Civil Aviation Authority
United Kingdom

TYPE-CERTIFICATE DATA SHEET

UK.TC.A.00112

for

EA 300

Type Certificate Holder

EXTRA Flugzeugproduktions- und Vertriebs- GmbH
Schwarze Heide 21
46569, Hünxe
Germany

Model(s):	EA 300 EA 300/S EA 300/L EA 300/200 EA 300/SC EA 300/LT EA 300/LC EA 300/SX
Issue:	1
Date of issue:	02 July 2024

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Section 1 EA 300 (SALES DESIGNATION: EXTRA 300)

I. General

1. Type/ Model/ Variant	
1.1 Type	EA 300 (Sales designation EXTRA 300)
1.2 Model	-/-
1.3 Variant	-/-
2. Airworthiness Category	Normal, Aerobatic
3. Manufacturer (see Note 6)	EXTRA Flugzeugproduktions- und Vertriebs- GmbH Schwarze Heide 21 46569 Hünxe Germany
4. Type Certification Application Date	18 December 1986

Note: State of Design Authority certification application date for grandfathered products

5. State of Design Authority	Luftfahrt-Bundesamt (Germany)
6. State of Design Authority Type Certificate Date	16 May 1990
7. EASA Type Certification Date	17 July 2008

II. Certification Basis

1. Reference Date for determining the applicable requirements	Updated on 12 March 1993 (from initial 14 CFR eff. Feb. 1, 1965, incl. Amdt. 23-1 through 23-33)
2. Airworthiness Requirements	14 CFR eff. 1 February 1965, incl. Amdt. 23-1 through 23-34, effective 14 September 1987
3. Special Conditions	C-1, Ermüdungs-/Schadens-Toleranznachweis der Faserverbundstruktur (Fatique/Damage Tolerance Substantiation of Composite Structure) and C-4, Structural Design and Loads Criteria (LBA I 311-1086/93, dated 12-March-1993 & FAA Issue Paper C-1 and C-4, Project N° CA581EU) Smoke System (optional equipment) (LBA I 311-1086/96, dated 07-February-1996) Lufttüchtigkeitsforderungen für den Schleppflug (Airworthiness Requirement for Glider Towing) (LBA I 23-60/100, dated February-1971)
4. Exemptions	None
5. (Reserved) Deviations	None
6. Equivalent Safety Findings	None
7. Environmental Protection	ICAO, Annex 16, Volume 1

III. Technical Characteristic and Operating Limitations

1. Type Design Definition — EA-03102.1 Description and Operation of Aircraft and Systems (most current issue);

2. Description — Single engine, mid wing cantilever monoplane with reciprocating engine and fixed main gear in tail-wheel configuration; wing, empennage and landing gear in fibre-composite construction; fuselage and engine mount in conventional steel tube construction.

3. Equipment — Equipment List, refer to POH/AFM Doc. N° EA-03701, (See Note 4)

4. Dimensions

Span:	8.00 m	(26.25 ft)
Length:	7.12 m	(23.36 ft)
Height:	2.62 m	(8.60 ft)
Wing area:	10.72 m²	(115.39 sq.ft.)

5. Engine

5.1 Engine Model 1

5.1.1 Model	Lycoming	AEIO-540-L1B5	
5.1.2 Type Certificate	LBA No.	4535	
5.1.3 Limitations	Rated power at 2700 RPM		224 kW / 300 BHP
	Maximum RPM (take-off and max. continuous):		
	With propeller model 1: Aerobatic Category only		2700 RPM
	With propeller model 2: Normal and Aerobatic Category		2700 RPM
	(See Note 3)		

5.2 Engine Model 2

5.2.1 Model	Lycoming	AEIO-540-L1B5D	
5.2.2 Type Certificate	LBA No.	4535	
5.2.3 Limitations	Rated power at 2700 RPM		224 kW / 300 BHP
	Maximum RPM (Take-off and max. continuous):		
	With propeller model 1: Aerobatic Category only		2700 RPM
	With propeller model 2: Normal and Aerobatic Category		2700 RPM
	(See Note 3)		

6. Load factors

Normal category	+6 / -3
Aerobatic category	
Single Seat Operation / ACRO I	±10
Double Seat Operation / ACRO II	± 8

7. Propeller

7.1 Propeller Model 1

7.1.1	Model	MT Propeller	MTV-9-B-C/C200-15
7.1.2	Type Certificate	EASA.P.096	(replaced LBA No. 32.130/65)
7.1.3	Number of blades	3	
7.1.4	Diameter	2000 mm ± 0 mm	
7.1.5	Sense of Rotation	Right-hand tractor (viewed in direction of flight)	

7.2 Propeller Model 2

7.2.1	Model	MT Propeller	MTV-14-B-C/C190-17
7.2.2	Type Certificate	EASA.P.017	
7.2.3	Number of blades	4	
7.2.4	Diameter	1900 mm ± 0 mm	
7.2.5	Sense of Rotation	Right-hand tractor (viewed in direction of flight)	

8. Fluids

8.1	Fuel	100/100LL minimum grade aviation gasoline
8.2	Oil	Single or multi – viscosity aviation grade oils see latest issue of Textron Lycoming S.I. N° 1014
8.3	Coolant	None
8.4	Smoke Oil	Straight paraffin oil, kin. viscosity 30-50 cSt at 20°C (68°F), initial boiling point >330°C (626°F); For example: Fauth FC05, Texaco Canopus 13 or equivalent.

9. Fluid capacities

9.1 Fuel

9.1.1	Fuel – Standard	Total capacity	160 Litre	(42.3 US.gal)
		Usable capacity	158 Litre	(41.7 US.gal)
		Usable capacity for aerobatics	38 Litre	(10.0 US.gal)
9.1.2	Fuel – Long Range	Total capacity	194 Litre	(51.2 US.gal)
		Usable capacity	192 Litre	(50.7 US.gal)
		Usable capacity for aerobatics	38 Litre	(10.0 US.gal)
9.2	Oil	Max. sump capacity	15.1 Litre	(16 qts)
		Min. sump capacity aerobatic	11.3 Litre	(12 qts)
		Min. sump capacity normal	8.5 Litre	(9 qts)
9.3	Coolant system capacity	None		
9.4	Smoke Oil:		35 Litre	(9.2 US.gal)

10. Air Speeds

Design Manoeuvring Speed V_A:

Aerobatic category	158 KIAS
Normal category	140 KIAS

Max. Structural Cruising Speed V_{NO}:

Aerobatic category	158 KIAS
Normal category	140 KIAS

Never Exceed Speed V_{NE} : 220 KIAS

11. Flight Envelope

Max. operating altitude 4877 m (16000 ft)

12. Approved Operations Capability

Day-VFR

13. Maximum Masses

Take-off and Landing:

Normal category	950 kg	(2095 lbs)
Aerobatic category		
Single Seat Operation / ACRO I	820 kg	(1808 lbs)
Double Seat Operation / ACRO II	870 kg	(1918 lbs)

Empty:

Normal category		
Standard	745 kg	(1643 lbs)
Long Range	724 kg	(1596 lbs)
Aerobatic category		
Single Seat Operation / ACRO I	701 kg	(1546 lbs)
Double Seat Operation / ACRO II	665 kg	(1466 lbs)

14. Centre of Gravity Range

Forward limit (aft of datum):

at 820 kg (1808 lbs) or below	75.0 cm	(29.53”)
Normal category		
at 950 kg (2095 lbs)	78.0 cm	(30.71”)
Aerobatic category at 870 kg (1918 lbs)		
Double Seat Operation / ACRO II	76.5 cm	(30.12”)

Rear limit (aft of datum):

at 820 kg (1808 lbs) or below	89.8 cm	(35.35”)
Normal category		
at 950 kg (2095 lbs) or below	86.0 cm	(33.86”)
Aerobatic category at 870 kg (1918 lbs)		
Double Seat Operation / ACRO II	88.5 cm	(34.84”)

Straight line variation between mass limits.

15. Datum

Plane of Firewall

16. Control surface deflections

Aileron:	30°±2° upward,	30°±2° downward
Elevator:	25°±2° upward,	25°±2° downward
Rudder:	30°±2° left,	30°±2° right
Elevator trim tab:	40°±5° upward,	50°±5° downward

17. Levelling Means

Upper fuselage longeron

18. Minimum Flight Crew

1 Pilot (rear seat)

19. Maximum Passenger Seating Capacity

1 (front seat)

20. Baggage/Cargo Compartments

None

21. Wheels and Tyres

Main Wheel Tyre Size: 5.00-5 6ply

Tail Wheel Tyre Size:	Solid rubber 125/50-75 ZL or 6” (optional)

22. (Reserved)

IV. Operating and Service Instructions

1. Flight Manual

Flughandbuch (FHB)	Doc. No. EA-03701D
Pilot’s operating Handbook (POH) & Airplane Flight Manual (AFM)	Doc. No. EA-03701
Manuel de Vol (MdV)	Doc. No. EA-03701F

2. Maintenance Manual

Service Manual	Doc. No. EA-03702

3. Structural Repair Manual

Service Manual	Doc. No. EA-03702

4. Weight and Balance Manual

Flughandbuch (FHB)	Doc. No. EA-03701D
Pilot’s operating Handbook (POH) & Airplane Flight Manual (AFM)	Doc. No. EA-03701
Manuel de Vol (MdV)	Doc. No. EA-03701F

5. Illustrated Parts Catalogue

Parts Catalogue	Doc. No. EA-03703

V. Notes

1. This certification applies to Serial numbers V1, 03, 05, 06, 015 and on.
2. The use of an exhaust silencer system type ACC Columbia EA300-606500 is certified. The installation of the exhaust silencer system has to be in accordance with the Retrofit-Instruction UA-300-1-92. For service of the optional system the instructions of the appendix to the Service Manual EA 300 are obligatory.
3. A standard Certificate of Airworthiness can only be issued for an aircraft which is equipped with the 4-blade propeller MTV-14-B-C/C190-17 in combination with the exhaust silencer system type ACC Columbia EA300-606500 or EA300-606000. Otherwise a Certificate of Airworthiness can only be issued for aerial work.
4. For more certified optional equipment refer to EXTRA Doc. No. EA-03707, or AFM/POH latest revision. The applicable Retrofit-Instructions and supplements of the AFM are to be observed. Available: At manufacturer.
5. Structure is qualified up to 72°C (161.6°F). Structure temperatures (composite) above 72°C (161.6°F) are not permitted. Not to exceed this temperature limit, colour specification for composite structure of the manufacturer (document EA-03205.19) has to be complied with.
6. The address of the design and production organization until September 15th, 2003 is:
   Extra Flugzeugbau GmbH
   Flugplatz Dinslaken
   46569 Hünxe
   Germany

7. Model EA 300 serial numbers V1, 03, 05, 06 and 15 to 67 manufactured by Extra Flugzeugbau GmbH; serial numbers 1068 and on by Extra Flugzeugproduktions- und Vertriebs- GmbH (continuation of manufacture). See type certificate holder record (Administrative Section II).

