5701, -3		3 000	
269A6040-7, -9, 9M           6 000           269ASK09         3 000            Spline Adapter Fitting         269ASK04          5 000            Blade Assy - T/R         269A6035, -17, -21          5 000            269A6035M          5 000          269A6035           269A6035-9, -19, -23          5 000            269A6124          9 000         269-6100         960            269A6124-9          960          269A605          269A605          2800         3 540           269A6065          2 800         3 540         269A6065          2 800         3 540           269A6065          2 800         3 540         269A605         2 800         3 540           269A6065          2 800         3 540         269A605          2 800         3 540           269A605          2 800         5 100         369A1706          2 800         5 100           Torsion Shaft - T/R Blade (	Shaft Assy - T/R Drive	269A6040,-BSC M		3 000	
269ASK09          3 000            Spline Adapter Fitting         269ASK04          5 000            Blade Assy - T/R         269A6035, -17, -21          5 000            269A6035M          5 000            269A6035-9, -19, -23          5 000            269A6124          9 000         269-6100         960            269A6124          960          269A6025          960            269A6124-9          960          269A6025         2800         3 540           269A6065          2 800         3 540         269A6065         280         5 100           269A6065          2 800         5 100         369A1706-505, -507          2 800         5 100           369A1706-505, -507          2 800         5 100         369A1706-505, -507             269A6108          2 800         5 100             269A6108          1 200		269A6040-5, -5M		3 000	
Spline Adapter Fitting Blade Assy - T/R         269ASK04          20 000            269A6035, -17, -21          5 000            269A6035M          5 000            269A60359          5 000            269A60359, -19, -23          5 000            269A6035-9, -19, -23          9 000         269-6100         960            269A6124          960          269A6124          960            269A6124-9          960          269A6055         2800         3540           269A6065          2800         3540         269A6065         2800         5100           269A6065          2800         5100         369A1706-505, -507          2800         5100           Torsion Shaft - T/R Blade (NOTE 2)         269-6108         1200             269A6219          1200             269A6221          960            269A6221		269A6040-7, -9, 9M			6 000
Blade Assy - T/R       269A6035, -17, -21        5 000          269A6035M        5 000          269A6035-9, -19, -23        5 000          269A6035-9, -19, -23        9000       269-6100       960          269A6124        960        269A6124       960          269A6124-9        960        269A6124       960          269A6124-9        960        269A6065       2800       3540         269A6065       269A6065        2800       3540       269A6065       2800       5100         269A6065       507        2800       5100       369A1706       2800       5100         269A6065       507        2800       5100       269A608        269A6108       1200          269A6108        1200        269A6219        269A6219          Hub - T/R       269A6221        960        269A6221       960          269A6221		269ASK09		3 000	
269A6035M        5 000          269A5X15        5 000          269A6035-9, -19, -23        9000         269-6100       960           269A6124        960          269A6124-9        960          269A6124-9        960          269A6055        2 800       3 540         269A6065-507        2 800       5 100         269A6108       1 200           269A6108       1 200           269A6219        1 200          269A6219        1 200          269A6219        1 200          269A6221        960          269A6221        960          269A6247       960           269A6247        960          269A6247       900           269-7506       900	Spline Adapter Fitting	269ASK04		20 000	
269ASK15        5 000          269A6035-9, -19, -23         9 000         269-6100       960           269A6124        960          269A6124-9        960          269A6124-9        960          269A6124-9        960          269A6055        2 800       3 540         269A6065        2 800       3 540         269A6065        2 800       5 100         269A6065        2 800       5 100         369A1706-505, -507        2 800       5 100         269A6065        2 800       5 100         369A1706-505, -507        2 800       5 100         269A6108       1 200           269A6219        1 200          Hub - T/R       269-6204       960          269A6221        960          269A6247        960          269A6247       900<	Blade Assy - T/R	269A6035, -17, -21		5 000	
269A6035-9, -19, -23         9000         269-6100       960           269A6124        960          269A6124-9        960          269A6124-9        960          Retention Straps - T/R       369A1706        2 800       3 540         269A6065        2 800       3 540         269A6065-507        2 800       5 100         269A6108        2 800       5 100         369A1706-505, -507        1 200          269A6108        1 200          269A6219        1 200          14b - T/R       269-6204       960          269A6221        960          269A6247        960		269A6035M		5 000	
