TCDS No.: EASA.A.367 DR 200, DR 300, and DR 400 series

Issue: 05 Date: 28 July 2023



TYPE-CERTIFICATE DATA SHEET

NO. EASA.A.367

for DR 200, DR300, AND DR400 SERIES

Type Certificate Holder C.E.A.P.R.

1b route de TROYES 21121, DAROIS FRANCE

For models:

DR 200 DR 221 DR 250 B-160	DR 220 DR 221 B DR 253	DR 220 A DR 250 DR 253 B	DR 220 B DR 250-160	DR 220 AB DR 250 B
DR 340 DR 300/180 R	DR 315 DR 300/140	DR 360 DR 300/125	DR 380 DR 300/120	DR 300/108
DR 400/125 DR 400/2+2 DR 400/160 D DR 400 NGL	DR 400/140 DR 400/120 DR 400/120 D DR 400/200 R	DR 400/160 DR 400/125i DR 400/180 S DR 400/500	DR 400/180 DR 400/140 B DR 400/100 DR 400/200 I	DR 400/180 R DR 400/120 A DR 400 RP



TCDS No.: EASA.A.367 DR 200, DR 300, and DR 400 series

Issue: 05 Date: 28 July 2023

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DR 200, DR 300, and DR 400 series

TCDS No.: EASA.A.367

Issue: 05 Date: 28 July 2023

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TCDS No.: EASA.A.367 DR 200, DR 300, and DR 400 series

Issue: 05 Date: 28 July 2023

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TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

Issue: 05 - DR 200 Date: 28 July 2023

Section A: DR 200

A.I General

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 200

2. Airworthiness Category: Normal Category

Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification Date: April 06, 1965

7. EASA Type Certification Date: Transferred by Commission Regulation (EC) No.

1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 34.

A.II Certification Basis

1. Reference Date for determining

the applicable requirements: 30 October 1964

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052

6. Requirements elected to comply: None7. EASA Special Conditions: None

8. EASA Exemptions: None

9. EASA Equivalent Safety Findings:

10. EASA Environmental Standards: None.

A.III Technical Characteristics and Operational Limitations

1. Type Design Definition: Refer to the CEAPR document 1003343

None

2. Description: Single-engine, four-seat, low-wing airplane, wooden

construction, fixed conventional landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.



Date: 28 July 2023

TCDS No: EASA.A.367

Span8.72 m 4. Dimensions: (28.61 ft) Height 1.83 m (6.00 ft)Length......6.68 m (21.92 ft) Wing Area 14.15 m² (152.31 ft²)

POTEZ 4 E 20 B 5. Engines:

5.1 Engine Limits: Maximum Continuous Power:. 2750 rpm (105 HP, 77 kW)

6. Propellers:

Issue: 05

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Légère	2102 RA (pitch - 0,5)	1.80 m	2	2400 rpm
Ratier	FH-110R (pitch 62.5°)	1.75 m	2	2400 rpm

	Ratier	FH-110R (pitch 6	32.5°)	1.75 m	2	2400 rpm
7.	Fluids:					
	7.1 Fuel:	100	octane r	minimum a	aviation gas	oline grade.
	7.2 Engine Oil:					SAE 30 (AERO 65) SAE 40 (AERO 80)
8.	Fluid capacities:					
	8.1 Fuel:	Wing	g tanks:			2 x 40 litres
						55 litres usable during level flight
	8.2 Oil:	Oil s	ump ca _l	pacity	4	.5 litres (4.8 U.S. quarts)
9.	Air speeds:	V _{NO} V _A V _{FE}			2 1 1	5 km/h (148.5 knots IAS) 40 km/h (130 knots IAS) 78 km/h (92 knots IAS) 50 km/h (81 knots IAS) 40 km/h (130 knots IAS)
10.	Maximum Operating Alti	tude: Refe	r to app	roved airc	raft flight m	anual.
11.	Operational Capability:	Refe	r to app	roved airc	raft flight m	anual.
12.	Maximum Masses:					850 kg (1874 lb) 850 kg (1874 lb)
13.	Centre of Gravity Range	Forw	/ard limi mit (32.	t (18.1 % ı 1 % ref.): .	ref.):	0.31 m aft of datum 0.55 m aft of datum
14.	Datum:				ctangular p	part of the wing. : 1.71 m.
15.	Load factor (n) at maxim	Flap		•	e n /e n	+ 3.8 1.52
16.	Levelling Means:	Horiz	zontal re	eference u	pper fusela	ge spar
17.	Minimum Flight Crew:	1 (pi	lot) at 0.	.42 ±0.05 ı	m aft of dat	um
18.	Maximum Passenger Se	eating Capacity: 1	at 0.42	±0.05 m a	and 2 at 1.1	6m aft of datum.
19.	Baggage/cargo compart				ompartmen nd balance	t 20 kg at 1.85m aft of limits.

TCDS No: EASA.A.367 DR 200, DR300, and DR400 series Issue: 05 DR 200 Date: 28 July 2023 20. Wheels and Tires Wheel tire size main gear wheel:.....380 x 150 tail wheel:6 x 2 Tire pressure......Refer to the maintenance manual 21. Control surface movements: up...... $9.5^{\circ} \pm 0.5^{\circ}$ Elevator: down 12° ± 0.5° up...... 12° ± 0.5° Ailerons: down 12° ± 0.5° before differential braking.....right: 18° left: 15° Elevator trim tab (manual): Elevator nose down Tab down position:4° ± 1° Tab up position: 30° ± 1° Elevator nose up Tab down position: - 11° ± 1° Tab up position: - 16° ± 1° 1^{st} notch $20^{\circ} \pm 3^{\circ}$ Wing Flaps: 2^{nd} notch45° \pm 3° 22. (Reserved) A.IV Operating and Service Instructions

Airplane Flight Manual......Refer to latest amendment of service letter n°6 Airplane Maintenance Manual......Refer to latest amendment of service letter n°6 Airplane Major Inspection Schedule......Refer to latest amendment of service letter n°6

A.V Note:



TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

Issue: 05 - DR 220 Date: 28 July 2023

Section B: DR 220

B.I General

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 220

2. Airworthiness Category: **N**ormal Category and **U**tility Category

Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification Date: June 24, 1966

7. EASA Type Certification Date: Transferred by Commission Regulation (EC) No.

1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 40.

B.II Certification Basis

1. Reference Date for determining

the applicable requirements: 15 November 1965

(Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment May 1st 1965

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None
 EASA Environmental Standards: None

B.III Technical Characteristics and Operational Limitations

1. Type Design Definition Refer to the CEAPR document 1003343

2. Description: Single-engine, four-seat, low-wing airplane, wooden

construction, fixed conventional landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.



Issue: 05 - DR 220 Date: 28 July 2023

4. Dimensions:

TCDS No: EASA.A.367

 Span
 8.72 m
 (28.6 ft)

 Height
 1.90 m
 (6.2 ft)

 Length
 6.80 m
 (22.3 ft)

 Wing Area
 13.60 m²
 (146.4 ft²)

5. Engines: Continental (or Rolls Royce) O-200A

The EASA type certification standard includes that of FAA TC E-252, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior

to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2750 rpm (100 HP, 74.5

kW)

6. Propellers:

Make	Make	α	Number	Minimum static RPM
Make	Model	Ø	of blades	at sea level
Légère	2102 RA (pitch – 0.5)	1.80 m	2	2350 rpm
Légère	2102 RA (pitch - 0°)	1.80 m	2	2300 rpm
Mac Cauley	1B90ECM7250	1.83 m	2	2250 rpm
Jodel Evra	D11-28-7C	1.76 m	2	2250 rpm
Ratier	FH-110R (pitch-3)	1.74 m	2	2300 rpm
Ratier	FH110 (cal-3)	1.85 m	2	2300 rpm

7. Fluids:

7.1 Fuel: 80/87 octane, minimum aviation grade gasoline.

7.2 Engine Oil:

Temperature	Aviation grade	SAE grade	Multi-viscosity
below 40°F (5°C)	65	30	10W-30 / 15W-50 / 20W-50
above 40°F (5°C)	80	50	15W-50 / 20W-50 / 20W-60

8. Fluid capacities:

8.1 Fuel: Main fuel tank capacity:110 litres

Usable:.. the last 5 litres are only usable during level flight

Supplemental fuel tank capacity:.....50 litres

8.2 Oil: Oil sump capacity 5.7 litres (6 U.S. quarts)

9. Air speeds:

V_{NE}	270 km/h (146 knots IAS
V_{NO}	210 km/h (113 knots IAS
V_A	170 km/h (92 knots IAS
V_{FE}	150 km/h (81 knots IAS
V_{C}	210 km/h (113 knots IAS

10. Maximum Operating Altitude: Refer to approved aircraft flight manual.

11. Operational Capability: Refer to approved aircraft flight manual.

. In Normal Category, all aerobatic manoeuvres, including

spins, are forbidden.



TCDS No: EASA.A.367

Issue: 05 **DR 220** Date: 28 July 2023

12. Maximum Masses: **Normal Category** Take-Off......780 kg

Landing741 kg

Utility category T/O & Ldg......700 kg

13. Centre of Gravity Range: Normal category

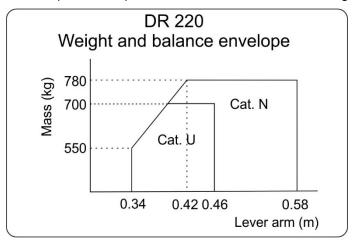
> Forward limit (19.9 % ref.): . 0.34 m aft of datum at 550 kg Intermediate limit (24.5 % ref.):0.42 m aft of datum at 780

> Aft limit (33.9 % ref.): 0.58 m aft of datum at 780 kg

Utility categories

Forward limit (19.9 % ref.): . 0.34 m aft of datum at 550 kg Intermediate limit (22.8 % ref.):0.39 m aft of datum at 700

Aft limit (26.9 % ref.): 0.46 m aft of datum at 700 kg



14. Datum: Leading edge of the rectangular part of the wing. Chord length at reference section: 1.71 m.

15. Load factor (n) at maximum weight: Normal Category:

Flaps up.....+ 3.8 Flaps up.....- 1.52 Flaps down+ 2

Utility Category:

Flaps up+ 4.4 Flaps up.....- 1.76 Flaps down+ 2

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 1 at 0.43 ±0.05 m aft of datum

18. Maximum Passenger Seating Capacity:

1 at 0.43 ±0.05 m and 2 (maximum 90kg) at 1.22m aft of datum

The rear seats can be used only if seat belts are provided and if weight and balance are respected.

Rear seats must not be used in utility category.

19. Wheels and Tires

Wheel tire size main gear wheel:.....380 x 150 tail wheel:6 x 2

Tire pressure......Refer to the maintenance manual

TCDS No: EASA.A.367 DR 220 Issue: 05

20. Control surface movements:

Elevator:	up
Ailerons:	up
flaps	neadan daming eage angilea en
	28° (0°; +2°) ifferential braking15° (0°; +2°)
	osition: 28° tion: 6°30'
Elevator down Tab down p Tab up posi	osition:

 1^{st} notch20° \pm 2°

 2^{nd} notch $60^{\circ} \pm 2^{\circ}$

Date: 28 July 2023

21. (Reserved)

B.IV Operating and Service Instructions

Airplane Flight Manual......Refer to latest amendment of service letter n°6 Airplane Maintenance Manual......Refer to latest amendment of service letter n°6 Airplane Major Inspection Schedule......Refer to latest amendment of service letter n°6

Wing Flaps:

B.V Note:

(Reserved)



TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

Issue: 05 - DR 220 A Date: 28 July 2023

Section C: DR 220 A

C.I General

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 220 A

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification Date: January 4, 1967

7. EASA Type Certification Date: Transferred by Commission Regulation (EC) No.

1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 40.