Section 2 EA 300/S (Sales Designation: Extra 300S)

I. General

1. Type/ Model/ Variant	
1.1 Type	EA 300
1.2 Model	EA 300/S (Sales designation EXTRA 300S)
1.3 Variant	-/-
2. Airworthiness Category	Normal, Aerobatic
3. Manufacturer (see Note 6)	EXTRA Flugzeugproduktions- und Vertriebs- GmbH Schwarze Heide 21 46569 Hünxe Germany
4. Type Certification Application Date	17 September 1991

Note: State of Design Authority certification application date for grandfathered products

5. State of Design Authority	Luftfahrt-Bundesamt (Germany)
6. State of Design Authority Type Certificate Date	19 March 1993
7. EASA Type Certification Date	17 July 2008

II. Certification Basis

1. Reference Date for determining the applicable requirements	Updated on 12 March 1993 (from initial 14 CFR eff. Feb. 1, 1965, incl. Amdt. 23-1 through 23-33)
2. Airworthiness Requirements	14 CFR eff. 1 February 1965, incl. Amdt. 23-1 through 23-34, effective 14 September 1987
3. Special Conditions	C-1, Ermüdungs-/Schadens-Toleranznachweis der Faserverbundstruktur (Fatique/Damage Tolerance Substantiation of Composite Structure) and C-4, Structural Design and Loads Criteria (LBA I 311-1086/93, dated 12-March-1993 & FAA Issue Paper C-1 and C-4, Project N° CA581EU) Smoke System (optional equipment) (LBA I 311-1086/96, dated 07-February-1996) Lufttüchtigkeitsforderungen für den Schleppflug (Airworthiness Requirement for Glider Towing) (LBA I 23-60/100, dated February-1971)
4. Exemptions	None
5. (Reserved) Deviations	None
6. Equivalent Safety Findings	None
7. Environmental Protection	ICAO, Annex 16, Volume 1

III. Technical Characteristic and Operating Limitations

1. Type Design Definition

EA-04102.1 Description and Operation of Aircraft and Systems (most current issue);

2. Description

Single engine, mid wing cantilever monoplane with reciprocating engine and fixed main gear in tail-wheel configuration; wing, empennage and landing gear in fibre-composite construction; fuselage and engine mount in conventional steel tube construction.

3. Equipment

Equipment List, refer to POH/AFM Doc. N° EA-04701, (See Note 4)

4. Dimensions

Span:	7.50 m	(24.61 ft)
Length:	6.65 m	(21.82 ft)
Height:	2.62 m	(8.60 ft)
Wing area:	10.44 m²	(112.38 sq.ft.)

5. Engine

5.1 Engine Model 1

5.1.1 Model	Lycoming	AEIO-540-L1B5	
5.1.2 Type Certificate	LBA No.	4535	
5.1.3 Limitations	Rated power at 2700 RPM		224 kW / 300 BHP
	Maximum RPM (take-off and max. continuous):		
	With propeller model 1:		
	Aerobatic Category only		2700 RPM
	With propeller model 2:		
	Normal and Aerobatic Category		2700 RPM
	(See Note 3)		

5.2 Engine Model 2

5.2.1 Model	Lycoming	AEIO-540-L1B5D	
5.2.2 Type Certificate	LBA No.	4535	
5.2.3 Limitations	Rated power at 2700 RPM		224 kW / 300 BHP
	Maximum RPM (Take-off and max. continuous):		
	With propeller model 1:		
	Aerobatic Category only		2700 RPM
	With propeller model 2:		
	Normal and Aerobatic Category		2700 RPM
	(See Note 3)		

6. Load factors

Normal category	+6 / -3
Aerobatic category	±10

7. Propeller

7.1 Propeller Model 1

7.1.1 Model	MT Propeller	MTV-9-B-C/C200-15
7.1.2 Type Certificate	EASA.P.096	(replaced LBA No. 32.130/65)
7.1.3 Number of blades	3	
7.1.4 Diameter	2000 mm ± 0 mm	
7.1.5 Sense of Rotation	Right-hand tractor (viewed in direction of flight)	

7.2 Propeller Model 2

7.2.1 Model	MT Propeller	MTV-14-B-C/C190-17
7.2.2 Type Certificate	EASA.P.017	
7.2.3 Number of blades	4	
7.2.4 Diameter	1900 mm ± 0 mm	
7.2.5 Sense of Rotation	Right-hand tractor (viewed in direction of flight)	

8. Fluids

8.1 Fuel	100/100LL minimum grade aviation gasoline
8.2 Oil	Single or multi – viscosity aviation grade oils see latest issue of Textron Lycoming S.I. N° 1014
8.3 Coolant	None
8.4 Smoke Oil	Straight paraffin oil, kin. viscosity 30-50 cSt at 20°C (68°F), initial boiling point >330°C (626°F); For example: Fauth FC05, Texaco Canopus 13 or equivalent.

9. Fluid capacities

9.1 Fuel

9.1.1 Fuel – Standard	Total capacity	171 Litre	(45.1 US.gal)
	Usable capacity	169 Litre	(44.6 US.gal)
	Usable capacity for aerobatics	49 Litre	(12.9 US.gal)
9.1.2 Fuel – Long Range	Total capacity	205 Litre	(54.1 US.gal)
	Usable capacity	203 Litre	(53.5 US.gal)
	Usable capacity for aerobatics	49 Litre	(12.9 US.gal)
9.2 Oil	Max. sump capacity	15.1 Litre	(16 qts)
	Min. sump capacity aerobatic	11.3 Litre	(12 qts)
	Min. sump capacity normal	8.5 Litre	(9 qts)
9.3 Coolant system capacity	None		
9.4 Smoke Oil:		35 Litre	(9.2 US.gal)

10. Air Speeds

Design Manoeuvring Speed V_A:		
Aerobatic category		158 KIAS
Normal category		140 KIAS
Max. Structural Cruising Speed V_{NO}:		
Aerobatic category		158 KIAS
Normal category		140 KIAS
Never Exceed Speed V_{NE} :		220 KIAS

11. Flight Envelope

Max. operating altitude	4877 m	(16000 ft)

12. Approved Operations Capability

Day-VFR

13. Maximum Masses

Take-off and Landing:		
Normal category	920 kg	(2028 lbs)
Aerobatic category	820 kg	(1808 lbs)
Empty:		
Normal category		
Standard	711 kg	(1568 lbs)
Long Range	686 kg	(1513 lbs)
Aerobatic category	697 kg	(1537 lbs)

14. Centre of Gravity Range

Forward limit (aft of datum):		
920 kg (2028 lbs) or below	48.9 cm	(19.3”)
Rear limit (aft of datum):		
920 kg (2028 lbs) or below	71.4 cm	(28.1”)

15. Datum

Plane of Firewall

16. Control surface deflections

Aileron:	30°±2° upward;	30°±2° downward
Elevator:	25°±2° upward,	25°±2° downward
Rudder:	30°±2° left,	30°±2° right
Elevator trim tab:	40°±5° upward,	50°±5° downward

17. Levelling Means

Upper fuselage longeron

18. Minimum Flight Crew

1 Pilot

19. Maximum Passenger Seating Capacity

None

20. Baggage/Cargo Compartments

None

21. Wheels and Tyres

Main Wheel Tyre Size:	5.00-5 6ply
Tail Wheel Tyre Size:	Solid rubber 125/50-75 ZL or 6” (optional)

22. (Reserved)

IV. Operating and Service Instructions

1. Flight Manual

Flughandbuch (FHB)	Doc. No. EA-04701D
Pilot's operating Handbook (POH) & Airplane Flight Manual (AFM)	Doc. No. EA-04701

2. Maintenance Manual

Service Manual	Doc. No. EA-04702

3. Structural Repair Manual

Service Manual	Doc. No. EA-04702

4. Weight and Balance Manual

Flughandbuch (FHB)	Doc. No. EA-04701D
Pilot's operating Handbook (POH) & Airplane Flight Manual (AFM)	Doc. No. EA-04701

5. Illustrated Parts Catalogue

Parts Catalogue	Doc. No. EA-04703

V. Notes

1. This certification applies to Serial numbers 01 and on.
2. The use of an exhaust silencer system type ACC Columbia EA300-606500 is certified. The installation of the exhaust silencer system has to be in accordance with the Retrofit-Instruction UA-300-1-92. For service of the optional system the instructions of the appendix to the Service Manual EA 300 are obligatory.
3. A standard Certificate of Airworthiness can only be issued for an aircraft which is equipped with the 4-blade propeller MTV-14-B-C/C190-17 in combination with the exhaust silencer system type ACC Columbia EA300-606500 or EA300-606000. Otherwise a Certificate of Airworthiness can only be issued for aerial work.
4. For more certified optional equipment refer to EXTRA Doc. No. EA-04707, or AFM/POH latest revision. The applicable Retrofit-Instructions and supplements of the AFM are to be observed. Available: At manufacturer
5. Structure is qualified up to 72°C (161.6°F). Structure temperatures (composite) above 72°C (161.6°F) are not permitted. Not to exceed this temperature limit, colour specification for composite structure of the manufacturer (document EA-03205.19) has to be complied with.
6. The address of the design and production organization until September 15th, 2003 is:
   Extra Flugzeugbau GmbH
   Flugplatz Dinslaken
   46569 Hünxe
   Germany
7. Model EA 300/S serial numbers 1 to 31 manufactured by Extra Flugzeugbau GmbH; serial numbers 1032 and on by Extra Flugzeugproduktions- und Vertriebs- GmbH (continuation of manufacture). See type certificate holder record (Administrative Section II).