269-6100       960        960          269A6124        960          269A6124-9        960          Retention Straps - T/R       369A1706        2 800       3 540         269A6055        2 800       3 540         269A6065-507        2 800       5 100         269A6065-507        2 800       5 100         369A1706-505, -507        2 800       5 100         Torsion Shaft - T/R Blade (NOTE 2)       269-6108       1 200          269A6219        1 200          Hub - T/R       269-6204       960          269A6221        960          269A6247        960 <td< td=""><td></td><td>269ASK15</td><td></td><td>5 000</td><td></td></td<>		269ASK15		5 000	
269A6124        960          269A6124-9        960          Retention Straps - T/R       369A1706        2 800       3 540         269A6055        2 800       3 540         269A6065        2 800       3 540         269A6065-507        2 800       5 100         369A1706-505, -507        2 800       5 100         369A1706-505, -507        2 800       5 100         369A1706-505, -507        2 800       5 100         Torsion Shaft - T/R Blade (NOTE 2)       269-6108       1 200          269A6108        1 200          269A6219        1 200          Hub - T/R       269-6204       960          269A6221        960          269A6247        960          269A6247        960          Bellcrank - Lat. Pitch       269-7506       900		269A6035-9, -19, -23			9 000
269A6124-9          960            Retention Straps - T/R         369A1706          2 800         3 540           269A6065          2 800         3 540           269A6065.507          2 800         5 100           369A1706-505, -507          2 800         5 100           369A1706-505, -507          2 800         5 100           369A1706-505, -507          2 800         5 100           Torsion Shaft - T/R Blade (NOTE 2)         269-6108         1 200            269A6219          1 200            269A6221          960            269A6247         960             269A6247         900             Bellcrank - Lat. Pitch         269-7506         900		269-6100	960		
Retention Straps - T/R       369A1706        2 800       3 540         269A6065        2 800       3 540         269A6065-507        2 800       5 100         369A1706-505, -507        2 800       5 100         369A1706-505, -507        2 800       5 100         Torsion Shaft - T/R Blade (NOTE 2)       269-6108       1 200          269A6108        1 200          269A6219        1 200          Hub - T/R       269-6204       960          269A6221        960          269A6247        960          Bellcrank - Lat. Pitch       269-7506       900		269A6124		960	
269A6065        2 800       3 540         269A6065-507        2 800       5 100         369A1706-505, -507        2 800       5 100         Torsion Shaft - T/R Blade (NOTE 2)       269-6108       1 200          269A6108        1 200          269A6219        1 200          Hub - T/R       269-6204       960          269A6221        960          269A6247        960          Bellcrank - Lat. Pitch       269-7506       900		269A6124-9		960	
269A6065-507        2 800       5 100         369A1706-505, -507        2 800       5 100         Torsion Shaft - T/R Blade (NOTE 2)       269-6108       1 200          269A6108        1 200          269A6219        1 200          Hub - T/R       269-6204       960          269A6221        960          269A6247        960          Bellcrank - Lat. Pitch       269-7506       900	Retention Straps - T/R	369A1706		2 800	3 540
369A1706-505, -507          2 800         5 100           Torsion Shaft - T/R Blade (NOTE 2)         269-6108         1 200             269A6108          1 200             269A6219          1 200             Hub - T/R         269-6204         960             269A6221          960             269A6247          960             Bellcrank - Lat. Pitch         269-7506         900		269A6065		2 800	3 540
Torsion Shaft - T/R Blade (NOTE 2)       269-6108       1 200           269A6108        1 200          269A6219        1 200          Hub - T/R       269-6204       960           269A6221        960          269A6221        960          269A6247        960          Bellcrank - Lat. Pitch       269-7506       900		269A6065-507		2 800	5 100
269A6108        1 200          269A6219       1 200          Hub - T/R       269-6204       960          269A6221        960          269A6221        960          269A6247        960          Bellcrank - Lat. Pitch       269-7506       900		369A1706-505, -507		2 800	5 100
269A6219          1 200            Hub - T/R         269-6204         960             269A6221          960             269A6221          960             269A6247          960             Bellcrank - Lat. Pitch         269-7506         900	Torsion Shaft - T/R Blade (NOTE 2)	269-6108	1 200		
Hub - T/R       269-6204       960           269A6221        960          269A6247        960          Bellcrank - Lat. Pitch       269-7506       900		269A6108		1 200	
269A6221          960            269A6247          960            Bellcrank - Lat. Pitch         269-7506         900		269A6219		1 200	
269A6247          960            Bellcrank - Lat. Pitch         269-7506         900	Hub - T/R	269-6204	960		
Bellcrank - Lat. Pitch 269-7506 900		269A6221		960	
		269A6247		960	
Idler Mixer 269A7506 900	Bellcrank - Lat. Pitch	269-7506	900		
	Idler Mixer	269A7506		900	