C.II Certification Basis

1. Reference Date for determining

the applicable requirements: 15 November 1965

(Reserved)
 (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment May 1st 1965

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None
 EASA Environmental Standards: None

C.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition Refer to the CEAPR document 1003343

2. Description: Single-engine, four-seat, low-wing airplane, wooden

construction, fixed conventional landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.



Issue: 05 - DR 220 A Date: 28 July 2023

4. Dimensions:

 Span
 8.72 m
 (28.6 ft)

 Height
 1.90 m
 (6.2 ft)

 Length
 6.80 m
 (22.3 ft)

 Wing Area
 13.60 m²
 (146.4 ft²)

5. Engines: Continental (or Rolls Royce) O-200A

The EASA type certification standard includes that of FAA TC E-252, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior

to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2750 rpm (100 HP, 74.5

kW)

6. Propellers:

Make	Model	Ø	Number of blades	Minimum static RPM at sea level
Jodel Evra	D11-28-7C	1.76 m	2	2250 rpm
Légère	2102RA (pitch - 0°)	1.80 m	2	2300 rpm
Mac Cauley	1B90ECM7250	1.83 m	2	2250 rpm
Ratier	FH-110R (pitch - 3)	1.74 m	2	2300 rpm
Ratier	FH110 (cal - 3)	1.85 m	2	2300 rpm

7. Fluids:

7.1 Fuel: 80/87 octane, minimum aviation grade gasoline.

7.2 Engine Oil:

Temperature	Aviation grade	SAE grade	Multi-viscosity
below 40°F (5°C)	65	30	10W-30 / 15W-50 / 20W-50
above 40°F (5°C)	80	50	15W-50 / 20W-50 / 20W-60

8. Fluid capacities:

Usable:.. the last 5 litres are only usable during level flight Supplemental fuel tank capacity:......50 litres

8.2 Oil: Oil sump capacity 5.7 litres (6 U.S. quarts)

9. Air speeds:

 VNE
 290 km/h (157 knots IAS)

 VNO
 216 km/h (117 knots IAS)

 VA
 190 km/h (103 knots IAS)

 VFE
 150 km/h (81 knots IAS)

 Vc
 216 km/h (117 knots IAS)

10. Maximum Operating Altitude: Refer to approved aircraft flight manual.

11. Operational Capability: Refer to approved aircraft flight manual.

. In Normal Category, all aerobatic manoeuvres, including

spins, are forbidden.



Issue: 05

12. Maximum Masses:

Normal Category Take-Off......780 kg Landing780 kg

Utility category T/O & Ldg......780 kg

13. Centre of Gravity Range:

Normal category

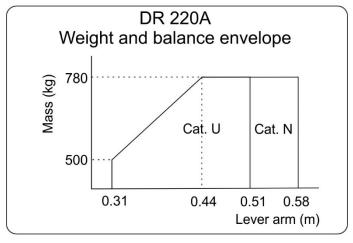
Forward limit (18.1 % ref.): . 0.31 m aft of datum at 500 kg Intermediate limit (25.7 % ref.):0.44 m aft of datum at 780

Aft limit (33.9 % ref.): 0.58 m aft of datum at 780 kg

Utility categories

Forward limit (18.1 % ref.): . 0.31 m aft of datum at 500 kg Intermediate limit (25.7 % ref.):0.44 m aft of datum at 780

Aft limit (29.8 % ref.): 0.51 m aft of datum at 780 kg



14. Datum:

Leading edge of the rectangular part of the wing. Chord length at reference section: 1.71 m.

15. Load factor (n) at maximum weight: Normal Category:

Flaps up.....+ 3.8 Flaps up.....- 1.52

Flaps down+ 2

Utility Category:

Flaps up+ 4.4 Flaps up.....- 1.76 Flaps down+ 2

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.43 ±0.05 m aft of datum

18. Maximum Passenger Seating Capacity:

19. Wheels and Tires

1 at 0.43 ±0.05 m and 2 (maximum 110kg) at 1.22m aft of

datum

The rear seats can be used only if seat belts are provided and if weight and balance are respected.

Rear seats must not be used in utility category.

Wheel tire size main gear wheel:380 x 150

tail wheel:6 x 2

Tire pressure......Refer to the maintenance manual

Issue: 05 - DR 220 A Date: 28 July 2023

20. Control surface movements:

Elevator:	up
Ailerons:	up
flaps	5 5 5
	28° (0°; +2°) fferential braking15° (0°; +2°)
Elevator anti tab: Elevator up Tab down p Tab up posi	osition:
	osition:
Wing Flaps:	$\begin{array}{lll} 1^{st} \ notch & & 20^{\circ} \pm 2^{\circ} \\ 2^{nd} \ notch & & .60^{\circ} \pm 2^{\circ} \end{array}$

21. (Reserved)

C.IV Operating and Service Instructions

Airplane Flight Manual	Refer to latest amendment of service letter n°6
Airplane Maintenance Manual	Refer to latest amendment of service letter n°6
Airplane Major Inspection Schedule	Refer to latest amendment of service letter n°6

C.V Note:

1. This model is identical to DR220 except wing structure and landing gear.

TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

Issue: 05 - DR 220 B Date: 28 July 2023

Section D: DR 220 B

D.I General

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 220 B

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification Date: July 20, 1968

7. EASA Type Certification Date: Transferred by Commission Regulation (EC) No.

1702/2003

D.II Certification Basis

1. Reference Date for determining

the applicable requirements: 15 November 1965

(Reserved)
 (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment May 1st 1965

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None
 EASA Environmental Standards: None

D.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition Refer to the CEAPR document 1003343

2. Description: Single-engine, four-seat, low-wing airplane, wooden

construction, fixed conventional landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.



Issue: 05 - DR 220 B Date: 28 July 2023

4. Dimensions:

TCDS No: EASA.A.367

 Span
 8.72 m
 (28.6 ft)

 Height
 1.90 m
 (6.2 ft)

 Length
 6.80 m
 (22.3 ft)

 Wing Area
 13.60 m²
 (146.4 ft²)

5. Engines: Continental (or Rolls Royce) O-200A

The EASA type certification standard includes that of FAA TC E-252, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior

to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2750 rpm (100 HP, 74.5

kW)

6. Propellers:

Make	Model	Ø	Number of blades	Minimum static RPM at sea level
Légère	2102 RA (pitch - 0,5)	1.80 m	2	2350 rpm
Légère	2102 RA (pitch - 0°)	1.80 m	2	2300 rpm
Mac Cauley	1B90ECM7250	1.83 m	2	2250 rpm
Jodel Evra	D11-28-7C	1.76 m	2	2250 rpm
Ratier	FH-110R (pitch-3)	1.74 m	2	2300 rpm
Ratier	FH110 (cal-3)	1.85 m	2	2300 rpm

7. Fluids:

7.1 Fuel: 80/87 octane, minimum aviation grade gasoline.

7.2 Engine Oil:

Temperature	Aviation grade	SAE grade	Multi-viscosity
below 40°F (5°C)	65	30	10W-30 / 15W-50 / 20W-50
above 40°F (5°C)	80	50	15W-50 / 20W-50 / 20W-60

8. Fluid capacities:

8.1 Fuel: Main fuel tank capacity:110 litres

Usable:.. the last 5 litres are only usable during level flight

Supplemental fuel tank capacity:50 litres

9. Air speeds:

V_{NE}	270 km/r	ı (146	knots	IAS)
V_{NO}	210 km/h	ı (113	knots	IAS)
V_A	170 km/h	i (92	knots	IAS)
V_{FE}	150 km/h	i (81	knots	IAS)
Vc	210 km/h	ı (113	knots	IAS)

10. Maximum Operating Altitude: Refer to approved aircraft flight manual.

11. Operational Capability: Refer to approved aircraft flight manual.

. In Normal Category, all aerobatic manoeuvres, including

spins, are forbidden.



Issue: 05 DR 220 B Date: 28 July 2023

12. Maximum Masses: **Normal Category** Take-Off......780 kg

Landing741 kg

Utility category T/O & Ldg......700 kg

13. Centre of Gravity Range: Normal category

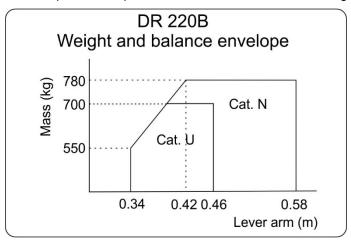
> Forward limit (19.9 % ref.): . 0.34 m aft of datum at 550 kg Intermediate limit (25.6 % ref.):0.42 m aft of datum at 780

Aft limit (33.9 % ref.): 0.58 m aft of datum at 780 kg

Utility categories

Forward limit (19.9 % ref.): . 0.34 m aft of datum at 550 kg Intermediate limit (22.8 % ref.):0.39 m aft of datum at 700

Aft limit (26.9 % ref.): 0.46 m aft of datum at 700 kg



14. Datum: Leading edge of the rectangular part of the wing. Chord length at reference section: 1.71 m.

15. Load factor (n) at maximum weight: Normal Category:

Flaps up	.+ 3.8
Flaps up	
Flaps down	.+ 2

Utility Category:

Flaps up+ 4.4 Flaps up.....- 1.76 Flaps down+ 2

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.43 ±0.05 m aft of datum

18. Maximum Passenger Seating Capacity:

1 at 0.43 ±0.05 m and 2 (maximum 90kg) at 1.22m aft of datum

The rear seats can be used only if seat belts are provided

and if weight and balance are respected.

Rear seats must not be used in utility category.

19. Wheels and Tires

Wheel tire size main gear wheel:380 x 150 tail wheel:6 x 2

Tire pressure......Refer to the maintenance manual

Issue: 05 - DR 220 B Date: 28 July 2023

20. Control surface movements:

TCDS No: EASA.A.367

Elevator:	up
Ailerons:	up
flaps	5 5 5
	28° (0°; +2°) ifferential braking15° (0°; +2°)
Elevator anti tab: Elevator up Tab down p Tab up posi	osition:
Elevator down Tab down p Tab up posi	osition:
Wing Flaps:	$\begin{array}{cccccccccccccccccccccccccccccccccccc$

21. (Reserved)

D.IV Operating and Service Instructions

Airplane Flight Manual	Refer to latest amendment of service letter n°6
Airplane Maintenance Manual	Refer to latest amendment of service letter n°6
Airplane Major Inspection Schedule	Refer to latest amendment of service letter n°6

D.V Note:

1. This model is identical to DR220 except leading edge profile of trapezoidal wing part.



TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

Issue: 05 - DR 220 AB Date: 28 July 2023

Section E: DR 220 AB

E.I General

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 220 AB

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification Date: July 20, 1968

7. EASA Type Certification Date: Transferred by Commission Regulation (EC) No.

1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 40.

E.II Certification Basis

1. Reference Date for determining

the applicable requirements: 15 November 1965

(Reserved)
 (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment May 1st 1965

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None
 EASA Environmental Standards: None

E.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition Refer to the CEAPR document 1003343

2. Description: Single-engine, four-seat, low-wing airplane, wooden

construction, fixed conventional landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.



Issue: 05 - DR 220 AB Date: 28 July 2023

4. Dimensions:

 Span
 8.72 m
 (28.6 ft)

 Height
 1.90 m
 (6.2 ft)

 Length
 6.80 m
 (22.3 ft)

 Wing Area
 13.60 m²
 (146.4 ft²)

5. Engines: Continental (or Rolls Royce) O-200A

The EASA type certification standard includes that of FAA TC E-252, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior

to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2750 rpm - 100 HP (74.5

kW)

6. Propellers:

Make	Model	Ø	Number of blades	Minimum static RPM at sea level
Légère	2102 RA (pitch - 0°)	1.80 m	2	2300 rpm
Mac Cauley	1B90ECM7250	1.83 m	2	2250 rpm
Jodel Evra	D11-28-7C	1.76 m	2	2250 rpm
Ratier	FH-110R (pitch-3)	1.74 m	2	2300 rpm
Ratier	FH110 (cal-3)	1.85 m	2	2300 rpm

7. Fluids:

7.1 Fuel:

80/87 octane, minimum aviation grade gasoline.