Section 3 EA300/L (Sales Designation: EXTRA 300L)

I. General

1. Type/ Model/ Variant	
1.1 Type	EA 300
1.2 Model	EA 300/L (Sales designation EXTRA 300L)
1.3 Variant	-/-
2. Airworthiness Category	Normal, Aerobatic
3. Manufacturer (see Note 6)	EXTRA Flugzeugproduktions- und Vertriebs- GmbH Schwarze Heide 21 46569 Hünxe Germany
4. Type Certification Application Date	02 February 1994

Note: State of Design Authority certification application date for grandfathered products

5. State of Design Authority	Luftfahrt-Bundesamt (Germany)
6. State of Design Authority Type Certificate Date	31 January 1995
7. EASA Type Certification Date	17 July 2008

II. Certification Basis

1. Reference Date for determining the applicable requirements	03 February 1994
2. Airworthiness Requirements	14 CFR eff. 1 February 1965, incl. Amdt. 23-1 through 23-34, effective 14 September 1987
3. Special Conditions	C-1, Ermüdungs-/Schadens-Toleranznachweis der Faserverbundstruktur (Fatique/Damage Tolerance Substantiation of Composite Structure) and C-4, Structural Design and Loads Criteria (LBA I 311-1086/93, dated 12-March-1993 & FAA Issue Paper C-1 and C-4, Project N° CA581EU) Smoke System (optional equipment) (LBA I 311-1086/96, dated 07-February-1996) Lufttüchtigkeitsforderungen für den Schleppflug (Airworthiness Requirement for Glider Towing) (LBA I 23-60/100, dated February-1971)
4. Exemptions	None
5. (Reserved) Deviations	None
6. Equivalent Safety Findings	None
7. Environmental Protection	ICAO, Annex 16, Volume 1

III. Technical Characteristic and Operating Limitations

1. Type Design Definition EA-06102.1 Description and Operation of Aircraft and Systems (most current issue);

2. Description Single engine, low wing cantilever monoplane with reciprocating engine and fixed main gear in tail-wheel configuration; wing, empennage and landing gear in fibre-composite construction; fuselage and engine mount in conventional steel tube construction.

3. Equipment Equipment List, refer to POH/AFM Doc. N° EA-06701, (See Note 4)

4. Dimensions

Span:	8.00 m	(26.25 ft)
Length:	6.96 m	(22.83 ft)
Height:	2.62 m	(8.60 ft)
Wing area:	10.84 m²	(116.68 sq.ft.)

5. Engine

5.1 Engine Model 1

5.1.1 Model	Lycoming	AEIO-540-L1B5	
5.1.2 Type Certificate	LBA No.	4535	
5.1.3 Limitations	Rated power at 2700 RPM		224 kW / 300 BHP
	Maximum RPM (take-off and max. continuous):		
	With propeller model 1: Aerobatic Category only		2700 RPM
	With propeller model 2: Normal and Aerobatic Category		2700 RPM
	(See Note 3)		

5.2 Engine Model 2

5.2.1 Model	Lycoming	AEIO-540-L1B5D	
5.2.2 Type Certificate	LBA No.	4535	
5.2.3 Limitations	Rated power at 2700 RPM		224 kW / 300 BHP
	Maximum RPM (Take-off and max. continuous):		
	With propeller model 1: Aerobatic Category only		2700 RPM
	With propeller model 2: Normal and Aerobatic Category		2700 RPM
	(See Note 3)		

5.3 Engine Model 3

5.3.1 Model	Lycoming	AEIO-580-B1A
5.3.2 Type Certificate	IM.E.027	

5.3.3 Limitations	Rated power at 2700 RPM	235 kW / 315 BHP
	Rated power at 2600 RPM	226 kW / 303 BHP
	Maximum RPM (Take-off and max. continuous):	
	With propeller model 2:	
	Normal and Aerobatic Category	2700 RPM
	With propeller model 3:	
	Normal and Aerobatic Category	2600 RPM
	Aerobatic Category only	2700 RPM
	(See Note 3 and Note 8)	

6. Load factors

Normal category	+6 / -3
Aerobatic category	
Single Seat Operation / ACRO I	±10
Double Seat Operation / ACRO II	± 8
Double Seat Operation / ACRO III	± 6

7. Propeller

7.1 Propeller Model 1

7.1.1 Model	MT Propeller	MTV-9-B-C/C200-15
7.1.2 Type Certificate	EASA.P.096	(replaced LBA No. 32.130/65)
7.1.3 Number of blades	3	
7.1.4 Diameter	2000 mm ± 0 mm	
7.1.5 Sense of Rotation	Right-hand tractor (viewed in direction of flight)	

7.2 Propeller Model 2

7.2.1 Model	MT Propeller	MTV-14-B-C/C190-17
7.2.2 Type Certificate	EASA.P.017	
7.2.3 Number of blades	4	
7.2.4 Diameter	1900 mm ± 0 mm	
7.2.5 Sense of Rotation	Right-hand tractor (viewed in direction of flight)	

7.3 Propeller Model 3

7.3.1 Model	MT Propeller	MTV-9-B-C/C198-25
7.3.2 Type Certificate	EASA.P.096	(replaced LBA No. 32.130/65)
7.3.3 Number of blades	3	
7.3.4 Diameter	1980 mm ± 5 mm	
7.3.5 Sense of Rotation	Right-hand tractor (viewed in direction of flight)	

8. Fluids

8.1 Fuel	100/100LL minimum grade aviation gasoline
8.2 Oil	Single or multi – viscosity aviation grade oils see latest issue of Textron Lycoming S.I. N° 1014
8.3 Coolant	None
8.4 Smoke Oil	Straight paraffin oil, kin. viscosity 30-50 cSt at 20°C (68°F), initial boiling point >330°C (626°F); For example: Fauth FC05, Texaco Canopus 13 or equivalent.

9. Fluid capacities

9.1 Fuel				
9.1.1	Fuel – Standard	Total capacity	171.0 Litre	(45.1 US.gal)
		Usable capacity	165.5 Litre	(43.7 US.gal)
		Usable capacity for aerobatics	45.5 Litre	(12.0 US.gal)
9.1.2	Fuel – Long Range	Total capacity	205.0 Litre	(54.1 US.gal)
		Usable capacity	199.5 Litre	(52.7 US.gal)
		Usable capacity for aerobatics	45.5 Litre	(12.0 US.gal)
9.1.3	Fuel – Raised Standard see Note 9	Total capacity	189 Litre	(49.9 US.gal)
		Usable capacity	187 Litre	(49.4 US.gal)
		Usable capacity for aerobatics	67 Litre	(17.7 US.gal)
9.2 Oil				
9.2.1	Oil – Engine Model 1 & 2	Max. sump capacity	15.1 Litre	(16 qts)
		Min. sump capacity aerobatic	11.3 Litre	(12 qts)
		Min. sump capacity normal	8.5 Litre	(9 qts)
9.2.2	Oil – Engine Model 3	Max. sump capacity	15.1 Litre	(16 qts)
		Min. sump capacity normal	8.5 Litre	(9 qts)
9.3 Coolant system capacity		None		
9.4 Smoke Oil:			31 Litre	(8.2 US.gal)

10. Air Speeds

Design Manoeuvring Speed V_A:	
Aerobatic category	158 KIAS
Normal category	140 KIAS
Max. Structural Cruising Speed V_{NO}:	
Aerobatic category	158 KIAS
Normal category	140 KIAS
Never Exceed Speed V_{NE} :	220 KIAS

11. Flight Envelope

Max. operating altitude	4877 m	(16000 ft)

12. Approved Operations Capability

Day-VFR

13. Maximum Masses

Take-off and Landing:		
Normal category	950 kg	(2095 lbs)
Aerobatic category		
Single Seat Operation / ACRO I	820 kg	(1808 lbs)
Double Seat Operation / ACRO II	870 kg	(1918 lbs)
Double Seat Operation / ACRO III	950 kg	(2095 lbs)

