Civil Aviation Authority

United Kingdom

FR11
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AGUSTA
A109 Series

TYPE CERTIFICATE DATA SHEET NO. FR11

This data sheet is a part of CAA Type Certificate No.FR11, and gives the basis upon which the Certificate was issued, together with other information to define the product.

Aircraft Manufacturer:	Agusta S.p.A Cascina Costa Gallarate (Va) Italy
Type to which the Certificate relates:	Agusta A109A
Approved variants:	Models A109A II, A109C, A109E

CERTIFICATION BASIS

The following requirements were the basis of certification of the type design:

Italian requirements, FAR Part 27 including amendments 1 to 8 inclusive, FAR Part 29 paragraph 29.903(b). FAA Special Condition No. 27-54-EU-17 and the requirements of United Kingdom CAA Special Survey Nos. 4783, 5198, 5739 and Airworthiness Approval Note 25750 as contained in CAA Additional Requirements Document.

FLIGHT MANUAL

The limitations, recommended procedures and information required are contained in the following Flight Manuals:

CAA approved Rotorcraft Flight Manuals Agusta A109A issued 5 December 1979 and Agusta A109A II issued 7 June 1985 and Agusta A109C issued 12 July 1991 and RAI approved Rotorcraft Flight Manual Agusta A109E third Issue dated 31 May 1996 Revision 6 including CAA green pages approved by RAI at initial issue dated 9 June 1998.

AIRWORTHINESS CATEGORY

UK: Transport Category (Passenger)

PERFORMANCE GROUP

UK: Group A and Group B

MINIMUM CREW

UK: One pilot.

MAXIMUM NUMBER OF OCCUPANTS

Number of persons, including crew, not to exceed eight. The number is limited by the capacity of the fuselage.

ROTORS

	A109C Part Nos.	*A109A* Part Nos.	*A109A* II Part Nos.	*A109E* Part Nos.
Main Rotor, four bladed	709.0103.01-101/-109	109.0103.01-9	109.0103.01-9	709-0103-01-105/109
Main rotor hub assembly	109.0101.01-115	109.1010.01-5/-7	109.0101.01-9/-111/-113	109-0111-02-101
Main transmission	109.0400.02-107	109.0400.02-3	109.0400.02-5	109-0400-03-109
Tail rotor,two-bladed, Semi-rigid	109.8132.01-1	109.0132.02-11	109.0132.02-15	109-8132-01-109

POWER UNITS

A109A and A109 AII: Two Detroit Diesel Allison Model 250-C20B engines.
A109C: Two Detroit Diesel Allison Model 250-C20R/1 engines
A109E: Two Pratt and Whitney Canada Inc. PW206C engines.

FUEL CAPACITY

A109A, AII and C

The capacity of the fuel tanks, in Imp. gallons is:

Tank	Unusable	Unusable	Total
No. 1 Left	60.5	1	61.5
No. 2 Right	60.5	1	61.5
Total	121	2	123

A109E

The capacity of the fuel tanks is litres is:

Tank	Unusable	Unusable	Total
No. 1 Left No. 2 Right	297.5 297.5	5 5	302.5 302.5
Total	595	10	605

OIL CAPACITY

The capacity of each Allison Model 250-C20 engine oil system is 13.5 Imp. gallons.

The capacity of each PW206C engine oil system is 5.12 litres.

MAXIMUM WEIGHTS

	A109A & A109A II	*A109C*	*A109E*
Maximum take-off and landing weight:	5730 lb (2600kg)	5997 lb (2720kg)	2850 kg (6283 lb)
Maximum zero fuel weight:	4800 lb (2180 kg)	5291 lb (2400kg)	2220 kg (4894 lb)

NUMBER OF WHEELS AND MAXIMUM TYRE PRESSURES

Number of wheels in the nose-wheel unit 1
Number of main-wheel units 2
Number of wheels in each main-wheel unit 1
Maximum tyre pressure: Nose and main wheels 85 lbf/in^2

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