7.2 Engine Oil:

Temperature		Aviation grade	SAE grade	Multi-viscosity
	below 40°F (5°C)	65	30	10W-30 / 15W-50 / 20W-50
	above 40°F (5°C)	80	50	15W-50 / 20W-50 / 20W-60

8. Fluid capacities:

8.1 Fuel:	Main fuel tank capacity:	110 litres
	Usable: the last 5 litres are only	usable during level flight

, , , , ,

Supplemental fuel tank capacity:.....50 litres

8.2 Oil: Oil sump capacity 5.7 litres (6 U.S. quarts)

9. Air speeds:

V_{NE}	290 km/h (157 knots IAS)
V_{NO}	216 km/h (117 knots IAS)
V_A	190 km/h (103 knots IAS)
V_{FE}	150 km/h (81 knots IAS)
V_{C}	216 km/h (117 knots IAS)

10. Maximum Operating Altitude: Refer to approved aircraft flight manual.

11. Operational Capability: Refer to approved aircraft flight manual.

In Normal Category, all aerobatic manoeuvres, including

spins, are forbidden.

Issue: 05 - DR 220 AB Date: 28 July 2023

12. Maximum Masses: Normal Category Take-Off......780 kg

Landing780 kg

Utility category T/O & Ldg......780 kg

13. Centre of Gravity Range: Normal category

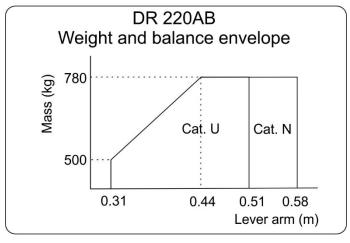
Forward limit (18.1 % ref.): . 0.31 m aft of datum at 500 kg Intermediate limit (25.7 % ref.):0.44 m aft of datum at 780 kg

Aft limit (33.9 % ref.): 0.58 m aft of datum at 780 kg

Utility categories

Forward limit (18.1 % ref.): . 0.31 m aft of datum at 500 kg Intermediate limit (25.7 % ref.): 0.44 m aft of datum at 780 kg

Aft limit (29.8 % ref.): 0.51 m aft of datum at 780 kg



14. Datum:

Leading edge of the rectangular part of the wing. Chord length at reference section: 1.71 m.

15. Load factor (n) at maximum weight: Normal Category:

ai Galegory.	
Flaps up	+ 3.8
Flaps up	1.52
Flaps down	+ 2

Utility Category:

Flaps up + 4.4
Flaps up - 1.76
Flaps down + 2

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.43 ±0.05 m aft of datum

18. Maximum Passenger Seating Capacity:

1 at 0.43 \pm 0.05 m and 2 (maximum 110kg) at 1.22m aft of datum

The rear seats can be used only if seat belts are provided

and if weight and balance are respected.

Rear seats must not be used in utility category.

Wheel tire size main gear wheel:380 x 150 tail wheel:6 x 2

Tire pressure......Refer to the maintenance manual

TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

Issue: 05 - DR 220 AB Date: 28 July 2023

20. Control surface movements:

down 12° (+0°; -0°30')

neutral: trailing edge aligned on

flaps

Rudder L & R:28° (0°; +2°)

before differential braking......15° (0°; +2°)

Elevator anti tab:

Elevator up

Tab down position: 28°
Tab up position: 6°30

Elevator down

21. (Reserved)

E.IV Operating and Service Instructions

E.V Note:

1. This model is identical to DR220A except leading edge profile of trapezoidal wing part.



Issue: 05 - DR 221 Date: 28 July 2023

Section F: DR 221

F.I General

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 221

2. Airworthiness Category: Normal Category and Utility Category

Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

DGAC Type Certification Date: April 25, 1967

7. EASA Type Certification Date: Transferred by Commission Regulation (EC) No.

1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 40.

F.II Certification Basis

1. Reference Date for determining

the applicable requirements: 24 march 1967

(Reserved)
 (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment May 1st 1965

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None

10. EASA Environmental Standards:

F.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition Refer to the CEAPR document 1003343

None

2. Description: Single-engine, four-seat, low-wing airplane, wooden

construction, fixed conventional landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.

4. Dimensions:

 Span
 8.72 m
 (28.6 ft)

 Height
 1.90 m
 (6.2 ft)

 Length
 6.80 m
 (22.3 ft)

 Wing Area
 13.60 m²
 (146.4 ft²)

5. Engines: Lycoming O-235-C2A

The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS



TCDS No: EASA.A.367 DR 221

standards certificated by individual EU member state prior

Date: 28 July 2023

to 28 September 2003 are also acceptable.

Takeoff (max 5 minutes): 2800 rpm - 115 HP (85 kW) 5.1 Engine Limits:

Maximum Continuous power: 2600 rpm - 108 HP (79.5

6. Propellers:

Issue: 05

Make	Model	Ø	Number of blades	Minimum static RPM at sea level
Mac Cauley	1A105BCM7056	1.78 m	2	2400 rpm (Note 1)
Jodel Evra	88-75-34-F	1.76 m	2	2250 rpm

Note 1: Maximum RPM: 2600 rpm

7. Fluids:

7.1 Fuel: 80/87 octane, minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070.

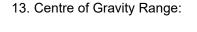
7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No 1014

110. 1014.		
Air temperature	Ashless dispersant (AD)	Mineral
All temperature	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or	SAE30
	SAE20W40	
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

٥.	i iaia sapasitios.				
	8.1 Fuel:	Main fuel tank capacity:110 litres Usable: the last 5 litres are only usable during level fligh			
		Supplemental fuel	tank capacity:50 litres		
	8.2 Oil:				
9.	Air speeds:	V _{NO} V _A V _{FE}			
10.	Maximum Operating Altitude:	Refer to approved aircraft flight manual.			
11.	Operational Capability:	Refer to approved aircraft flight manual In Normal Category, all aerobatic manoeuvres, including spins, are forbidden.			
12.	Maximum Masses:	Normal Category	Take-Off840 kg Landing840 kg		
		Utility category	T/O & Ldg780 kg		



Normal category Forward limit (18.1 % ref.): . 0.31 m aft of datum at 600 kg Intermediate limit (27.5 % ref.):0.47 m aft of datum at 840 kg



TCDS No: EASA.A.367

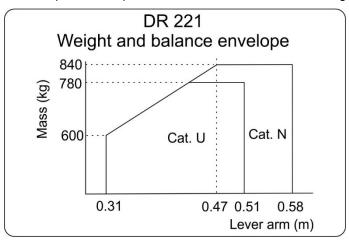
lssue: 05 - DR 221 Date: 28 July 2023

Aft limit (33.9 % ref.): 0.58 m aft of datum at 840 kg

Utility categories

Forward limit (18.1 % ref.): . 0.31 m aft of datum at 600 kg Intermediate limit (25.1 % ref.):0.43 m aft of datum at 780 kg

Aft limit (29.8 % ref.): 0.51 m aft of datum at 780 kg



14. Datum:

Leading edge of the rectangular part of the wing. Chord length at reference section: 1.71 m.

15. Load factor (n) at maximum weight: Normal Category:

Flaps up + 3.8
Flaps up - 1.52
Flaps down + 2

Utility Category:

Flaps up + 4.4 Flaps up - 1.76 Flaps down + 2

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.43 ±0.05 m aft of datum

18. Maximum Passenger Seating Capacity:

1 at 0.43 ±0.05 m and 2 (maximum 120kg) at 1.22m aft of

datum

The rear seats can be used only if seat belts are provided and if weight and balance are respected.

Rear seats must not be used in utility category

Tire pressure......Refer to the maintenance manual

Issue: 05 - DR 221 Date: 28 July 2023

20. Control surface movements:

TCDS No: EASA.A.367

Elevator:	up
Ailerons:	up
flaps	0 0
	28° (0°; +2°) fferential braking15° (0°; +2°)
Elevator anti tab: Elevator up Tab down p Tab up posi	osition:
	osition:
Wing Flaps:	$\begin{array}{lll} 1^{st} \ notch & & 20^{\circ} \pm 2^{\circ} \\ 2^{nd} \ notch & & 60^{\circ} \pm 2^{\circ} \end{array}$

21. (Reserved)

F.IV Operating and Service Instructions

Airplane Flight Manual	Refer to latest amendment of service letter n°6
Airplane Minor inspection Schedule	Refer to latest amendment of service letter n°6
Airplane Major inspection Schedule	Refer to latest amendment of service letter n°6

F.V Note:

1. This model is identical to DR220A except power plant and pitch tab deflection.

Issue: 05 - DR 221 B Date: 28 July 2023

Section G: DR 221 B

G.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 221 B

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification Date: July 20, 1968

7. EASA Type Certification Date: Transferred by Commission Regulation (EC) No.

1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 40.

G.II Certification Basis

1. Reference Date for determining

the applicable requirements: 24 march 1967

(Reserved)
 (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052, amendment May 1st 1965

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None
 EASA Environmental Standards: None

G.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition Refer to the CEAPR document 1003343

2. Description: Single-engine, four-seat, low-wing airplane, wooden

construction, fixed conventional landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.

4. Dimensions:

5. Engines: Lycoming O-235-C2A

The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS



TCDS No: EASA.A.367

standards certificated by individual EU member state prior

Date: 28 July 2023

to 28 September 2003 are also acceptable.

5.1 Engine Limits: Takeoff (max 5 minutes): 2800 rpm - 115 HP (85 kW)

Maximum Continuous Power: 2600 rpm - 108 HP (79.5

6. Propellers:

Issue: 05

Make	Model	Ø	Number of blades	Minimum static RPM at sea level
Mac Cauley	1A105BCM7056	1.78 m	2	2400 rpm (Note 1)
Jodel Evra	88-75-34-F	1.76 m	2	2250 rpm

Note 1: Maximum RPM: 2600 rpm

7. Fluids:

7.1 Fuel: 80/87 octane, minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070.

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014.

110. 101 11					
Air temperature	Ashless dispersant (AD)	Mineral			
Air temperature	grades	grades			
All temperature	SAE15W50 or SAE20W50				
Above 80°F (+25°C)	SAE60	SAE60			
Above 60°F (+15°C)	SAE40 or SAE50	SAE50			
30°F to 90°F (O°C à +30°C)	SAE40	SAE40			
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30			
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50			
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20			

8. Fluid capacities:

0 4 E...I

8.1 Fuel:	iviain tuei tank capacity:	T10 litre			Hitres	

Usable:.. the last 5 litres are only usable during level flight Supplemental fuel tank capacity:.....50 litres

Oil sump capacity 6 U.S. quarts (5.7 litres) 8.2 Oil:

V_{NE}290 km/h (157 knots IAS) 9. Air speeds:

> V_{NO}216 km/h (117 knots IAS) V_A190 km/h (103 knots IAS) V_{FE}150 km/h (81 knots IAS) V_C216 km/h (117 knots IAS)

10. Maximum Operating Altitude: Refer to approved aircraft flight manual.

11. Operational Capability: Refer to approved aircraft flight manual.

In Normal Category, all aerobatic manoeuvres, including

spins, are forbidden.