	Empty (with Engine Model 1 & 2):		
	Normal category		
	Standard	745 kg	(1643 lbs)
	Raised Standard	729 kg	(1607 lbs)
	Long Range	720 kg	(1588 lbs)
	Aerobatic category		
	Single Seat Operation / ACRO I	701 kg	(1546 lbs)
	ACRO I (raised Standard)	686 kg	(1513 lbs)
	Double Seat Operation / ACRO II	665 kg	(1466 lbs)
	Double Seat Operation / ACRO III	745 kg	(1643 lbs)
	Empty (with Engine Model 3):		
	Normal category		
	Standard	742 kg	(1636 lbs)
	Raised Standard	729 kg	(1607 lbs)
	Long Range	720 kg	(1588 lbs)
	Aerobatic category		
	Single Seat Operation / ACRO I	698 kg	(1540 lbs)
	ACRO I (raised Standard)	686 kg	(1513 lbs)
	Double Seat Operation / ACRO II	662 kg	(1460 lbs)
	Double Seat Operation / ACRO III	742 kg	(1636 lbs)
14. Centre of Gravity Range	Forward limit (aft of datum):		
	at 950 kg (2095 lbs) or below	67.1 cm	(29.4”)
	Rear limit (aft of datum):		
	at 950 kg (2095 lbs) or below	84.1 cm	(33.10”)
15. Datum	Plane of Firewall		
16. Control surface deflections	Aileron:	30°±2° upward;	30°±2° downward
	Elevator:	25°±2° upward,	25°±2° downward
	Rudder:	30°±2° left,	30°±2° right
	Elevator trim tab:	40°±5° upward,	50°±5° downward
17. Levelling Means	Upper fuselage longeron		
18. Minimum Flight Crew	1 Pilot (rear seat)		
19. Maximum Passenger Seating Capacity	1 (front seat)		
20. Baggage/Cargo Compartments	None		
21. Wheels and Tyres	Main Wheel Tyre Size:	5.00-5 6ply	
	Tail Wheel Tyre Size:	Solid rubber 125/50-75 ZL or 6” (optional)	
22. (Reserved)			

IV. Operating and Service Instructions

1. Flight Manual

Flughandbuch (FHB)	Doc. No. EA-06701D
Pilot's operating Handbook (POH) & Airplane Flight Manual (AFM)	Doc. No. EA-06701

2. Maintenance Manual

Service Manual	Doc. No. EA-06702

3. Structural Repair Manual

Service Manual	Doc. No. EA-06702

4. Weight and Balance Manual

Flughandbuch (FHB)	Doc. No. EA-06701D
Pilot's operating Handbook (POH) & Airplane Flight Manual (AFM)	Doc. No. EA-06701

5. Illustrated Parts Catalogue

Parts Catalogue	Doc. No. EA-06703

V. Notes

1. This certification applies to Serial numbers 01 and on.
2. The use of an exhaust silencer system type ACC Columbia EA300-606500 is certified. The installation of the exhaust silencer system has to be in accordance with the Retrofit-Instruction UA-300-1-92. For service of the optional system the instructions of the appendix to the Service Manual EA 300 are obligatory.
3. A standard Certificate of Airworthiness can only be issued for an aircraft which is equipped with
   - the 4-blade propeller MTV-14-B-C/C190-17 in combination with the exhaust silencer system type ACC Columbia EA300-606500 or EA300-606000 or
   - the 3-blade propeller MTV-9-B-C/C198-25 in combination with the exhaust silencer type ACC Columbia EA300-606000 and a reduced max. take-off engine rotational speed of 2600 RPM.

   Otherwise a Certificate of Airworthiness can only be issued for aerial work.
4. For more certified optional equipment refer to EXTRA Doc. No. EA-06707, or AFM/POH latest revision. The applicable Retrofit-Instructions and supplements of the AFM are to be observed. Available: At manufacturer
5. Structure is qualified up to 72°C (161.6°F). Structure temperatures (composite) above 72°C (161.6°F) are not permitted. Not to exceed this temperature limit, colour specification for composite structure of the manufacturer (document EA-03205.19) has to be complied with.
6. The address of the design and production organization until September 15th, 2003 is:
   Extra Flugzeugbau GmbH
   Flugplatz Dinslaken
   46569 Hünxe
   Germany
7. Model EA 300/L serial numbers 1 to 167 manufactured by Extra Flugzeugbau GmbH; serial numbers 168 to 170, 1171, 172, 173, 1174 an on by Extra Flugzeugproduktions- und Vertriebs- GmbH (continuation of manufacture). See type certificate holder record (Administrative Section II).

   Serial number 166 & 167 are under warranty of Extra Flugzeugproduktions- und Vertriebs- GmbH although manufactured by Extra Flugzeugbau GmbH.
8. The 3-blade propeller MTV-9-B-C/C198-25 is only approved in combination with the Lycoming engine AEIO-580-B1A specified in section C.III 5.3.1.
9. The raised-standard fuel system provides an increased fuel capacity of the center fuel tank approved for operation in the normal and aerobatic category delivered ex factory. It can not be combined with the increased fuel capacity of the wing fuel tank of the long range tank option specified in section C.III 9.1.2.

Section 4 EA 300/200 (Sales Designation: EXTRA 200)

I. General

1. Type/ Model/ Variant	
1.1 Type	EA 300
1.2 Model	EA 300/200 (Sales designation EXTRA 200)
1.3 Variant	-/-
2. Airworthiness Category	Normal, Aerobatic
3. Manufacturer (see Note 6)	EXTRA Flugzeugproduktions- und Vertriebs- GmbH Schwarze Heide 21 46569 Hünxe Germany
4. Type Certification Application Date	26 May 1995

Note: State of Design Authority certification application date for grandfathered products

5. State of Design Authority	Luftfahrt-Bundesamt (Germany)
6. State of Design Authority Type Certificate Date	12 August 1996
7. EASA Type Certification Date	17 July 2008

17 Certification Basis

1. Reference Date for determining the applicable requirements	26 May 1995
2. Airworthiness Requirements	14 CFR eff. 1 February 1965, incl. Amdt. 23-1 through 23-34, effective 14 September 1987
3. Special Conditions	C-1, Ermüdungs-/Schadens-Toleranznachweis der Faserverbundstruktur (Fatique/Damage Tolerance Substantiation of Composite Structure) and C-4, Structural Design and Loads Criteria (LBA I 311-1086/93, dated 12-March-1993 & FAA Issue Paper C-1 and C-4, Project N° CA581EU) Lufttüchtigkeitsforderungen für den Schleppflug (Airworthiness Requirement for Glider Towing) (LBA I 23-60/100, dated February-1971)
4. Exemptions	None
5. (Reserved) Deviations	None
6. Equivalent Safety Findings	EA-07406.1 issued 31 May 1999 and ACE-96-6, dated December 4, 1996, for paragraphs 23.963(e), 23.1337(b), and 23.1553
7. Environmental Protection	ICAO, Annex 16, Volume 1

18 Technical Characteristic and Operating Limitations

1. Type Design Definition	EA-07102.1 Description and Operation of Aircraft and Systems (most current issue);	
2. Description	Single engine, low wing cantilever monoplane with reciprocating engine and fixed main gear in tail-wheel configuration; wing, empennage and landing gear in fibre-composite construction; fuselage and engine mount in conventional steel tube construction.	
3. Equipment	Equipment List, refer to POH/AFM Doc. N° EA-07701, (See Note 3)	
4. Dimensions	Span:	7.50 m (24.61 ft)
	Length:	6.65 m (21.82 ft)
	Height:	2.62 m (8.60 ft)
	Wing area:	10.44 m² (112.38 sq.ft.)

5. Engine

5.1.1 Model	Lycoming	AEIO-360-A1E	
5.1.2 Type Certificate	LBA No.	4569	
5.1.3 Limitations	Rated power at 2700 RPM		149 kW / 200 BHP
	Rated power at 2500 RPM		138 kW / 185 BHP
	Maximum RPM (take-off)		2700 RPM
	Maximum RPM (max. continuous)		2500 RPM

6. Load factors	Normal category	+6 / -3
	Aerobatic category	
	Single Seat Operation / ACRO I	±10
	Double Seat Operation / ACRO II	± 8

7. Propeller

7.1.1 Model	MT Propeller	MTV-12-B-C/C183-17e
7.1.2 Type Certificate	EASA.P.013	
7.1.3 Number of blades	3	
7.1.4 Diameter	1830 mm ± 0 mm	
7.1.5 Sense of Rotation	Right-hand tractor (viewed in direction of flight)	

8. Fluids

8.1 Fuel	100/100LL minimum grade aviation gasoline	
8.2 Oil	Single or multi – viscosity aviation grade oils see latest issue of Textron Lycoming S.I. N° 1014	
8.3 Coolant	None	

9. Fluid capacities

9.1 Fuel				
	9.1.1 Fuel – Standard	Total capacity	122 Litre	(32.1 US.gal)
		Usable capacity	117 Litre	(30.8 US.gal)
		Usable capacity for aerobatics	32 Litre	(8.5 US.gal)
	9.1.2 Fuel – Long Range	Total capacity	190 Litre	(50.2 US.gal)
		Usable capacity	185 Litre	(48.9 US.gal)
		Usable capacity for aerobatics	32 Litre	(8.5 US.gal)
9.2 Oil		Max. sump capacity	7.6 Litre	(8 qts)
		Min. sump capacity aerobatic	5.7 Litre	(6 qts)
		Min. sump capacity normal	3.8 Litre	(4 qts)
9.3 Coolant system capacity		None		
9.4 Smoke Oil:			35 Litre	(9.2 US.gal)