(b) It is prohibited to interchange life limited components between different series of helicopters (i.e. 369/269). Components which have been interchanged between series of helicopters prior to revision 19 of FAA TCDS 4H12 may continue in service to their respective retirement lives. Life limited components interchanged between Models, configurations, or previously between series must be restricted to the lowest service life indicated for the Models or configurations affected. Parts are applicable only on Models under which a service life is listed. Interchanged components with known service hours but without Model application identification may not exceed the lowest life listed for any applicable Model. If the service hours are not known, regardless of Model application, the component cannot be interchanged to Models that list the component as limited life.

(c) Life limited components removed when life limit has been reached must be destroyed or permanently marked to prevent return to service.

(d) Input Gearshaft assy. T/R, P/N 369A5406 (Input Only), 369A5425 and 369A5425-3 having accumulated any Military (OH-6A Model 369A) time in service must be limited to a total service life of 530 hours.

(e) (Elastomeric Dampers) Mandatory inspection required in accordance with the 269 Series "Helicopter Maintenance Instruction" (HMI) requirements at 600-hour intervals for operation up to 4 200 hours and at 300-hour intervals thereafter to a total damper operational service time of 6 000 hours. For



Models 269A and 269B Main Rotor Elastomeric Dampers P/N 269A1290 can only be used with Main Rotor Blades P/N 269A1190-1.

(f) AD 76-18-01 required modifying 269A2015-5 to 269A2015-11 configuration within 500 hours or by September 7, 1977 in any case.

(g) Alpha and/or numeric suffixes added to part numbers denote special manufacturing or handling procedures and do not alter the replacement requirements of the part. For example, 269A5305-11 and 269A5305-11M2 are subject to the same requirements.

(h) 269A5193 Carrier is part of 269A5194 Carrier Assembly

 The limited service life for all P/N 369A1706 or 269A6065 tension torsion strap assemblies used on any 269A Configuration d (TH-55A) series helicopter, while the helicopter was operated by the U.S. Army, is reduced to 1 531 hours as defined in Schweizer Service Information Notice No.N-214. All such parts in service or spares inventory, which have exceeded 1 531 hours total time in service, must be removed and scrapped.

The TH-55A is a military helicopter with no civil counterpart. For conversion to the Model 269A, contact the manufacturer.

3. (a) The retirement times of critical parts for Model 269D are listed in the Handbook of Maintenance Instructions, Appendix B, CSP-D-4, Airworthiness Limitations Section, dated March 11, 2010. These values of retirement or service life cannot be increased without EASA approval.

(b) The retirement times of critical parts for Model 269D Configuration "A" are listed in the Handbook of Maintenance Instructions, Appendix B, CSP-D-11, Airworthiness Limitations Section, dated March 11, 2010. These values of retirement or service life cannot be increased without EASA approval.

#### (c) reserved

(d) It is prohibited to interchange life limited components between different series of helicopters (i.e. 369/269). Components which have been interchanged between series of helicopters prior to revision 19 of FAA TCDS 4H12 may continue in service to their respective retirement lives. Life limited components interchanged between Models, configurations, or previously between series must be restricted to the lowest service life indicated for the Models or configurations affected. Parts are applicable only on Models under which a service life is listed. Interchanged components with known service hours but without Model application identification may not exceed the lowest life listed for any applicable Model. If the service hours are not known, regardless of Model application, the component cannot be interchanged to Models that list the component as limited life.

(e) Life limited components removed when life limit has been reached must be destroyed or permanently marked to prevent return to service.

(f) Alpha and/or numeric suffixes added to part numbers denote special manufacturing or handling procedures and do not alter the replacement requirements of the part. For example, 269A5305-11 and 269A5305-11M2 are subject to the same requirements.

4. (a) The retirement times of critical parts for Model 269C-1 are listed in the following table. These values

of retirement or service life cannot be increased without EASA approval.	
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		Model 269C-1 s/n 0001 & subs.
Description	p/n	[h]
Main Rotor Blade	269A1185-1,-7	5 500
	269A1185-9	3 050
Pitch Bearing Shaft	269A1240-7	4 000
Elastometic Dampers	269A1290-3	6 000
M/R Input Pinion	269A5103-51, -55	8 000
M/R Drive Shaft (bolted)	269A5305-111	2 000
M/R Drive Shaft (splined)	269A5326-1	4 000
M/R Hub (splined)	269A5325-1	8 000



		Model 269C-1 s/n 0001 & subs.
Description	p/n	[h]
T/R Drive Shaft	269A6040-7,-9,-9M	6 000
Shaft-Input T/R GB	269A5626-5	8 600
T/R Blade	269A6035-23	9 000
T/R T-T Straps	269A6065-507	5 100
Main Rotor Mast	269A2010-5, -15	13 590
Tail Boom Assy	269A2320-13	2 100
	269A2320-15	4 200
Tail Boom Strut	269A2015-11, -13, -15, -17, - 113, -213, -215	10 700
Horizontal Stab.	269A2516-21	4 200
Lower Pulley Coupling Shaft	269A5559-3	6 000
Thrust Bearing-M/R	269A5050-63, -95	4 200
Carrier Assy-Ring Gear see NOTE 4(h)	269A5194	8 000

(c) It is prohibited to interchange life limited components between different series of helicopters (i.e. 369/269). Components which have been interchanged between series of helicopters prior to revision 19 of FAA TCDS 4H12 may continue in service to their respective retirement lives. Life limited components interchanged between Models, configurations, or previously between series must be restricted to the lowest service life indicated for the Models or configurations affected. Parts are applicable only on Models under which a service life is listed. Interchanged components with known service hours but without Model application identification may not exceed the lowest life listed for any applicable Model. If the service hours are not known, regardless of Model application, the component cannot be interchanged to Models that list the component as limited life.