Take-Off......840 kg 12. Maximum Masses: Normal Category

Landing840 kg

T/O & Ldg......780 kg Utility category

13. Centre of Gravity Range: Normal category



TCDS No: EASA.A.367

Issue: 05

DR 221 B

Forward limit (18.1 % ref.): . 0.31 m aft of datum at 600 kg

Aft limit (33.9 % ref.): 0.58 m aft of datum at 840 kg

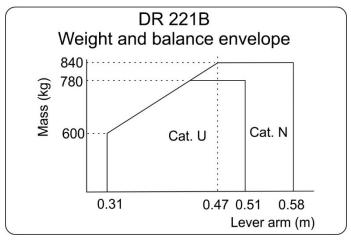
Intermediate limit (27.5 % ref.):0.47 m aft of datum at 840

Date: 28 July 2023

Utility categories

Forward limit (18.1 % ref.): . 0.31 m aft of datum at 600 kg Intermediate limit (25.1 % ref.):0.43 m aft of datum at 780

Aft limit (29.8 % ref.): 0.51 m aft of datum at 780 kg



- Leading edge of the rectangular part of the wing. 14. Datum: Chord length at reference section: 1.71 m.
- 15. Load factor (n) at maximum weight: Normal Category:

Flaps up.....+ 3.8 Flaps up.....- 1.52 Flaps down+ 2

Utility Category:

Flaps up+ 4.4 Flaps up.....- 1.76 Flaps down+ 2

- 16. Levelling Means: Horizontal reference upper fuselage spar
- 17. Minimum Flight Crew: 1 (pilot) at 0.43 ±0.05 m aft of datum
- 18. Maximum Passenger Seating Capacity:

1 at 0.43 ±0.05 m and 2 (maximum 120kg) at 1.22m aft of datum

The rear seats can be used only if seat belts are provided and if weight and balance are respected.

Rear seats must not be used in utility category

19. Wheels and Tires:

> Wheel tire size main gear wheel:.....380 x 150

tail wheel:6 x 2

Tire pressure......Refer to the maintenance manual

20. Control surface movements:

Elevator: down 12° (+0°; -0°30') DR 200, DR300, and DR400 series

Issue: 05 - DR 221 B Date: 28 July 2023

up...... 12° (+0°; -0°30') Ailerons: down 12° (+0°; -0°30') neutral: trailing edge aligned on flaps Rudder L & R:28° (0°; +2°) before differential braking......15° (0°; +2°) Elevator anti tab: Elevator up Tab down position: 29°30 Tab up position:10° Elevator down Tab down position: 8°30 Tab up position: 13°30 1^{st} notch $20^{\circ} \pm 2^{\circ}$ Wing Flaps: 2^{nd} notch $60^{\circ} \pm 2^{\circ}$

21. (Reserved)

TCDS No: EASA.A.367

G.IV Operating and Service Instructions

Airplane Flight Manual	.Refer to latest amendment of service letter n°6
	.Refer to latest amendment of service letter n°6
Airplane Major inspection Schedule	.Refer to latest amendment of service letter n°6

G.V Note:

1. This model is identical to DR221 except leading edge profile of trapezoidal wing part.



Issue: 05 - DR 250 Date: 28 July 2023

Section H: DR 250

H.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 250

2. Airworthiness Category: Normal Category

Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification Date: May 25, 1965

7. EASA Type Certification Date: Transferred by Commission Regulation (EC) No.

1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 34.

H.II Certification Basis

1. Reference Date for determining

the applicable requirements: 18 May 1965

(Reserved)
 (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment May 1st 1965

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None
 EASA Environmental Standards: None.

H.III <u>Technical Characteristics and Operational Limitations</u>

Type Design Definition Refer to the CEAPR document 1003344

2. Description: Single-engine, four-seat, low-wing airplane, wooden

construction, fixed conventional landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.

4. Dimensions:

 Span
 8.72 m
 (28.61 ft)

 Height
 1.86 m
 (6.10 ft)

 Length
 6.98 m
 (22.90 ft)

 Wing Area
 14.15 m²
 (152.31 ft²)

5. Engines: Lycoming O-320 E2A (150 HP)

The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS



TCDS No: EASA.A.367 Issue: 05 DR 250

> standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Date: 28 July 2023

5.1 Engine Limits:

Maximum Continuous Power:2700 rpm (152 HP, 112 kW)

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
HOFFMANN Croisière	FH2/LC23 180 155-6.5R	1.80 m	2	2150 rpm
HOFFMANN	FH2/LC23 180-140-6,5 R	1.80 m	2	2250 rpm
JODEL EVRA	91-78-34	1.84 m	2	2250 rpm
SENSENICH	M74 DMS-2-64	1.83 m (*)	2	2150 rpm
SENSENICH	74 DM 6S5-2-64	1.83 m (*)	2	2150 rpm
SENSENICH	M74 DMS-0-64	1.88 m	2	2200 rpm
SENSENICH	74 DM 6S5-0-64	1.88 m	2	2200 rpm

Remark: (*) no diameter reduction for repairs.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Manufacturer	Model	Ø	Number of blades	Governor	Minimum static RPM at sea level
HARTZELL	HC-C2YL-1 7663-4	1.83 m	2	Hartzell H1	Constant speed

The EASA type certification standard includes that of FAA TC P-920, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 80/87 octane minimum aviation gasoline grade. Refer to

latest revision of Service Instruction Lycoming No. 1070

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014

140: 1014:				
Air temperature	Ashless dispersant	Mineral		
All temperature	(AD)grades	grades		
All temperature	SAE15W50 or SAE20W50			
Above 80°F (+25°C)	SAE60	SAE60		
Above 60°F (+15°C)	SAE40 or SAE50	SAE50		
30°F to 90°F (O°C à +30°C)	SAE40	SAE40		
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or	SAE30		
,	SAE20W40			
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50		
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20		

8. Fluid capacities:

8.1 Fuel: Wing tanks:2 x 40 litres Main fuel tank capacity:70 litres

Usable:.. the last 7 litres are only usable during level flight



TCDS No: EASA.A.367

Issue: 05 DR 250 Date: 28 July 2023 Supplemental fuel tank capacity:.....50 litres 8.2 Oil: Oil sump capacity 8 U.S. quarts (7.6 litres) Usable...... 6 U.S. quarts (5.7 litres) 9. Air speeds: V_{NE}295 km/h (159 knots IAS) V_{NO}260 km/h (140 knots IAS) V_{FE}170 km/h (92 knots IAS) V_C260 km/h (140 knots IAS) 10. Maximum Operating Altitude: Refer to approved aircraft flight manual. 11. Operational Capability: Refer to approved aircraft flight manual. 12. Maximum Masses: 13. Centre of Gravity Range: Forward limit (17 % ref.): 0.29 m aft of datum Aft limit (33 % ref.): 0.565 m aft of datum 14. Datum: Leading edge of the rectangular part of the wing. Chord length at reference section: 1.71 m. 15. Load factor (n) at maximum weight: Flaps retracted positive n+ 3.8 Flaps retracted negative n..... 1.52 16. Levelling Means: Horizontal reference upper fuselage spar 17. Minimum Flight Crew: 1 (pilot) at 0.42 ±0.05 m aft of datum 18. Maximum Passenger Seating Capacity: 1 at 0.42 ±0.05 m and 2 at 1.16m aft of datum. 19. Baggage/cargo compartment: Maximum baggage compartment 40 kg at 1.90m aft of datum, within weight and balance limits. See note 1. "Supplementary rear fuel tank". 20. Wheels and Tires: main gear wheel:380 x 150 Wheel tire size tail wheel:6 x 2 Tire pressure......Refer to the maintenance manual 21. Control surface movements: up...... $9.5^{\circ} \pm 0.5^{\circ}$ Elevator: down 12° ± 0.5° Ailerons: up...... 12° ± 0.5° down $12^{\circ} \pm 0.5^{\circ}$ before differential braking.....right: 18° Elevator trim tab (manual): Elevator nose down Tab down position:4° ± 1° Tab up position: 30° ± 1° Elevator nose up Tab down position: - 11° ± 1° Tab up position: - 16° ± 1° 1^{st} notch $20^{\circ} \pm 3^{\circ}$ Wing Flaps:

22. (Reserved)



 2^{nd} notch $60^{\circ} \pm 3^{\circ}$

TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

Issue: 05 - DR 250 Date: 28 July 2023

H.IV Operating and Service Instructions

H.V Note:

1. Supplementary rear fuel tank operation:

Always use first 50 litres of fuel from rear main tank (more or less 1h40min) then transfer fuel from the supplementary tank to the rear main tank (by pulling the valve control located on the instrument panel).

Balance limits with all 4 tanks full, load is generally limited to either:

- 0 kg in luggage compartment (normal load on rear seats: 154 kg)

Or

- 100 kg on rear seats (40 kg in luggage compartment).

FOR ALL LOADING, USE THE LOADING GRAPH

2. This model is identical to DR200 with Lycoming 0-320 E 2 A (150 HP) engine, longer and larger fuselage, flap defection up to 60°, fuel capacity increased (rear and supplemental tank).

Issue: 05 - DR 250 - 160 Date: 28 July 2023

Section I: <u>DR 250 - 160</u>

I.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 250 - 160

2. Airworthiness Category: Normal Category

Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification Date: September 09, 1965

7. EASA Type Certification Date: Transferred by Commission Regulation (EC) No.

1702/2003

8. The EASA Type Certificates replaces DGAC-France Type Certificate no. 34.

I.II Certification Basis

1. Reference Date for determining

the applicable requirements: May 1965

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment May 1st 1965

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None
 EASA Environmental Standards: None.

I.III <u>Technical Characteristics and Operational Limitations</u>

Type Design Definition Refer to the CEAPR document 1003344

2. Description: Single-engine, four-seat, low-wing airplane, wooden

construction, fixed conventional landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.

4. Dimensions:

 Span
 8.72 m
 (28.61 ft)

 Height
 1.86 m
 (6.10 ft)

 Length
 6.98 m
 (22.90 ft)

 Wing Area
 14.15 m²
 (152.31 ft²)

5. Engines: Lycoming O-320 D2A (160 HP)

The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state acceptance or certification of this standard prior to 28



TCDS No: EASA.A.367

September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior

Date: 28 July 2023

to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power:2700 rpm (162 HP, 119 kW)

6. Propellers:

Issue: 05

Manufacturer	Model	Ø	Number	Minimum static RPM
Maridiacturei	Model	Ø	of blades	at sea level
HOFFMANN	FH2/LC23 180-155-6,5R	1.80 m	2	2250 rpm
HOFFMANN	FH2/LC23 180-140-6,5R	1.80 m	2	2350 rpm
SENSENICH	M74 DMS-0-64	1.88 m	2	2200 rpm
SENSENICH	74 DM 6S5-0-64	1.88 m	2	2200 rpm
SENSENICH	74 DM 6S5-2-64	1.83 m (*)	2	2150 rpm
SENSENICH	M74 DMS-2-66	1.83 m (*)	2	2150 rpm
SENSENICH	74 DM 6S5-2-66	1.83 m (*)	2	2150 rpm
JODEL EVRA	91-86-34 F	1.82 m	2	2250 rpm
JODEL EVRA	91-78-34 F	1.84 m	2	2300 rpm

Remark: (*) no diameter reduction for repairs.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 91/96 octane minimum aviation gasoline grade. Refer to

latest revision of Service Instruction Lycoming No. 1070.

Take-Off: 960 kg (2116.4 lb)

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
All temperature	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

12. Maximum Masses:

8.1 Fuel:	Wing tanks:
	Supplemental fuel tank capacity:50 litres
8.2 Oil:	Oil sump capacity
9. Air speeds:	VNE .295 km/h (159 knots IAS) VNO .260 km/h (140 knots IAS) VA .186 km/h (100 knots IAS) VFE .170 km/h (92 knots IAS) VC .260 km/h (140 knots IAS)
10. Maximum Operating Altitude:	Refer to approved aircraft flight manual.
11. Operational Capability:	Refer to approved aircraft flight manual.