10. Air Speeds	Design Manoeuvring Speed V_A:	
	Aerobatic category	154 KIAS / 158 KCAS
	Normal category	138 KIAS / 140 KCAS
	Max. Structural Cruising Speed V_{NO}:	
	Aerobatic category	154 KIAS / 158 KCAS
	Normal category	138 KIAS / 140 KCAS
	Never Exceed Speed V_{NE} :	217 KIAS / 220 KCAS
11. Flight Envelope	Max. operating altitude	4877 m (16000 ft)
12. Approved Operations Capability	Day-VFR	
13. Maximum Masses	Take-off and Landing:	
	Normal category	840 kg (1852 lbs)
	Aerobatic category	
	Single Seat Operation / ACRO I	700 kg (1543 lbs)
	Double Seat Operation / ACRO II	800 kg (1764 lbs)
	Empty:	
	Normal category	
	Standard	646 kg (1424 lbs)
	Long Range	621 kg (1369 lbs)
	Aerobatic category	
	Single Seat Operation / ACRO I	591 kg (1303 lbs)
	Double Seat Operation / ACRO II	606 kg (1336 lbs)
14. Centre of Gravity Range	Forward limit (aft of datum):	
	at 840 kg (1852 lbs) or below	73.2 cm (28.8”)
	Rear limit (aft of datum):	
	at 840 kg (1852 lbs) or below	89.1 cm (35.1”)
15. Datum	Plane of Firewall	

16. Control surface deflections	Aileron:	30°±2° upward;	30°±2° downward
	Elevator:	25°±2° upward,	25°±2° downward
	Rudder:	30°±2° left,	30°±2° right
	Elevator trim tab:	40°±5° upward,	50°±5° downward
17. Levelling Means	Upper fuselage longeron		
18. Minimum Flight Crew	1 Pilot (rear seat)		
19. Maximum Passenger Seating Capacity	1 (front seat)		
20. Baggage/Cargo Compartments	None		
21. Wheels and Tyres	Main Wheel Tyre Size:	5.00-5 6ply	
	Tail Wheel Tyre Size:	Solid rubber 125/50-75 ZL or 6” (optional)	
22. (Reserved)			

19 Operating and Service Instructions

1. Flight Manual

Flughandbuch (FHB)	Doc. No. EA-07701D
Pilot’s operating Handbook (POH) & Airplane Flight Manual (AFM)	Doc. No. EA-07701

2. Maintenance Manual

Service Manual	Doc. No. EA-07702

3. Structural Repair Manual

Service Manual	Doc. No. EA-07702

4. Weight and Balance Manual

Flughandbuch (FHB)	Doc. No. EA-07701D
Pilot’s operating Handbook (POH) & Airplane Flight Manual (AFM)	Doc. No. EA-07701

5. Illustrated Parts Catalogue

Parts Catalogue	None

20 Notes

1. This certification applies to Serial numbers 01 and on.
2. The fuel capacity of the wing tank and the maneuvering speed of Serial Number 01 and 02 differ from the model design as follows:
   - Maneuvering speed (Acrobatic category): V_A = 138 KIAS
   - Wing- and acro tank:
     Total fuel capacity 156 L
     Usable fuel capacity 151 L
   - Operator's instruction:
     Supplement Airplane Flight Manual / POH EXTRA 300/200 Doc. No. EA–07701D.2
3. For more certified optional equipment refer to EXTRA Doc. No. EA-07707, or AFM/POH latest revision. The applicable Retrofit-Instructions and supplements of the AFM are to be observed. Available: At manufacturer
4. Structure is qualified up to 72°C (161.6°F). Structure temperatures (composite) above 72°C (161.6°F) are not permitted. Not to exceed this temperature limit, colour specification for composite structure of the manufacturer (document EA-03205.19) has to be complied with.
5. Export to USA:
   The airplanes to be registered in USA must comply with the provisions of "Modification Instruction for conformity to the Type Certificate A67EU EA 300/200" (EXTRA Doc. N° UA-300-1-96).
6. The address of the design and production organization until September 15th, 2003 is:
   Extra Flugzeugbau GmbH
   Flugplatz Dinslaken
   46569 Hünxe
   Germany
7. Model EA 300/200 serial numbers 1 to 31 manufactured by Extra Flugzeugbau GmbH; serial numbers 1032 and on by Extra Flugzeugproduktions- und Vertriebs- GmbH (continuation of manufacture). See type certificate holder record (Administrative Section II).

Section 5 EA 300/SC (Sales Designation: EXTRA 330SC)

I. General

1. Type/ Model/ Variant	
1.1 Type	EA 300
1.2 Model	EA 300/SC (Sales designation EXTRA 330SC)
1.3 Variant	-/-
2. Airworthiness Category	Normal, Aerobatic
3. Manufacturer	EXTRA Flugzeugproduktions- und Vertriebs- GmbH Schwarze Heide 21 46569 Hünxe Germany
4. EASA Type Certification Application Date	23 November 2007

Note: State of Design Authority certification application date for grandfathered products

5. State of Design Authority	-/-
6. State of Design Authority Type Certificate Date	-/-
7. EASA Type Certification Date	17 July 2008

II. Certification Basis

1. Reference Date for determining the applicable requirements	31 January 2008 (initial type board meeting at EASA)
2. Airworthiness Requirements	14 CFR eff. 1 February 1965, incl. Amdt. 23-1 through 23-34, effective 14 September 1987
3. Special Conditions	C-1, Ermüdungs-/Schadens-Toleranznachweis der Faserverbundstruktur (Fatique/Damage Tolerance Substantiation of Composite Structure) and C-4, Structural Design and Loads Criteria (LBA I 311-1086/93, dated 12-March-1993 & FAA Issue Paper C-1 and C-4, Project N° CA581EU) Smoke System (optional equipment) (LBA I 311-1086/96, dated 07-February-1996)
4. Exemptions	None
5. (Reserved) Deviations	None
6. Equivalent Safety Findings	a) Static longitudinal stability §§ 23.171, 23.173 & 23.175 b) Stall warning §23.207
7. Environmental Protection	ICAO, Annex 16, Volume 1, Fourth Edition, Amdt. 8

III. Technical Characteristic and Operating Limitations

1. Type Design Definition

EA 300/SC , EA-0C102.1 Description of differences to EA 300/S type design (most current issue);
based on
EA 300/S, EA-04102.1 Description and Operation of Aircraft and Systems (most current issue)

2. Description

Single engine, low wing cantilever monoplane with reciprocating engine and fixed main gear in tail-wheel configuration; wing, empennage and landing gear in fibre-composite construction; fuselage and engine mount in conventional steel tube construction.

3. Equipment

Equipment List, refer to POH/AFM Doc. N° EA-0C701,
(See Note 3)

4. Dimensions

Span:	7.50 m	(24.61 ft)
Length:	6.88 m	(22.55 ft)
Height:	2.55 m	(8.36 ft)
Wing area:	9.81 m²	(105.59 sq.ft.)

5. Engine

5.1 Model: Lycoming AEIO-580-B1A

5.2 Type Certificate: IM.E.027

5.3 Limitations:

Rated power at 2700 RPM: 235 kW/ 315 BHP
Rated power at 2600 RPM: 226 kW/ 303 BHP

Maximum RPM (take-off and max. continuous):

With propeller model 1 and model 2:

Normal & Aerobatic Category: 2600 RPM
Aerobatic Category only: 2700 RPM

(see Note 2)

6. Load factors

Normal category	+6 / -3
Aerobatic category (780 kg and below)	±10

7. Propeller

7.1 Propeller Model 1

7.1.1 Model: MT Propeller MTV-9-B-C/C198-25

7.1.2 Type Certificate: EASA.P.096 (replaced LBA No. 32.130/65)

7.1.3 Number of blades: 3

7.1.4 Diameter: 1980 mm ± 5 mm

7.1.5 Sense of Rotation: Right-hand tractor (viewed in direction of flight)

7.2 Propeller Model 2

7.2.1	Model	MT Propeller MTV-14-B-C/C190-130
7.2.2	Type Certificate	EASA.P.017
7.2.3	Number of blades	4
7.2.4	Diameter	1900 mm ± 5 mm
7.2.5	Sense of Rotation	Right-hand tractor (viewed in direction of flight)

8. Fluids

8.1	Fuel	100/100LL minimum grade aviation gasoline
8.2	Oil	Single or multi – viscosity aviation grade oils see latest issue of Textron Lycoming S.I. N° 1014
8.3	Coolant	None
8.4	Smoke Oil	Straight paraffin oil, kin. viscosity 30-50 cSt at 20°C (68°F), initial boiling point >330°C (626°F); For example: Fauth FC05, Texaco Canopus 13 or equivalent.

9. Fluid capacities

9.1	Fuel	Total capacity	224 Litre	(59.2 US.gal)
		Usable capacity	221 Litre	(58.4 US.gal)
		Usable capacity for aerobatics	101 Litre	(26.7 US.gal)
9.2	Oil	Max. sump capacity	15.1 Litre	(16 qts)
		Min. sump capacity normal	8.5 Litre	(9 qts)
9.3	Coolant system capacity	None		
9.4	Smoke Oil:		23 Litre	(6.1 US.gal)

10. Air Speeds

Design Manoeuvring Speed V_A:	
Aerobatic category	154 KIAS / 158 KCAS
Normal category	138 KIAS / 140 KCAS
Max. Structural Cruising Speed V_{NO}:	
Aerobatic category	154 KIAS / 158 KCAS
Normal category	138 KIAS / 140 KCAS
Never Exceed Speed V_{NE} :	219 KIAS / 220 KCAS

11. Flight Envelope

Max. operating altitude	3048 m	(10000 ft)

12. Approved Operations Capability

Day-VFR

13. Maximum Masses

Take-off and Landing:		
Normal category	870 kg	(1918 lbs)
Aerobatic category	780 kg	(1720 lbs)
Empty:		
Normal category	624 kg	(1377 lbs)
Aerobatic category	620 kg	(1367 lbs)

14. Centre of Gravity Range

Forward limit (aft of datum):

Normal category		
at 870 kg (1918 lbs)	54,5 cm	(21.5”)
at 820 kg (1808 lbs) or below	53.7 cm	(21.1”)
Aerobatic category		
at 780 kg (1720 lbs)	53.7 cm	(21.1”)

Rear limit (aft of datum):

Normal category		
at 870 kg (1918 lbs)	62.6 cm	(24.6”)
at 780 kg (1720 lbs) or below	66.8 cm	(26.3”)
Aerobatic category		
at 780 kg (1720 lbs) or below	66.8 cm	(26.3”)

Straight line variation between mass limits.

