European Aviation Safety Agency

EASA

TYPE-CERTIFICATE DATA SHEET

Number : P.176
Issue : 01
Date : 19 August 2010
Type : Aleš KŘEMEN
Z-42.6413 series propellers

Models
Z-42.6413

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I. General

1. Type/Models

Z-42.6413

2. Type Certificate Holder

Aleš KŘEMEN
Alšova 118
250 70 Odolena Voda - Dolínek
Czech Republic

Design Organisation Approval No.: EASA.AP250

3. Manufacturer

Aleš KŘEMEN
Alšova 118
250 70 Odolena Voda - Dolínek
Czech Republic

4. Date of Application

Z-42.6413
6.5.1970

5. Reference Date for determination of the applicable requirements

6 May 1970.

6. Certification Date

Z-42.6413
10.9.1970

Type certification of the Z-42.6413 series propeller model has been covered previously by Czech Republic Type Certificate No.70-07.
The Type Certificate has been transferred from “Moravan – Aeroplanes a.s., Czech Republic to Ales KREMEN on 19 August 2010.

II. Certification Basis

1. Airworthiness Standards

FAR Part 35-2 dated March 04, 1967

Note:
Application was made to CAA - Czech Republic (former Czechoslovakia) before EASA was established. The applicable airworthiness standards were established in accordance with the rule in Czech Republic (former Czechoslovakia) at the time of application.

III. Technical Characteristics

1. Type Design Definition

The Z-42.6413 propeller model covers the following design configuration. Design configuration is defined by a main assembly drawing and parts list.

Drawing No. 42-000-000, Rev.0, dated 02 August 2010 (*1)
Above mentioned drawing contains the parts list.

(*1) effective is the declared issue or a later approved revision.

2. Description

The Z-42.6413 series is a two blade, fixed pitch propeller constructed of wood composite structure. Leading edges of the propeller are protected against damage. The propeller surface is sprayed with a resistant polyurethane paint to increase lifetime.

3. Equipment

Spinner: 42-002-000

4. Dimensions

Propeller diameter: max. 2050 mm

5. Weight

Propeller weight: approx. 9,4 kg

6. Hub/Blade-Combinations

N/A (single piece propeller)

7. Control System

N/A (fixed pitch propeller)

8. Adaptation to Engine

Hub flanges as identified by a letter in the propeller designation (refer to note VI.3).

9. Direction of Rotation

Direction of rotation (viewed in flight direction) as identified by a letter-code in the propeller designation (refer to note VI.3)

IV. Operational Limits

1. Propeller Speed:

max. 2750 min^{-1}

2. Max.Take-Off Power:

132,5 kW

3. Max.Continous Power:

132,5 kW

4. Propeller Pitch Angle:

13° measured at reference station

V. Operating and Service Instructions

User´s manual	UM-03, Rev.0, dated 02 August 2010 [*]
Service Bulletins	as noted in the current List of Service Bulletins

[*] or later approved revision

VI. Notes

1. The suitability of the propeller for a given aircraft/engine-combination must be demonstrated within the scope of the type certification of the aircraft.

2. EASA Type Certificate and Type Certificate Data Sheet No.P.176 replace CAA - Czech Republic Type Certificate and Type Certificate Data Sheet No.70-07.

3. Propeller designation system

Z-42.6413( ) - ( ) - ( ) - ( ) - ( ) / ( )
1 2 3 4 5 6 7

1 – Propeller type

2 – variants of the Propeller model

3 – Propeller diameter in [cm]

4 – Letter code for the propeller sense of rotation / functioning
RT = right hand turning / tractor
LT = left hand turning / tractor
RP = right hand turning / pusher
LP = left hand turning / pusher

5 – Blade pitch in [°] defined to the shape tangent at 0,75 blade radius

6 – Type of propeller hub flange connecting dimensions – defined in type design

7 – Maximum permitted propeller speed in [min^{-1}].