DR 200, DR300, and DR400 series

Issue: 05 - DR 250 - 160 Date: 28 July 2023

Landing 920 kg (2028.3 lb) 13. Centre of Gravity Range: Forward limit (17 % ref.): 0.29 m aft of datum Aft limit (33 % ref.): 0.565 m aft of datum 14. Datum: Leading edge of the rectangular part of the wing. Chord length at reference section: 1.71 m. 15. Load factor (n) at maximum weight: Flaps retracted positive n+ 3.8 Flaps retracted negative n.....- 1.52 16. Levelling Means: Horizontal reference upper fuselage spar 17. Minimum Flight Crew: 1 (pilot) at 0.42 ±0.05 m aft of datum 18. Maximum Passenger Seating Capacity: 1 at 0.42 ±0.05 m aft and 2 at 1.16m aft of datum 19. Baggage/cargo compartment Maximum baggage compartment 40 kg at 1.90m aft of datum, within weight and balance limits. See note 1. "Supplementary rear fuel tank". 20. Wheels and Tires Wheel tire size main gear wheel:.....380 x 150 tail wheel:6 x 2 Tire pressure......Refer to the maintenance manual 21. Control surface movements: Elevator: up...... $9.5^{\circ} \pm 0.5^{\circ}$ down 12° ± 0.5° up...... 12° ± 0.5° Ailerons: down $12^{\circ} \pm 0.5^{\circ}$ before differential braking.....right: 18° left: 15° Elevator trim tab (manual): Elevator nose down Tab down position: $4^{\circ} \pm 1^{\circ}$ Tab up position: 30° ± 1° Elevator nose up Tab down position: - 11° ± 1° Tab up position: - 16° ± 1° 1st notch20° ± 3° Wing Flaps: 2^{nd} notch $60^{\circ} \pm 3^{\circ}$

22. (Reserved)

TCDS No: EASA.A.367



TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

lssue: 05 - DR 250 - 160 Date: 28 July 2023

I.IV Operating and Service Instructions

I.V Note:

1. Supplementary rear fuel tank operation:

Always use first 50 litres of fuel from rear main tank (more or less 1h40min) then transfer fuel from the supplementary tank to the rear main tank (by pulling the valve control located on the instrument panel).

Balance limits with all 4 tanks full, load is generally limited to either:

- 0 kg in luggage compartment (normal load on rear seats: 154 kg)

Or

- 100 kg on rear seats (40 kg in luggage compartment).

FOR ALL LOADING, USE THE LOADING GRAPH

2. This model is identical to DR250 with Lycoming 0-320 D 2 A (160 HP) engine.



Issue: 05 - DR 250 B Date: 28 July 2023

Section J: DR 250 B

J.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 250 B

2. Airworthiness Category: Normal Category

Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification Date: July 20, 1968

7. EASA Type Certification Date: Transferred by Commission Regulation (EC) No.

1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 34.

J.II Certification Basis

1. Reference Date for determining

the applicable requirements: May 1965

(Reserved)
 (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment May 1st 1965

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None
 EASA Environmental Standards: None.

J.III <u>Technical Characteristics and Operational Limitations</u>

Type Design Definition Refer to the CEAPR document 1003344

2. Description: Single-engine, four-seat, low-wing airplane, wooden

construction, fixed conventional landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.

4. Dimensions: Span8.72 m (28.61 ft)

5. Engines: Lycoming O-320 E2A (150 HP)

The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS



TCDS No: EASA.A.367 Issue: 05 DR 250 B

standards certificated by individual EU member state prior

Date: 28 July 2023

to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2700 rpm (152 HP, 112 kW)

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
HOFFMANN Croisière	FH2/LC23 180 155-6.5R	1.80 m	2	2150 rpm
HOFFMANN	FH2/LC23 180-140-6,5 R	1.80 m	2	2250 rpm
JODEL EVRA	91-78-34	1.84 m	2	2250 rpm
SENSENICH	M74 DMS-2-64	1.83 m (*)	2	2150 rpm
SENSENICH	74 DM 6S5-2-64	1.83 m (*)	2	2150 rpm
SENSENICH	M74 DMS-0-64	1.88 m	2	2200 rpm
SENSENICH	74 DM 6S5-0-64	1.88 m	2	2200 rpm

Remark: (*) no diameter reduction for repairs.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Manufacturer	Model	Ø	Number of blades	Governor	Minimum static RPM at sea level
HARTZELL	HC-C2YL-1 7663-4	1.83 m	2	Hartzell H1	Constant speed

The EASA type certification standard includes that of FAA TC P-920, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

80/87 octane minimum aviation gasoline grade. Refer to latest revision of Service Instruction Lycoming No. 1070.

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
7 til temperature	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à	SAE40	SAE40
+30°C)		
0°F to 70°F (-15°C à	SAE30, SAE40 or	SAE30
+20°C)	SAE20W40	
0°F to 90°F (-15°C à	SAE20W50 or SAE15W50	SAE20W50
+30°C)		
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

Fluid capacities

8.1 Fuel: Wing tanks:2 x 40 litres Main fuel tank capacity:70 litres



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			litres are only usable during level flight	
		Supplemental fuel t	ank capacity:50 litres	
	8.2 Oil:	Oil sump capacity . Usable		
9.	Air speeds:	V _{NO} V _A V _{FE}		
10.	Maximum Operating Altitude:	Refer to approved a	aircraft flight manual.	
11.	Operational Capability:	Refer to approved a	aircraft flight manual.	
12.	Maximum Masses:		960 kg (2116.4 lb) 920 kg (2028.3 lb)	
13.	Centre of Gravity Range:		5)	
			ref.):	
14.	Datum:		e rectangular part of the wing. erence section: 1.71 m.	
15.	Load factor (n) at maximum weight:			
			ed positive n+ 3.8 ed negative n1.52	
16.	Levelling Means:	Horizontal reference	e upper fuselage spar	
17.	Minimum Flight Crew:	1 (pilot) at 0.42 ±0.0	05 m aft of datum	
18.	Maximum Passenger Seating Capa	city: 1 at 0.42 ±0.05	m and 2 at 1.16m aft of datum.	
19.	Baggage/cargo compartment		e compartment 40 kg at 1.90m aft of ght and balance limits. See note 1. ar fuel tank".	
20.	Wheels and Tires	Wheel tire size	main gear wheel:	
21.	Control surface movements:			
		Elevator:	up	
		Ailerons:	up	
			differential brakingright: 18° left: 15°	
		Elevator trim tab (m	nanual):	
		Elevator nos	e down position:4° ± 1°	
			position: 30° ± 1°	
		Elevator nose up		
			position: 11° ± 1° bition: 16° ± 1°	
		Wing Flaps:	1^{st} notch	



TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

Issue: 05 - DR 250 B Date: 28 July 2023

22. (Reserved)

J.IV Operating and Service Instructions

J.V Note:

1. Supplementary rear fuel tank operation:

Always use first 50 litres of fuel from rear main tank (more or less 1h40min) then transfer fuel from the supplementary tank to the rear main tank (by pulling the valve control located on the instrument panel).

Balance limits with all 4 tanks full, load is generally limited to either:

- 0 kg in luggage compartment (normal load on rear seats: 154 kg)

Or

- 100 kg on rear seats (40 kg in luggage compartment).

FOR ALL LOADING, USE THE LOADING GRAPH

2. This model is identical to DR250 except the leading edge profile of trapezoidal wing part.



Issue: 05 - DR 250 B - 160 Date: 28 July 2023

Section K: <u>DR 250 B - 160</u>

K.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 250 B - 1602. Airworthiness Category: Normal Category

3. Type Certificate Holder: Refer to Note 2 Section PP4. Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification Date: July 20, 1968

7. EASA Type Certification Date: Transferred by Commission Regulation (EC) No.

1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 34.

K.II Certification Basis

1. Reference Date for determining

the applicable requirements: May 1965

(Reserved)
 (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment May 1st 1965

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None
 EASA Environmental Standards: None

K.III <u>Technical Characteristics and Operational Limitations</u>

Type Design Definition Refer to the CEAPR document 1003344

2. Description: Single-engine, four-seat, low-wing airplane, wooden

construction, fixed conventional landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 must be installed.

4. Dimensions:

 Span
 8.72 m
 (28.61 ft)

 Height
 1.86 m
 (6.10 ft)

 Length
 6.98 m
 (22.90 ft)

 Wing Area
 14.15 m²
 (152.31 ft²)

5. Engines: Lycoming O-320 D2A (160 HP)

The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state acceptance or certification of this standard prior to 28



TCDS No: EASA.A.367 Issue: 05 DR 250 B - 160

> September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Date: 28 July 2023

Maximum Continuous Power:2700 rpm (162 HP, 119 kW)

6. Propellers:

5.1 Engine Limits:

Manufacturer	Model	Ø	Number	Minimum static RPM
Manuaciurei	Wodel	Ø	of blades	at sea level
HOFFMANN	FH2/LC23 180-155-6,5R	1.80 m	2	2250 rpm
HOFFMANN	FH2/LC23 180-140-6,5R	1.80 m	2	2350 rpm
SENSENICH	M74 DMS-0-64	1.88 m	2	2200 rpm
SENSENICH	74 DM 6S5-0-64	1.88 m	2	2200 rpm
SENSENICH	74 DM 6S5-2-64	1.83 m (*)	2	2150 rpm
SENSENICH	M74 DMS-2-66	1.83 m (*)	2	2150 rpm
SENSENICH	74 DM 6S5-2-66	1.83 m (*)	2	2150 rpm
JODEL EVRA	91-86-34 F	1.82 m	2	2250 rpm
JODEL EVRA	91-78-34 F	1.84 m	2	2300 rpm

Remark: (*) no diameter reduction for repairs.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 91/96 octane minimum aviation gasoline grade. Refer to

latest revision of Service Instruction Lycoming No. 1070.

Refer to latest revision of Service Instruction Lycoming 7.2 Engine Oil:

No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
7 til tomporataro	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

Issue: 05 DR 250 B - 160

TCDS No: EASA.A.367

Date: 28 July 2023 8. Fluid capacities: Wing tanks:2 x 40 litres 8.1 Fuel: Main fuel tank capacity:70 litres Usable:.. the last 7 litres are only usable during level flight Supplemental fuel tank capacity:.....50 litres Oil sump capacity 8 U.S. guarts (7.6 litres) 8.2 Oil: Usable...... 6 U.S. quarts (5.7 litres) 9. Air speeds: V_{NE}295 km/h (159 knots IAS) V_{NO}260 km/h (140 knots IAS) V_A186 km/h (100 knots IAS) V_{FE}170 km/h (92 knots IAS) V_C260 km/h (140 knots IAS) 10. Maximum Operating Altitude: Refer to approved aircraft flight manual. 11. Operational Capability: Refer to approved aircraft flight manual. 12. Maximum Masses: Landing 920 kg (2028.3 lb) 13. Centre of Gravity Range: Forward limit (17 % ref.): 0.29 m aft of datum Aft limit (33 % ref.): 0.565 m aft of datum 14. Datum: Leading edge of the rectangular part of the wing. 15. Load factor (n) at maximum weight: Flaps retracted positive n+ 3.8 Flaps retracted negative n.....- 1.52 16. Levelling Means: Horizontal reference upper fuselage spar 17. Minimum Flight Crew: 1 (pilot) at 0.42 ±0.05 m aft of datum 18. Maximum Passenger Seating Capacity: 1 at 0.42 ±0.05 m and 2 at 1.16m aft of datum. 19. Baggage/cargo compartment: Maximum baggage compartment 40 kg at 1.90m aft of datum, within weight and balance limits. See note 1. "Supplementary rear fuel tank". 20. Wheels and Tires: Wheel tire size main gear wheel:380 x 150 tail wheel:6 x 2 Tire pressure......Refer to the maintenance manual 21. Control surface movements: up...... $9.5^{\circ} \pm 0.5^{\circ}$ Elevator: Ailerons: up...... $12^{\circ} \pm 0.5^{\circ}$ down $12^{\circ} \pm 0.5^{\circ}$ before differential braking.....right: 18° left: 15° Elevator trim tab (manual): Elevator nose down Tab down position:4° ± 1°



DR 200, DR300, and DR400 series

lssue: 05 - DR 250 B - 160 Date: 28 July 2023

Elevator nose up

Tab down position: $-11^{\circ} \pm 1^{\circ}$ Tab up position: $-16^{\circ} \pm 1^{\circ}$

22. (Reserved)

TCDS No: EASA.A.367

K.IV Operating and Service Instructions

K.V Note:

1. Supplementary rear fuel tank operation:

Always use first 50 litres of fuel from rear main tank (more or less 1h40min) then transfer fuel from the supplementary tank to the rear main tank (by pulling the valve control located on the instrument panel).