15. Datum

Plane of Firewall

16. Control surface deflections

Aileron:	30°±2° upward;	30°±2° downward
Elevator:	25°±1° upward,	25°±1° downward
Rudder:	30°-2° left,	30°-2° right
Elevator trim tab:	32°±2° upward,	32°±2° downward

17. Levelling Means

Upper fuselage longeron

18. Minimum Flight Crew

1 Pilot

19. Maximum Passenger Seating Capacity

None

20. Baggage/Cargo Compartments

None

21. Wheels and Tyres

Main Wheel Tyre Size:	5.00-5 6ply
Tail Wheel Tyre Size:	Solid rubber 125/50-75 ZL or 6” (optional)

22. (Reserved)

IV. Operating and Service Instructions

1. Flight Manual

   Pilot’s operating Handbook (POH) & Airplane Flight Manual (AFM) — Doc. No. EA-0C701

2. Maintenance Manual

   Service Manual — Doc. No. EA-0C702

3. Structural Repair Manual

   Service Manual — Doc. No. EA-0C702

4. Weight and Balance Manual

   Pilot’s operating Handbook (POH) & Airplane Flight Manual (AFM) — Doc. No. EA-0C701

5. Illustrated Parts Catalogue

   Parts Catalogue — Doc. No. EA-0C703

V. Notes

1. This certification applies to Serial Numbers SC003 and on.
2. A standard Certificate of Airworthiness can only be issued for an aircraft which is equipped with:
   - the 3-blade propeller MTV-9-B-C/C198-25 in combination with the exhaust silencer system type ACC Columbia EA300-606000 and a reduced max. takeoff engine rotational speed of 2600 RPM.
   - the 4-blade propeller MTV-14-B-C/C190-130 in combination with theexhaust silencer system type ACC Columbia EA300-606000 and a reduced max. take-off engine rotational speed of 2600 RPM.

   Otherwise a Certificate of Airworthiness can only be issued for aerial work.
3. For more certified optional equipment refer to approved AFM/POH Supplements latest revision.
4. Structure is qualified up to 72°C (161.6°F). Structure temperatures (composite) above 72°C (161.6°F) are not permitted. Not to exceed this temperature limit, colour specification for composite structure of the manufacturer (document EA- 03205.19) has to be complied with.

Section 6 EA 300/LT (Sales Designation: EXTRA 330LT)

I. General

1. Type/ Model/ Variant	
1.1 Type	EA 300
1.2 Model	EA 300/LT (Sales designation EXTRA 330LT)
1.3 Variant	-/-
2. Airworthiness Category	Normal, Aerobatic
3. Manufacturer	EXTRA Flugzeugproduktions- und Vertriebs- GmbH Schwarze Heide 21 46569 Hünxe Germany
4. EASA Type Certification Application Date	22 January 2009

Note: State of Design Authority certification application date for grandfathered products

5. State of Design Authority	-/-
6. State of Design Authority Type Certificate Date	-/-
7. EASA Type Certification Date	31 May 2010

II. Certification Basis

1. Reference Date for determining the applicable requirements	28 April 2009 (initial type board meeting at EASA)
2. Airworthiness Requirements	14 CFR eff. 1 February 1965, incl. Amdt. 23-1 through 23-34, effective 14 September 1987
3. Special Conditions	C-1, Ermüdungs-/Schadens-Toleranznachweis der Faserverbundstruktur (Fatique/Damage Tolerance Substantiation of Composite Structure) and C-4, Structural Design and Loads Criteria (LBA I 311-1086/93, dated 12-March-1993 & FAA Issue Paper C-1 and C-4, Project N° CA581EU) Smoke System (optional equipment) (LBA I 311-1086/96, dated 07-February-1996) Lufttüchtigkeitsforderungen für den Schleppflug (Airworthiness Requirement for Glider Towing) (LBA I 23-60/100, dated February-1971)
4. Exemptions	None
5. (Reserved) Deviations	None
6. Equivalent Safety Findings	Static longitudinal stability §§ 23.171, 23.173, 23.175 & 23.177
7. Environmental Protection	ICAO, Annex 16, Volume 1, Fourth Edition, Amdt. 8

III. Technical Characteristic and Operating Limitations

1. Type Design Definition	EA 300/LT, EA-0D102.1 Description of differences to EA 300/L type design (most current issue); based on: EA 300/L, EA-06102.1 Description and Operation of Aircraft and Systems (most current issue)	
2. Description	Single engine, low wing cantilever monoplane with reciprocating engine and fixed main gear in tail-wheel configuration; wing, empennage and landing gear in fibre-composite construction; fuselage and engine mount in conventional steel tube construction.	
3. Equipment	Equipment List, refer to POH/AFM Doc. N° EA-0D701, (See Note 3)	
4. Dimensions	Span:	8.00 m (26.25 ft)
	Length:	7.01 m (23.00 ft)
	Height:	2.62 m (8.60 ft)
	Wing area:	10.84 m² (116.68 sq.ft.)
5. Engine		
5.1 Model	Lycoming AEIO-580-B1A	
5.2 Type Certificate	IM.E.027	
5.3 Limitations	Rated power at 2700 RPM	235 kW / 315 BHP
	Maximum RPM (take-off and max. continuous): Normal and Aerobatic Category	2700 RPM
6. Load factors	Normal category	+6 / -3
	Aerobatic category	
	Single Seat Operation / ACRO I	±10
	Double Seat Operation / ACRO II	± 8
	Double Seat Operation / ACRO III	± 6
7. Propeller		
7.1 Model	MT Propeller MTV-9-B-C/C198-25	
7.2 Type Certificate	EASA.P.096 (replaced LBA No. 32.130/65)	
7.3 Number of blades	3	
7.4 Diameter	1980 mm ± 5 mm	
7.5 Sense of Rotation	Right-hand tractor (viewed in direction of flight)	
8. Fluids		
8.1 Fuel	100/100LL minimum grade aviation gasoline	
8.2 Oil	Single or multi – viscosity aviation grade oils see latest issue of Textron Lycoming S.I. N° 1014	
8.3 Coolant	None	
8.4 Smoke Oil	Straight paraffin oil, kin. viscosity 30-50 cSt at 20°C (68°F), initial boiling point >330°C (626°F); For example: Fauth FC05, Texaco Canopus 13 or equivalent.	

9. Fluid capacities

9.1 Fuel	Total capacity	221 Litre	(58.4 US.gal)
	Usable capacity	209 Litre	(55.2 US.gal)
	Usable capacity for aerobatics	67 Litre	(17.7 US.gal)
9.2 Oil	Max. sump capacity	15.1 Litre	(16 qts)
	Min. sump capacity normal	8.5 Litre	(9 qts)
9.3 Coolant system capacity	None		
9.4 Smoke Oil:		31 Litre	(8.2 US.gal)

10. Air Speeds

Design Manoeuvring Speed V_A:

Aerobatic category	160 KIAS / 158 KCAS
Normal category	143 KIAS / 140 KCAS

Max. Structural Cruising Speed V_{NO}:

Aerobatic category	160 KIAS / 158 KCAS
Normal category	143 KIAS / 140 KCAS

Never Exceed Speed V_{NE} : 221 KIAS / 220 KCAS

11. Flight Envelope

Max. operating altitude 3048 m (10000 ft)

12. Approved Operations Capability

Day-VFR

13. Maximum Masses

Take-off and Landing:

Normal category	950 kg	(2095 lbs)
Aerobatic category		
Single Seat Operation / ACRO I	820 kg	(1808 lbs)
Double Seat Operation / ACRO II	870 kg	(1918 lbs)
Double Seat Operation / ACRO III	950 kg	(2095 lbs)

Empty:

Normal category	723 kg	(1594 lbs)
Aerobatic category		
Single Seat Operation / ACRO I	686 kg	(1513 lbs)
Double Seat Operation / ACRO II	662 kg	(1460 lbs)
Double Seat Operation / ACRO III	742 kg	(1636 lbs)

14. Centre of Gravity Range

Forward limit (aft of datum):

at 820 kg (1808 lbs) or below	70.7 cm	(27.8”)
at 870 kg (1918 lbs)	71.6 cm	(28.2”)
at 950 kg (2095 lbs)	73.0 cm	(28.7”)

Rear limit (aft of datum):

at 915 kg (2018 lbs) or below	88.0 cm	(34.6”)
at 950 kg (2095 lbs)	84.1 cm	(33.1”)

Straight line variation between mass limits.