(d) Life limited components removed when life limit has been reached must be destroyed or permanently marked to prevent return to service.

(e) The 269A2402 Vertical Stabilizer is part of the 269A2320-13 Tail Boom Assembly. The Vertical Stabilizer has the same service life (2 100 hours) as does the Tail Boom and therefore the vertical stabilizer shall be retired with the Tail Boom Assembly.

(f) Some Parts may appear to be interchangeable between the Model 269C-1 and other 269 series helicopters. However due to differences in maintenance schedules, only the most current dash numbers as defined in Note 9(b) are applicable for installation on the Model 269C-1.

(g) Alpha and/or numeric suffixes added to part numbers denote special manufacturing or handling procedures and do not alter the replacement requirements of the part. For example, 269A5305-11 and 269A5305-11M2 are subject to the same requirements.

(h) 269A5193 Carrier is part of 269A5194 Carrier Assembly.



#### SECTION 7: OPERATIONAL SUITABILITY DATA (OSD)

The OSD elements listed below are approved by the European Union Aviation Safety Agency as per Commission Regulation (EU) 748/2012, as amended by Commission Regulation (EU) n° 69/2014.

#### I. OSD Certification Basis

I.1 Reference Date for determining the applicable OSD requirements

For 269A, 269B: n/a

For 269C, 269C-1, 269D, 269D Configuration 'A': reserved

I.2 MMEL - Certification Basis

For 269A, 269B: n/a

For 269C, 269C-1, 269D, 269D Configuration 'A': reserved

I.3 Flight Crew Data - Certification Basis

For 269A, 269B: n/a

For 269C, 269C-1, 269D, 269D Configuration 'A': reserved

#### II. OSD Elements

II.1 MMEL

For 269A, 269B: n/a

For 269C, 269C-1, 269D, 269D Configuration 'A': reserved

II.2 Flight Crew Data

For 269A, 269B: n/a

For 269C, 269C-1, 269D, 269D Configuration 'A': reserved



#### SECTION: ADMINISTRATIVE

#### I. Acronyms and Abbreviations

Amdt.	Amendment	Max	Maximum
B.L.	Butt Line	min	Minute
CAA SE	Luftfartsverket	OSD	Operational Suitablity Data
	(Civil Aviation Administration Sverige)	p/n	Part Number
CR	(European) Commission Regulation	PA	Pressure Altitude
DA	Density Altitude	PWR	Power
ENAC	Ente Nazionale per l'Aviazione Civile	s/n	Serial Number
	(Italian Civil Aviation Authority)	SAC	Sikorsky Aircraft Corporation
FAA	Federal Aviation Administration	STA	Station
Hg	Mercury ( <b>h</b> ydrar <b>g</b> yrum)	TCDSN	Type Certificate Data Sheet for Noise
HMI	Handbook of Maintenance Instructions	TKOF	Take-Off
hp	Horse Power	VDoors 'OFF'	Doors 'OFF' Speed
LBA	Luftfahrt-Bundesamt	VFR	Visual Flight Rules
	(German Federal Aviation Office)	VNE	Never Exceed Speed

#### II. Type Certificate Holder Record

Type Certificate Holder	Period
Schweizer RSG LLC 3901 N Main St. Fort Worth, Texas 76106, U.S.A.	Since 25 January 2018
Sikorsky Aircraft Corporation 6900 Main Street Stratford, CT 06497-9129, U.S.A.	Until 26 January 2018
Schweizer Aircraft Corporation P.O. Box 147 Elmira, New York 14902, U.S.A.	Until 25 September 2011
Hughes Tool Company Aircraft Division Culver City, CA 90094, U.S.A.	Until 20 November 1986

#### III. Change Record

Issue	Date	Changes	TC issue
Issue 1	3 Jun 2015	Transfer of grandfathered FAA TCDS 4H12 to EASA format	Initial EASA Issue 3 June 2015
Issue 2	4 Jul 2019	Transfer to new type certificate holder; I.6, I.7 and I.8 of Section 3, 5 amended; all II.8: reference to TCDSN added; all II.9: reference to 'no OSD required' added	Re-issued 4 July 2019

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