Balance limits with all 4 tanks full, load is generally limited to either:

- 0 kg in luggage compartment (normal load on rear seats: 154 kg)

Or

- 100 kg on rear seats (40 kg in luggage compartment).

FOR ALL LOADING, USE THE LOADING GRAPH

2. This model is identical to DR250/160 except the leading edge profile of trapezoidal wing part.

TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

Issue: 05 - DR 253 Date: 28 July 2023

Section L: DR 253

L.I General

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 253

2. Airworthiness Category: Normal Category

Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification Date: July 11, 1967

7. EASA Type Certification Date: Transferred by Commission Regulation (EC) No.

1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no 42

L.II Certification Basis

1. Reference Date for determining

the applicable requirements: June 1966

(Reserved)
 (Reserved)

4. Certification Basis: France AIR2052.

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966.

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None
 EASA Environmental Standards: None.

L.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition Refer to the CEAPR document 1003346

2. Description: Single-engine, four-seat, low-wing airplane, wooden

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Stall warning Safe Flight n°164

Issue: 05 - DR 253 Date: 28 July 2023

5. Engines: Lycoming O-360 A1A (variable-pitch propeller)

Lycoming O-360 A3A (Sensenich propeller)

The EASA type certification standard includes that of FAA TC E-286, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior

to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power:2700 rpm (133 kW, 183 HP)

6. Propellers:

TCDS No: EASA.A.367

Make	Model	Ø	Number of blades	Minimum static RPM at sea level
SENSENICH	M 76 EMMS-0-64 76 EM8S5-0-64	1,93 m (76 in.)	2	2300 rpm

The EASA type certification standard includes that of FAA TC P4EA, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Manufacturer	Model	Ø	Number of blades	Governor	Minimum static RPM at sea level
HARTZELL	HC-C2YK 7666-2	1,88 m (74 in.)	2	Hartzell D 16 or F3	Constant speed low pitch: 12° high: 28° 8 (*)

Remark: (*) Continuous operation between 2000 and 2250 rpm must be avoided.

The EASA type certification standard includes that of FAA TC P-920, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 100/100LL octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070.

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming No. 1014.

	•	
Air temperature	Ashless dispersant (AD)	Mineral
All temperature	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20



TCDS No: EASA.A.367 Issue: 05 DR 253

8. Fluid capacities:

Wing tank: 40 litres in each tank 8.1 Fuel:

> (Note: The last 7 litres are usable only in horizontal flight

Date: 28 July 2023

attitude)

Auxiliary tank: (see note 1)......50 or 60 litres

8.2 Oil: Oil sump capacity 8 U.S. quarts (7.6 litres)

Usable...... 6 U.S. quarts (5.7 litres)

9. Air speeds: V_{NE}310 km/h (167 knots IAS)

V_{NO}260 km/h (140 knots IAS) V_A203 km/h (109 knots IAS) V_{FE}170 km/h (91.8 knots IAS) V_C260 km/h (140 knots IAS)

Stall speed at maximum weight:

Flaps retracted:104 km/h (56 knots IAS)

Flaps extended:.....96 km/h (51.8 knots IAS)

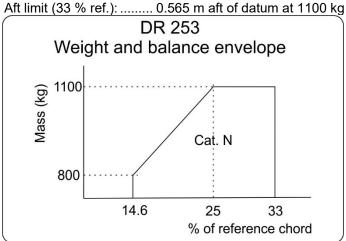
Refer to approved aircraft flight manual. 10. Maximum Operating Altitude:

Refer to approved aircraft flight manual. 11. Operational Capability:

12. Maximum Mass: take-off1100 kg landing1045 kg

13. Centre of Gravity Range: Normal category

Forward limit (14.6 % ref.): 0.250 m aft of datum at 800 kg Intermediate limit (25 % ref.):0.430 m aft of datum at 1100



14. Datum: Wing leading edge of the rectangular part of wing. Cord length at reference section: 1.71 m.

15. Load factor (n) at maximum weight: Normal Category:

Flaps up.....+ 3.8 Flaps up.....- 1.52

16. Levelling Means: Horizontal reference upper fuselage spar.

17. Minimum Flight Crew: 1 (pilot) at 0.47 ±0.05 m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.47 ±0.05 m and 2 at 1.25m aft of datum

DR 200, DR300, and DR400 series

Issue: 05 - DR 253 Date: 28 July 2023

19. Baggage / Cargo Compartment:		e limits of weight and bala + 2.		
20. Wheels and tires:		K		
	Wheel tire size		4	20 x 150
		380 x 150 or 5.	00-5 (Se	e note 2
		Refer to the mai	•	•
	Front wheel mo	ovements (left and right):.		. 25° +2° _{-0°}
21. Control surface movements:	Elevator:	n	ose up:	13° +0,5° -0°
		nos	se down:	6° +0,5° -0°
	Ailerons:		up:	12° -0° +0,5°
			down:	12° -0° +0,5°
	Rudder:		L&F	R: 25° +0°
	minir	num before differential bra	aking (L	& R): 15°
	Wing Flaps:	1 st notch (T/O)		15° +0° -5°
		2 nd notch (Ldg)		60° +0°
	Elevator tab:			
				ab
			Up move	ments Down
		Maximum "Nose up"	10°5	31°

22. (Reserved)

TCDS No: EASA.A.367

L.IV Operating and service instructions

Maximum "Nose down"

3°



TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

Issue: 05 - DR 253 Date: 28 July 2023

L.V Note:

1. Supplementary rear fuel tank operation:

Always use first 60 litres of fuel from rear main tank (more or less 1h30min) then transfer fuel from the supplementary tank to the rear main tank (by pulling the valve control located on the instrument panel).

Balance limits with all 4 tanks full, load is generally limited to either:

- 0 kg in luggage compartment (normal load on rear seats: 154 kg)

Or

- 100 kg on rear seats (40 kg in luggage compartment).

FOR ALL LOADING, USE THE LOADING GRAPH

2. Alternative Oleo Strut and Tires:

Tires size 380x150 or 5.00-5 can only be installed on the aircraft if the DR400 landing gear has been installed in accordance with the SB 160403.



TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

Issue: 05 - DR 253 B Date: 28 July 2023

Section M: DR 253 B

M.I General

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 253 B

2. Airworthiness Category: Normal Category

Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification Date: July 20, 1968

7. EASA Type Certification Date: Transferred by Commission Regulation (EC) No.

1702/2003

8. The EASA type Certificates replaces DGAC-France Type Certificate no 42

M.II Certification Basis

1. Reference date for determining

the applicable requirements: June 1966

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966.

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: None

M.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition Refer to the CEAPR document 1003346

2. Description: Single-engine, four-seat, low-wing airplane, wooden

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Stall warning Safe Flight n°164

Issue: 05 - DR 253 B Date: 28 July 2023

5. Engines: Lycoming O-360 A1A (variable-pitch propeller)

Lycoming O-360 A3A (Sensenich propeller)

The EASA type certification standard includes that of FAA TC E-286, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior

to 28 September 2003 are also acceptable.

5.1 Engine limits: Maximum Continuous Power:2700 rpm (133 kW, 183 HP)

6. Propellers:

TCDS No: EASA.A.367

Make	Model	Ø	Number of blades	Minimum static RPM at sea level
SENSENICH	M 76 EMMS-0-64 76 EM8S5-0-64	1,93 m (76 in.)	2	2300 rpm

The EASA type certification standard includes that of FAA TC P4EA, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Manufacturer	Model	Ø	Number of blades	Governor	Minimum static RPM at sea level
HARTZELL	HC-C2YK 7666-2	1,88 m (74 in.)	2	Hartzell D 16 or F3	Constant speed low pitch: 12° high: 28° 8 (*)

Remark: (*) Continuous operation between 2000 and 2250 rpm must be avoided.

The EASA type certification standard includes that of FAA TC P-920, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 100/100LL octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070.

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
All temperature	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20



Issue: 05 DR 253 B Date: 28 July 2023

8. Fluid capacities:

8.1 Fuel:	Wing tank:	40 litres in each tank
-----------	------------	------------------------

Fuselage tank:100 litres (Note: The last 7 litres are usable only in horizontal flight

attitude)

Auxiliary tank: (see note 1)......50 or 60 litres

8.2 Oil: Oil sump capacity 8 U.S. quarts (7.6 litres)

Usable...... 6 U.S. quarts (5.7 litres)

9. Air speeds: V_{NE}310 km/h (167 knots IAS)

V_{NO}......260 km/h (140 knots IAS) V_A203 km/h (109 knots IAS) V_{FE}170 km/h (91.8 knots IAS) V_C260 km/h (140 knots IAS)

Stall speed at maximum weight:

Flaps retracted:104 km/h (56 knots IAS) Flaps extended:.....96 km/h (51.8 knots IAS)

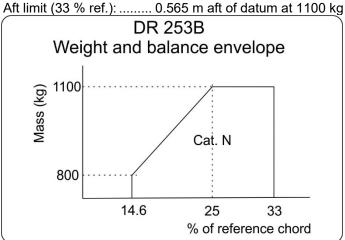
Refer to approved aircraft flight manual. 10. Maximum operating altitude:

Refer to approved aircraft flight manual. 11. Operational capability:

12. Maximum mass: take-off1100 kg landing1045 kg

13. Centre of gravity range: **Normal Category**

Forward limit (14.6 % ref.): 0.250 m aft of datum at 800 kg Intermediate limit (25 % ref.):0.430 m aft of datum at 1100



14. Datum: Wing leading edge of the rectangular part of wing. Cord length at reference section: 1.71 m.

15. Load factor (n) at maximum weight: Normal Category:

Flaps up.....+ 3.8 Flaps up.....- 1.52

16. Levelling means: Horizontal reference upper fuselage spar.

17. Minimum flight crew: 1 (pilot) at 0.47 ±0.05 m aft of datum

18. Maximum passenger seating capacity: 1 at 0.47 ±0.05 m and 2 at 1.25m aft of datum

19. Baggage / Cargo compartment 40 kg within the limits of weight and balance authorized.

Lever arm: + 2.1 m aft of datum

TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

Issue: 05 DR 253 B Date: 28 July 2023

20. Wheels and tires	Main gear track	
	Wheel tire size420 x 1	150
		e 3)
	PressureRefer to the maintenance man	ıual
	Front wheel movements (left and right): 25°	+2° -0°
21. Control surface movements:	Elevator:nose up: 13° +	0,5° -0°
	nose down: 6° +0	0,5° -0°
	Ailerons: up: 12° -	-0° 0,5°
	down: 12° -	-0° 0,5°
	Rudder:L & R: 25°	+0° -3°
	minimum before differential braking (L & R):	15°
	Wing Flaps: 1 st notch (T/O)15°	+0° -5°
	2 nd notch (Ldg)60°	+0° -5°
	Elevator tab:	
	Tab	

	Ta	ab
	move	ments
	Up	Down
Maximum "Nose up"	10°5	31°
Maximum "Nose down"	12°	3°

22. (Reserved)

M.IV Operating and Service Instructions

Airplane Flight Manual......Refer to latest amendment of service letter n°6 Airplane Maintenance Manual......Refer to latest amendment of service letter n°6 Airplane Major inspection schedule......Refer to latest amendment of service letter n°6

M.V Note:

1. Supplementary rear fuel tank operation:

Always use first 60 litres of fuel from rear main tank (more or less 1h30min) then transfer fuel from the supplementary tank to the rear main tank (by pulling the valve control located on the instrument panel).