15. Datum

Plane of Firewall

16. Control surface deflections

Aileron:	30°±2° upward;	20°±2° downward
Elevator:	25°±2° upward,	25°±2° downward
Rudder:	30°±2° left,	30°±2° right
Elevator trim tab:	35°±2° upward,	27°±2° downward

17. Levelling Means

Upper fuselage longeron

18. Minimum Flight Crew

1 Pilot (rear seat)

19. Maximum Passenger Seating Capacity	1 (front seat)		
20. Baggage/Cargo Compartments	1 baggage compartment within the upper aft fuselage section behind the rear seat. The baggage compartment must be empty for aerobatics.		
	Max. baggage mass:	10 kg	(22 lbs)
	C.G. (aft of datum):	331 cm	(130.1”)
21. Wheels and Tyres	Main Wheel Tyre Size:	5.00-5 6ply	
	Tail Wheel Tyre Size:	Solid rubber 125/50-75 ZL or 6” (optional)	
22. (Reserved)			

IV. Operating and Service Instructions

1. Flight Manual	
Pilot’s operating Handbook (POH) & Airplane Flight Manual (AFM)	Doc. No. EA-0D701
2. Maintenance Manual	
Service Manual	Doc. No. EA-0D702
3. Structural Repair Manual	
Service Manual	Doc. No. EA-0D702
4. Weight and Balance Manual	
Pilot’s operating Handbook (POH) & Airplane Flight Manual (AFM)	Doc. No. EA-0D701
5. Illustrated Parts Catalogue	
Parts Catalogue	None

V. Notes

1. This certification applies to Serial numbers LT001 and on.
2. res.
3. For more certified optional equipment refer to approved AFM/POH Supplements latest revision.
4. Structure is qualified up to 72°C (161.6°F). Structure temperatures (composite) above 72°C (161.6°F) are not permitted. Not to exceed this temperature limit, colour specification for composite structure of the manufacturer (document EA-03205.19) has to be complied with.

Section 7 EA 300/LC (Sales Designation: EXTRA 330LX)

I. General

1.	**Type/ Model/ Variant**	
	1.1 Type	EA 300
	1.2 Model	EA 300/LC (Sales designation: EXTRA 330LX)
	1.3 Variant	-/-
2.	**Airworthiness Category**	Normal, Aerobatic
3.	**Manufacturer**	EXTRA Flugzeugproduktions- und Vertriebs- GmbH Schwarze Heide 21 46569 Hünxe Germany
4.	**EASA Type Certification Application Date**	27 October 2009; amended on 02 December 2009

Note: State of Design Authority certification application date for grandfathered products

5.	**State of Design Authority**	-/-
6.	**State of Design Authority Type Certificate Date**	-/-
7.	**EASA Type Certification Date**	08 April 2011

II. Certification Basis

1.	**Reference Date for determining the applicable requirements**	21 January 2010 (initial type board meeting at EASA)
2.	**Airworthiness Requirements**	14 CFR eff. 1 February 1965, incl. Amdt. 23-1 through 23-34, effective 14 September 1987
3.	**Special Conditions**	C-1, Ermüdungs-/Schadens-Toleranznachweis der Faserverbundstruktur (Fatique/Damage Tolerance Substantiation of Composite Structure), and C-4, Structural Design and Loads Criteria (LBA I 311-1086/93, dated 12-March-1993 & FAA Issue Paper C-1 and C-4, Project N° CA581EU) Smoke System (optional equipment) (LBA I 311-1086/96, dated 07-February-1996) Lufttüchtigkeitsforderungen für den Schleppflug (Airworthiness Requirement for Glider Towing) (LBA I 23-60/100, dated February-1971)
4.	**Exemptions**	None
5.	**(Reserved) Deviations**	None
6.	**Equivalent Safety Findings**	a) Static longitudinal stability §§ 23.171, 23.173 & 23.175 b) Stall warning § 23.207
7.	**Environmental Protection**	ICAO, Annex 16, Volume 1, Fourth Edition, Amdt. 8

III. Technical Characteristic and Operating Limitations

1. Type Design Definition

EA 300/LC, EA-0E102.1 Description of differences to EA 300/L type design (most current issue);
based on:
EA 300/L, EA-06102.1 Description and Operation of Aircraft and Systems (most current issue)

2. Description

Single engine, low wing cantilever monoplane with reciprocating engine and fixed main gear in tail-wheel configuration; wing, empennage and landing gear in fibre-composite construction; fuselage and engine mount in conventional steel tube construction.

3. Equipment

Equipment List, refer to POH/AFM Doc. N° EA-0E701,
(See Note 3)

4. Dimensions

Span:	8.00 m	(26.25 ft)
Length:	7.20 m	(23.62 ft)
Height:	2.62 m	(8.60 ft)
Wing area:	10.72 m²	(115.39 sq.ft.)

5. Engine

5.1 Model: Lycoming AEIO-580-B1A

5.2 Type Certificate: IM.E.027

5.3 Limitations:

Rated power at 2700 RPM: 235 kW/ 315 BHP
Rated power at 2600 RPM: 226 kW/ 303 BHP

Maximum RPM (take-off and max. continuous):

With propeller model 1 and model 2:
Normal & Aerobatic Category: 2600 RPM
Aerobatic Category only: 2700 RPM

(see Note 2)

6. Load factors

Normal category	+6 / -3
Aerobatic category	
Single Seat Operation / ACRO I	±10
Double Seat Operation / ACRO II	± 8
Double Seat Operation / ACRO III	± 6

7. Propeller

7.1 Propeller Model 1

7.1.1 Model: MT Propeller MTV-9-B-C/C198-25

7.1.2 Type Certificate: EASA.P.096 (replaced LBA No. 32.130/65)

7.1.3 Number of blades: 3

7.1.4 Diameter: 1980 mm ± 5 mm

7.1.5 Sense of Rotation: Right-hand tractor (viewed in direction of flight)

7.2 Propeller Model 2

7.2.1	Model	MT Propeller MTV-14-B-C/C190-130
7.2.2	Type Certificate	EASA.P.017
7.2.3	Number of blades	4
7.2.4	Diameter	1900 mm ± 5 mm
7.2.5	Sense of Rotation	Right-hand tractor (viewed in direction of flight)

8. Fluids

8.1	Fuel	100/100LL minimum grade aviation gasoline
8.2	Oil	Single or multi – viscosity aviation grade oils see latest issue of Textron Lycoming S.I. N° 1014
8.3	Coolant	None
8.4	Smoke Oil	Straight paraffin oil, kin. viscosity 30-50 cSt at 20°C (68°F), initial boiling point >330°C (626°F); For example: Fauth FC05, Texaco Canopus 13 or equivalent.

9. Fluid capacities

9.1	Fuel (Standard)	Total capacity	189 Litre	(49.9 US.gal)
		Usable capacity	187 Litre	(49.4 US.gal)
		Usable capacity for aerobatics	67 Litre	(17.7 US.gal)
9.2	Oil	Max. sump capacity	15.1 Litre	(16 qts)
		Min. sump capacity normal	8.5 Litre	(9 qts)
9.3	Coolant system capacity	None		
9.4	Smoke Oil:		31 Litre	(8.2 US.gal)

10. Air Speeds

Design Manoeuvring Speed V_A:	
Aerobatic category	154 KIAS / 158 KCAS
Normal category	138 KIAS / 140 KCAS
Max. Structural Cruising Speed V_{NO}:	
Aerobatic category	154 KIAS / 158 KCAS
Normal category	138 KIAS / 140 KCAS
Never Exceed Speed V_{NE} :	219 KIAS / 220 KCAS

11. Flight Envelope

Max. operating altitude 3048 m (10000 ft)

12. Approved Operations Capability

Day-VFR

13. Maximum Masses

Take-off and Landing:		
Normal category	950 kg	(2095 lbs)
Aerobatic category		
Single Seat Operation / ACRO I	820 kg	(1808 lbs)
Double Seat Operation / ACRO II	870 kg	(1918 lbs)
Double Seat Operation / ACRO III	950 kg	(2095 lbs)
Empty:		
Normal category	738 kg	(1627 lbs)
Aerobatic category		
Single Seat Operation / ACRO I	686 kg	(1513 lbs)
Double Seat Operation / ACRO II	662 kg	(1460 lbs)
Double Seat Operation / ACRO III	742 kg	(1636 lbs)

14. Centre of Gravity Range

Forward limit (aft of datum):		
at 950 kg (2095 lbs) or below	67.1 cm	(26.4”)
Rear limit (aft of datum):		
at 950 kg (2095 lbs) or below	84.1 cm	(33.1”)

15. Datum

Plane of Firewall

16. Control surface deflections

Aileron:	30°±2° upward;	30°±2° downward
Elevator:	25°±2° upward,	25°-2° downward
Rudder:	30°±2° left,	30°±2° right
Elevator trim tab:	35°±2° upward,	27°±2° downward

17. Levelling Means

Upper fuselage longeron

18. Minimum Flight Crew

1 Pilot (rear seat)

19. Maximum Passenger Seating Capacity

1 (front seat)

20. Baggage/Cargo Compartments

None

21. Wheels and Tyres

Main Wheel Tyre Size:	5.00-5 6ply
Tail Wheel Tyre Size:	Solid rubber 125/50-75 ZL or 6” (optional)

22. (Reserved)

IV. Operating and Service Instructions

1. Flight Manual

Pilot's operating Handbook (POH) & Airplane Flight Manual (AFM)	Doc. No. EA-0E701

2. Maintenance Manual

Service Manual	Doc. No. EA-0E702

3. Structural Repair Manual

Service Manual	Doc. No. EA-0E702

4. Weight and Balance Manual

Pilot's operating Handbook (POH) & Airplane Flight Manual (AFM)	Doc. No. EA-0E701

5. Illustrated Parts Catalogue

Parts Catalogue	None

V. Notes

1. This certification applies to Serial numbers LC001 and on.
2. A standard Certificate of Airworthiness can only be issued for an aircraft which is equipped with:
   - the 3-blade propeller MTV-9-B-C/C198-25 in combination with the exhaust silencer system type ACC Columbia EA300-606000 and a reduced max. take-off engine rotational speed of 2600 RPM.
   - the 4-blade propeller MTV-14-B-C/C190-130 in combination with the exhaust silencer system type ACC Columbia EA300-606000 and a reduced max. take-off engine rotational speed of 2600 RPM.