Balance limits with all 4 tanks full, load is generally limited to either:

- 0 kg in luggage compartment (normal load on rear seats: 154 kg)

Or

- 100 kg on rear seats (40 kg in luggage compartment).

FOR ALL LOADING, USE THE LOADING GRAPH

2. This model is identical to the DR 253 except the leading edge profile of trapezoidal wing part.



DR 200, DR300, and DR400 series

Issue: 05 - DR 253 B Date: 28 July 2023

3. Alternative Oleo Strut and Tires:

TCDS No: EASA.A.367

Tires size 380x150 or 5.00-5 can only be installed on the aircraft if the DR400 landing gear has been installed in accordance with the SB 160403.



Issue: 05 - DR 340 Date: 28 July 2023

Section N: DR 340

N.I General

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 340

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: May 21, 1968

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

N.II <u>Certification Basis</u>

1. Reference Date for determining

the applicable requirements: 22 December 1967

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: None

N.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition Refer to the CEAPR document 1003349

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

4. Dimensions: Span8.72 m (28.61 ft)

5. Engines: Lycoming O-320-E2A

The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state acceptance or certification of this standard prior to 28



TCDS No: EASA.A.367

September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior

Date: 28 July 2023

to 28 September 2003 are also acceptable.

5.1 Engine Limits:

6. Propellers:

Issue: 05

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Sensenich	M 74 DMS-2-64 or M 74 DM-6S5-2-64	1.83 m (*)	2	2200 rpm
Sensenion	M 74 DMS-0-64 or M 74 DM-6S5-0-64	1.88 m	2	2200 rpm

Remarks: (*) No acceptable diameter reduction for repair. The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 80/87 octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD) grades	Mineral grades
All towns a material		grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:

Maii	n tank	RH wir	ng tank	LH wir	ng tank	Auxilia	ry tank
(lit	res)	(litr	es)	(litr	es)	(optiona	l) (litres)
Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable
75	65	40	40	40	40	50	50

Oil sump capacity 8 U.S. quarts (7.57 litres) 8.2 Oil:

9. Air speeds:

V_{NE}	295 km/h (159 knots IAS)
	260 km/h (140 knots IAS)
	260 km/h (140 knots IAS)
	200 km/h (108 knots IAS)
V_{FE}	

10. Maximum Operating Altitude: Refer to approved aircraft flight manual.

11. Operational Capability: Refer to approved aircraft flight manual.

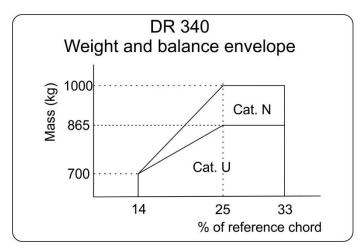
12. Maximum Masses:

"N" Category			"U" Category
	Take-off	Landing	
100	00 kg (2205 lb)	950 kg (2094 lb)	865 kg (1907 lb)



TCDS No: EASA.A.367 Issue: 05 DR 340

13. Centre of Gravity Range:



Date: 28 July 2023

Normal Category

Forward limit (14 % ref.): .. 0.240 m aft of datum at 700 kg Intermediate limit (25 % ref.):0.427 m aft of datum at 1000

Aft limit (33 % ref.): 0.564 m aft of datum at 1000 kg

Utility Category

Forward limit (14 % ref.): .. 0.240 m aft of datum at 700 kg Intermediate limit (25 % ref.):0.427 m aft of datum at 865

Aft limit (33 % ref.): 0.564 m aft of datum at 865 kg

14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor (n) at maximum weight: Normal Category:

Flaps up n	+ 3.8
Flaps up n	- 1.9
Flaps down n	+ 2
Flaps down n	

Utility Category:

Flaps up n	+ 4.4
Flaps up n	- 2.2
Flaps down n	+ 2
Flaps down n	0

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

- 18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of
- Maximum baggage compartment: 40 kg (88 lb) at 1.90m 19. Baggage / Cargo Compartment aft of datum

DR 200, DR300, and DR400 series

TCDS No: EASA.A.367 Issue: 05 DR 340 Date: 28 July 2023 Main gear track......2.58 m (8.46 ft) 20. Wheels and Tires Front gear angular movementleft: 27°right: 27° Tire pressure..... refer to the maintenance manual Oleo strut pressure refer to the maintenance manual 21. Control surface movements: Elevator:.....up 9°30′ -30′ down 12° -30' Relative to the trailing edge of the wings Ailerons: Refer to following table neutral up down 2°30' 11°10' 16°30' 15° 1°45' 9°45' Elevator tab:..... Elevator up: $25^{\circ}30' \pm 1^{\circ}$, $6^{\circ} \pm 1^{\circ}$ Elevator down: $10^{\circ}30' \pm 1^{\circ}$, $16^{\circ}30' \pm 1^{\circ}$ 2nd notch: 60° -5° Rudder: 25° -0° 22. (Reserved)

N.IV Operating and Service Instructions

Airplane Flight Manual......Refer to latest amendment of service letter n°6 Airplane Maintenance Manual......Refer to latest amendment of service letter n°6

N.V Note:

- 1. This plane is identical to DR315 except:
 - powerplant
 - addition of leading edge fuel tanks and 75 litres rear fuel
 - wings profile at rectangular part

lssue: 05 - DR 315 Date: 28 July 2023

Section O: DR 315

O.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 315

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: June 24, 1968

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

O.II Certification Basis

1. Reference Date for determining

the applicable requirements: 22 December 1967

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None
 EASA Environmental Standards: None

O.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition Refer to the CEAPR document 1003348

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

4. Dimensions:

5. Engines: Lycoming O-235-C2A or O-235-C2C

The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28



TCDS No: EASA.A.367 DR 315

> September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior

Date: 28 July 2023

to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power:

Propeller Manufacturer	Model	Maximum Continuous Power RPM
Evra	88-75-34 F	2800
	1 A 105 BCM 70-60	2600
Mac Cauley	1 A 105 BCM 70-56	2600
	1 B 90 ECM 72-50	2800

6. Propellers:

Issue: 05

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Evra	88-75-34 F	1.76 m	2	2250
	1 A 105 BCM 70-60	1.78 m	2	2250
Mac Cauley	1 A 105 BCM 70-56	1.78 m	2	2250
_	1 B 90 ECM 72-50	1.83 m	2	2300

The EASA type certification standard includes that of FAA TC P-918 and FAA TC P-842, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 80/87 octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070.

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
All temperature	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

Issue: 05 DR 315 Date: 28 July 2023

- 8. Fluid capacities:
 - 8.1 Fuel:

TCDS No: EASA.A.367

Main tank		Auxiliary tank	
(litres)		(optional) (litres)	
Capacity Usable		Capacity	Usable
110 100		50	50

Oil sump capacity 6 U.S. quarts (5.68 litres) 8.2 Oil:

9. Air speeds:

V_{NE}	295 km/h (159 knots IAS)
V_{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
V_A	200 km/h (108 knots IAS)
V_{FE}	170 km/h (92 knots IAS)

- 10. Maximum Operating Altitude: Refer to approved aircraft flight manual.
- 11. Operational Capability:

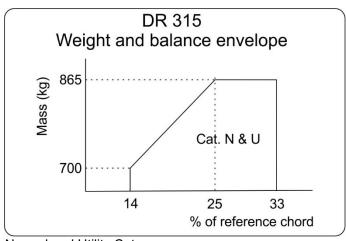
Refer to approved aircraft flight manual.

12. Maximum Masses:

14. Datum:

"N" Category		"U" Category
Take-off	Landing	
865 kg (1907 lb)	865 kg (1907 lb)	865 kg (1907 lb)

13. Centre of Gravity Range:



Normal and Utility Category

Forward limit (14 % ref.): .. 0.240 m aft of datum at 700 kg Intermediate limit (25 % ref.):0.427 m aft of datum at 865

Aft limit (33 % ref.): 0.564 m aft of datum at 865 kg

Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight: Normal Category:

Flaps up n	. + 3.8
Flaps up n	1.9
Flaps down n	. + 2
Flaps down n	
tility Category:	
Flancun	+ 1 1

Ut

+ 4.4
- 2.2
+ 2
0



TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

Issue: 05 - DR 315 Date: 28 July 2023

16. Levelling Means:	Horizontal reference upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 (maximum 120kg (265lb)) at 1.19m aft of datum.

19. Baggage / Cargo Compartment Maximum baggage compartment: 40 kg (88 lb) at 1.90m

aft of datum

20. Wheels and Tires

Main gear track	2.58 m (8.46 ft)
Wheel tire size3	880 x 150 or 500-5
Front gear angular movement	left: 27°
	right: 27°
Tire pressure refer to the ma	intenance manual
Oleo strut pressure refer to the ma	intenance manual

21. Control surface movements:

	+0°
Elevator:	up 9°30' -30'
	· +0°
	down 12° -30'

Ailerons:..... Relative to the trailing edge of the wings

up	neutral	down	
16°30'	2°30'	11°10'	
15°	1°45'	9°45'	

Elevator tab:

Elevator up:25°30' \pm 1° 6° \pm 1° Elevator down:10°30' \pm 1°16°30' \pm 1°

Flaps: 1st notch:15° \pm 5°

2nd notch:60° -5°

22. (Reserved)

O.IV Operating and Service Instructions

O.V Note:



Issue: 05 - DR 360 Date: 28 July 2023

Section P: DR 360

P.I General

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 360

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: July 19, 1968

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

P.II Certification Basis

1. Reference Date for determining

the applicable requirements: 22 December 1967

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None
 EASA Environmental Standards: None

P.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition Refer to the CEAPR document 1003350

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

4. Dimensions:

 Span
 8.72 m
 (28.61 ft)

 Height
 2.23 m
 (7.32 ft)

 Length
 7.08 m
 (23.23 ft)

 Wing Area
 14.20 m²
 (152.85 foot²)



Issue: 05 - DR 360 Date: 28 July 2023

5. Engines: Lycoming O-320-D2A

The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2700 rpm

6. Propellers:

TCDS No: EASA.A.367

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Sensenich	M 74 DMS-2-66 or M 74 DM-6S5-2-66	1.83 m (*)	2	2150 rpm

Remarks: (*) No acceptable diameter reduction for repair.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 91/96 or 100/130 octane minimum aviation grade

gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014

NO. 1014.	110. 1014.				
Air temperature	Ashless dispersant (AD)	Mineral			
All temperature	grades	grades			
All temperature	SAE15W50 or SAE20W50				
Above 80°F (+25°C)	SAE60	SAE60			
Above 60°F (+15°C)	SAE40 or SAE50	SAE50			
30°F to 90°F (O°C à +30°C)	SAE40	SAE40			
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30			
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50			
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20			

8. Fluid capacities:

8.1 Fuel:

Main tank		RH wir	ng tank	LH wing tank		Auxiliary tank	
(litres)		(litr	es)	(litres)		(optional) (litres)	
Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable
75	65	40	40	40	40	50	50

8.2 Oil: Oil sump capacity 8 U.S. quarts (7.57 litres)

 VNO
 260 km/h (140 knots IAS)

 Vc
 260 km/h (140 knots IAS)

 VA
 200 km/h (108 knots IAS)

 VFE
 170 km/h (92 knots IAS)

10. Maximum Operating Altitude: Refer to approved aircraft flight manual.

11. Operational Capability: Refer to approved aircraft flight manual.



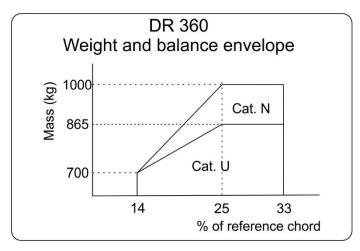
TCDS No: EASA.A.367

Issue: 05 - DR 360 Date: 28 July 2023

12. Maximum Masses:

"N" Ca	"U" Category	
Take-off	Landing	
1000 kg (2205 lb)	950 kg (2094 lb)	865 kg (1907 lb)

13. Centre of Gravity Range:



Normal Category

Forward limit (14 % ref.): .. 0.240 m aft of datum at 700 kg Intermediate limit (25 % ref.):0.427 m aft of datum at 1000 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 1000 kg

Utility category

Forward limit (14 % ref.): .. 0.240 m aft of datum at 700 kg Intermediate limit (25 % ref.):0.427 m aft of datum at 865 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 865 kg

14. Datum: Wing leading edge of the rectangular part of the wings Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

Normal Category:

Flaps up	n	+ 3.8
Flaps up	n	1.9
Flaps do	wn n	+ 2
Flaps do	wn n	0

Utility Category:

Flaps up n	+ 4.4
Flaps up n	2.2
Flaps down n	+ 2
Flaps down n	0

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.