   Otherwise a Certificate of Airworthiness can only be issued for aerial work.
3. For more certified optional equipment refer to approved AFM/POH Supplements latest revision.
4. Structure is qualified up to 72°C (161.6°F). Structure temperatures (composite) above 72°C (161.6°F) are not permitted. Not to exceed this temperature limit, colour specification for composite structure of the manufacturer (document EA-03205.19) has to be complied with.

Section 8 EA 300/SX (Sales Designation: EXTRA 330SX)

I. General

1. Type/ Model/ Variant		
	1.1 Type	EA 300
	1.2 Model	EA 300/SX (Sales designation EXTRA 330SX)
	1.3 Variant	-/-
2. Airworthiness Category		Normal, Aerobatic
3. Manufacturer		EXTRA Flugzeugproduktions- und Vertriebs- GmbH Schwarze Heide 21 46569 Hünxe Germany
4. EASA Type Certification Application Date		02 December 2022

Note: State of Design Authority certification application date for grandfathered products

5. State of Design Authority	-/-
6. State of Design Authority Type Certificate Date	-/-
7. EASA Type Certification Date	29 February 2024

II. Certification Basis

1. Reference Date for determining the applicable requirements	02 December 2022
2. Airworthiness Requirements	14 CFR eff. 1 February 1965, incl. Amdt. 23-1 through 23-34, effective 14 September 1987
3. Special Conditions	C-1, Ermüdungs-/Schadens-Toleranznachweis der Faserverbundstruktur (Fatique/Damage Tolerance Substantiation of Composite Structure) and C-4, Structural Design and Loads Criteria (LBA I 311-1086/93, dated 12-March-1993 & FAA Issue Paper C-1 and C-4, Project N° CA581EU) Smoke System (optional equipment) (LBA I 311-1086/96, dated 07-February-1996)
4. Exemptions	None
5. (Reserved) Deviations	None
6. Equivalent Safety Findings	a) ESF-B23-0207-01 Stall warning §23.207 b) ESF-D23-777/781 Position and Shape of Engine Controls §23.777(d) & §23.781(b)
7. Environmental Protection	ICAO, Annex 16, Volume 1, Eigth Edition, Amdt. 13

III. Technical Characteristic and Operating Limitations

1. Type Design Definition

EA 300/SX, EA-0F102.1 Description of differences to EA 300/SC type design (most current issue);
based on:
EA 300/SC, EA-0C102.1 Description of differences to EA 300/S type design (most current issue); and
EA 300/S, EA-04102.1 Description and Operation of Aircraft and Systems (most current issue)

2. Description

Single engine, low wing cantilever monoplane with reciprocating engine and fixed main gear in tail-wheel configuration; wing, empennage and landing gear in fibre-composite construction; fuselage and engine mount in conventional steel tube construction.

3. Equipment

Equipment List, refer to POH/AFM Doc. N° EA-0F701,
(See Note 3)

4. Dimensions

Span:	7.80 m	(25.59 ft)
Length:	6.65 m	(21.82 ft)
Height:	2.55 m	(8.36 ft)
Wing area:	10.03 m²	(108.00 sq.ft.)

5. Engine

5.1 Model: Lycoming AEIO-580-B1A

5.2 Type Certificate: IM.E.027

5.3 Limitations

Rated power at 2700 RPM: 235 kW/ 315 BHP
Rated power at 2600 RPM: 226 kW/ 303 BHP

Maximum RPM (take-off and max. continuous):

With propeller model 1 and model 2:

Normal & Aerobatic Category: 2600 RPM
Aerobatic Category only: 2700 RPM

(see Note 2)

6. Load factors

Normal category	+6 / -3
Aerobatic category	
at 780 kg (1720 lbs) or below / ACRO I	±10
at 820 kg (1808 lbs) or below / ACRO II	± 8

7. Propeller

7.1 Propeller Model 1

7.1.1 Model: MT Propeller MTV-9-B-C/C198-25

7.1.2 Type Certificate: EASA.P.096 (replaced LBA No. 32.130/65)

7.1.3 Number of blades: 3

7.1.4 Diameter: 1980 mm ± 5 mm

7.1.5 Sense of Rotation: Right-hand tractor (viewed in direction of flight)

7.2 Propeller Model 2

7.2.1	Model	MT Propeller MTV-14-B-C/C190-130
7.2.2	Type Certificate	EASA.P.017
7.2.3	Number of blades	4
7.2.4	Diameter	1900 mm ± 5 mm
7.2.5	Sense of Rotation	Right-hand tractor (viewed in direction of flight)

8. Fluids

8.1	Fuel	100/100LL minimum grade aviation gasoline
8.2	Oil	Single or multi – viscosity aviation grade oils see latest issue of Textron Lycoming S.I. N° 1014
8.3	Coolant	None
8.4	Smoke Oil	Straight paraffin oil, kin. viscosity 30-50 cSt at 20°C (68°F), initial boiling point >330°C (626°F); For example: Fauth FC05, Texaco Canopus 13 or equivalent.

9. Fluid capacities

9.1	Fuel	Total capacity	189.1 Litre	(49.9 US.gal)
		Usable capacity	187.5 Litre	(49.5 US.gal)
		Usable capacity for aerobatics	67.5 Litre	(17.8 US.gal)
9.2	Oil	Max. sump capacity	15.1 Litre	(16 qts)
		Min. sump capacity normal	8.5 Litre	(9 qts)
9.3	Coolant system capacity	None		
9.4	Smoke Oil:		21 Litre	(5.5 US.gal)

10. Air Speeds

Design Manoeuvring Speed V_A:	
Aerobatic category	154 KIAS / 158 KCAS
Normal category	138 KIAS / 140 KCAS
Max. Structural Cruising Speed V_{NO}:	
Aerobatic category	154 KIAS / 158 KCAS
Normal category	138 KIAS / 140 KCAS
Never Exceed Speed V_{NE} :	217 KIAS / 220 KCAS

11. Flight Envelope

Max. operating altitude	3048 m	(10000 ft)

12. Approved Operations Capability

Day-VFR

13. Maximum Masses

Take-off and Landing:		
Normal category	870 kg	(1918 lbs)
Aerobatic category		
ACRO I	780 kg	(1720 lbs)
ACRO II	820 kg	(1808 lbs)

	Empty:		
	Normal category	658 kg	(1451 lbs)
	Aerobatic category		
	ACRO I	645 kg	(1422 lbs)
	ACRO II	685 kg	(1510 lbs)
14. Centre of Gravity Range	Forward limit (aft of datum):		
	at 870 kg (1918 lbs) or below	34.4 cm	(13.54”)
	Rear limit (aft of datum):		
	at 870 kg (1918 lbs) or below	48.2 cm	(18.98”)
15. Datum	Plane of Firewall		
16. Control surface deflections	Aileron:	30°±2° upward;	30°±2° downward
	Elevator:	25°±1° upward,	25°±1° downward
	Rudder:	30°-2° left,	30°-2° right
	Elevator trim tab:	32°±2° upward,	32°±2° downward
17. Levelling Means	Upper fuselage longeron		
18. Minimum Flight Crew	1 Pilot		
19. Maximum Passenger Seating Capacity	None		
20. Baggage/Cargo Compartments	None		
21. Wheels and Tyres	Main Wheel Tyre Size:	5.00-5 6ply	
	Tail Wheel Tyre Size:	Solid rubber 125/50-75 ZL or 6” (optional)	
22. (Reserved)			

IV. Operating and Service Instructions

1.	Flight Manual	
	Pilot’s operating Handbook (POH) & Airplane Flight Manual (AFM)	Doc. No. EA-0F701
2.	Maintenance Manual	
	Service Manual	Doc. No. EA-0F702
3.	Structural Repair Manual	
	Service Manual	Doc. No. EA-0F702
4.	Weight and Balance Manual	
	Pilot’s operating Handbook (POH) & Airplane Flight Manual (AFM)	Doc. No. EA-0F701
5.	Illustrated Parts Catalogue	
	Parts Catalogue	None

V. Notes

1. This certification applies to Serial Numbers SX001 and on.
2. A standard Certificate of Airworthiness can only be issued for an aircraft which is equipped with:
   - the 3-blade propeller MTV-9-B-C/C198-25 in combination with the exhaust silencer system type ACC Columbia EA300-606000 and a reduced max. takeoff engine rotational speed of 2600 RPM.
   - the 4-blade propeller MTV-14-B-C/C190-130 in combination with theexhaust silencer system type ACC Columbia EA300-606000 and a reduced max. take-off engine rotational speed of 2600 RPM.

   Otherwise a Certificate of Airworthiness can only be issued for aerial work.
3. For more certified optional equipment refer to approved AFM/POH Supplements latest revision.
4. Structure temperatures (composite) above 72°C (161.6°F) are not permitted. Not to exceed this temperature limit, colour specification for composite structure of the manufacturer (document EA-03205.19) has to be complied with.

Section 9 Administration

I. Acronyms and Abbreviations

Acronym / Abbreviation	Definition
CAA	Civil Aviation Authority
EASA	European Union Aviation Safety Authority
TCH	Type Certificate Holder
TCDS	Type Certificate Data Sheet
TC	Type Certificate

II. Type Certificate Holder Record

TCH Record	Period
Extra Flugzeugbau GmbH Flugplatz Dinslaken 46569 Hünxe Germany	until 15 September 2003
EXTRA Flugzeugproduktions- und Vertriebs- GmbH Schwarze Heide 21 46569, Hünxe Germany	from 15 September 2003

III. Amendment Record

TCDS Issue No.	TCDS Issue Date	Changes	TC Issue and Date
1	28 Jun 2024	This certificate supersedes EASA.A.362. All data taken from EASA.A.362 Issue 08.	Issue 1 28 Jun 2024

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