19. Baggage / Cargo Compartment Maximum baggage compartment: 40 kg (88 lb) at 1.90m aft of datum

Issue: 05 - DR 360 Date: 28 July 2023

20. Wheels and Tires

TCDS No: EASA.A.367

21. Control surface movements

Elevator: up $9^{\circ}30^{,}_{-30}^{,0^{\circ}}$

down 12° +0° -30′

Ailerons: Relative to the trailing edge of the

wings.

 up
 neutral
 down

 16°30'
 2°30'
 11°10'

 15°
 1°45'
 9°45'

Elevator tab:

Elevator up:25°30' \pm 1° 6° \pm 1°

Elevator down:10°30' ± 1°16°30' ± 1°

Flaps: 1st notch:15 $^{\circ} \pm 5^{\circ}$

2nd notch: $60^{\circ} \, {}^{+0^{\circ}}_{-5^{\circ}}$

22. (Reserved)

P.IV Operating and Service Instructions

P.V Note:

1. This plane is identical to DR 340 except powerplant.

lssue: 05 - DR 380 Date: 28 July 2023

Section Q: DR 380

Q.I General

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 380

2. Airworthiness Category: Normal Category

Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: May 29, 1969

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

Q.II Certification Basis

1. Reference Date for determining

the applicable requirements: 22 December 1967

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

6. Requirements elected to comply: None
7. EASA Special Conditions: None
8. EASA Exemptions: None
9. EASA Equivalent Safety Findings: None
10. EASA Environmental Standards: None

Q.III Technical Characteristics and Operational Limitations

Type Design Definition Refer to the CEAPR document 1003350

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

4. Dimensions:

 Span
 8.72 m
 (28.61 ft)

 Height
 2.23 m
 (7.32 ft)

 Length
 7.08 m
 (23.23 ft)

 Wing Area
 14.20 m²
 (152.85 foot²)

5. Engines: Lycoming O-360-A3A

The EASA type certification standard includes that of FAA TC E-286, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS



TCDS No: EASA.A.367

standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Date: 28 July 2023

5.1 Engine Limits:

6. Propellers:

Issue: 05

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Sensenich	76 EM 8S5-0-64 76 EM 8S5-0-68	1.93 m (1)	2	2250 (2)

Remarks:

- (1) No acceptable diameter reduction for repair.
- (2) Do not continuous operate between 2025 rpm and

The EASA type certification standard includes that of FAA TC P4EA, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 100/100LL octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014.

1101 101 11					
Air temperature	Ashless dispersant (AD)	Mineral			
7 til temperature	grades	grades			
All temperature	SAE15W50 or SAE20W50				
Above 80°F (+25°C)	SAE60	SAE60			
Above 60°F (+15°C)	SAE40 or SAE50	SAE50			
30°F to 90°F (O°C à +30°C)	SAE40	SAE40			
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30			
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50			
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20			

8. Fluid capacities:

8.1 Fuel:

Main tank		RH tank		LH tank		Auxiliary tank	
(litres)		(litr	litres) (litres)		(optional) (litres)		
Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable
75	65	40	40	40	40	50	50

Oil sump capacity 8 U.S. quarts (7.57 litres) 8.2 Oil: Usable...... 6 U.S. quarts (5.68 litres)

9. Air speeds:

V_{NE}	305 km/h (165 knots IAS)
V_{NO}	270 km/h (146 knots IAS)
Vc	270 km/h (146 knots IAS)
V_A	200 km/h (108 knots IAS)
V_{FE}	170 km/h (92 knots IAS)

10. Maximum Operating Altitude: Refer to approved aircraft flight manual.

11. Operational Capability: Refer to approved aircraft flight manual.

12. Maximum Masses:

"N" Ca	tegory
Take-off	Landing

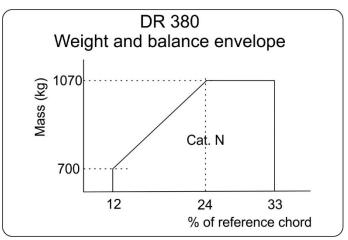


TCDS No: EASA.A.367

Issue: 05 - DR 380 Date: 28 July 2023

1070 kg (2359 lb) | 1020 kg (2249 lb)

13. Centre of Gravity Range:



Normal category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 700 kg Intermediate limit (24 % ref.):0.410 m aft of datum at 1070 kg $\,$

Aft limit (33 % ref.): 0.564 m aft of datum at 1070 kg

- 14. Datum: Wing leading edge of the rectangular part of the wings Cord length at reference section: 1.71 m (5.61 ft)
- 15. Load factor at maximum weight: Normal Category:

Flaps up n	+ 3.8
Flaps up n	1.9
Flaps down n	+ 2
Flaps down n	0

- 16. Levelling means: Horizontal reference upper fuselage spar
- 17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum
- 18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.
- 19. Baggage / Cargo Compartment Maximum baggage compartment: 60 kg (132 lb) at 1.90m aft of datum
- 20. Wheels and Tires

Main gear track	2.58 m (8.46 ft)
Wheel tire size	380 x 150 or 5.00-5
Front gear angular	movement left: 27°
	right: 27°
	refer to the maintenance manual
Oleo strut pressure	refer to the maintenance manual

21. Control surface movements

	+0°
Elevator:	up 9°30' -30'
	+0°
	down 12° -30'
Ailerons: wings	Relative to the trailing edge of the

up 16°30

15°

Elevator up:25°30' \pm 1° 6° \pm 1°

neutral

2°30'

1°45'

down

11°10

9°45'



Elevator tab:

TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

Issue: 05 - DR 380 Date: 28 July 2023

22. (Reserved)

Q.IV Operating and Service Instructions

Q.V Note:

- 1. This plane is identical to DR 340 except:
 - Powerplant
 - Structure
 - Landing gears

lssue: 05 - DR 300/108 Date: 28 July 2023

Section R: <u>DR 300/108</u>

R.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 300/108

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: June 18, 1970

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

R.II <u>Certification Basis</u>

1. Reference Date for determining

the applicable requirements: 22 December 1967

(Reserved)
 (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 6.

R.III <u>Technical Characteristics and Operational Limitations</u>

Type Design Definition Refer to the CEAPR document 1003347

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

4. Dimensions:

 Span
 8.72 m
 (28.61 ft)

 Height
 2.23 m
 (7.32 ft)

 Length
 6.96 m
 (22.83 ft)

 Wing Area
 13.60 m²
 (146.39 foot²)

5. Engines: Lycoming O-235-C2A or Lycoming O-235-C2C

The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS



Issue: 05 - DR 300/108 Date: 28 July 2023

standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2600 rpm

Remark: Maximum continuous power limited by noise

regulation.

6. Propellers:

Manufactur er	Model	Ø	Number of blades	Minimum static RPM at sea level
	1 A 105 BCM 70-56	1.78 m	2	
Mac Cauley	1 A 105 BCM 70-60	1.78 m	2	2250 rpm (*)
	1 B 90 ECM 72-50	1.83 m	2	2250 rpm (*)
Evra	88-75-34 F	1.76 m	2	

Remarks: (*) Maximum authorized RPM: 2600 rpm
The EASA type certification standard includes that of FAA
TC P-918 and FAA TC P-842, based on individual EU
member state acceptance or certification of this standard
prior to 28 September 2003. Other standards confirming to
TC/TCDS standards certificated by individual EU member
state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 80/87 octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral		
All temperature	grades	grades		
All temperature	SAE15W50 or SAE20W50			
Above 80°F (+25°C)	SAE60	SAE60		
Above 60°F (+15°C)	SAE40 or SAE50	SAE50		
30°F to 90°F (O°C à +30°C)	SAE40	SAE40		
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30		
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50		
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20		

8. Fluid capacities:

8.1 Fuel:

Main	tank	Auxiliary tank		
(lit	res)	(optional) (litres)		
Capacity	Usable	Capacity	Usable	
110	100	50	50	

8.2 Oil: Oil sump capacity 6 U.S. quarts (5.68 litres)

9. Air speeds:

V_{NE}295 km/h (159 knots IAS) V_{NO}260 km/h (140 knots IAS) V_C......260 km/h (140 knots IAS) V_A200 km/h (108 knots IAS) V_{FE}170 km/h (92 knots IAS)

Date: 28 July 2023

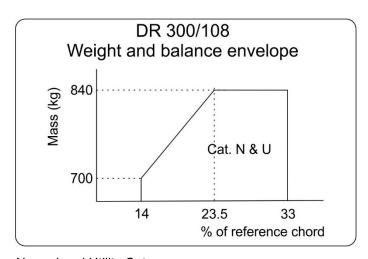
Refer to approved aircraft flight manual. 10. Maximum Operating Altitude:

11. Operational Capability: Refer to approved aircraft flight manual.

12. Maximum Masses:

"N" Ca	"U" Category	
Take-off Landing		
840 kg (1852 lb)	840 kg (1852 lb)	840 kg (1852 lb)

13. Centre of Gravity Range:



Normal and Utility Category

Forward limit (14 % ref.): .. 0.240 m aft of datum at 700 kg Intermediate limit (23.5 % ref.):0.401 m aft of datum at 840

Aft limit (33 % ref.): 0.564 m aft of datum at 840 kg

Wing leading edge of the rectangular part of the wings. 14. Datum: Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

Normal Category: Flaps up n + 3.8 Flaps up n - 1.9 Flaps down n..... + 2 Flaps down n.....0 **Utility Category:** Flaps up n + 4.4 Flaps up n - 2.2

Flaps down n..... + 2 Flaps down n.....0

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 (maximum 100kg (220lb)) at 1.19m aft of datum.

19. Baggage / Cargo Compartment: Not applicable DR 200, DR300, and DR400 series

Issue: 05 - DR 300/108 Date: 28 July 2023

20. Wheels and Tires:

TCDS No: EASA.A.367

21. Control surface movements:

wings

 up
 neutral
 down

 16°30'
 2°30'
 11°10'

 15°
 1°45'
 9°45'

Elevator tab: Elevator up:.....25°30' \pm 1°......6° \pm 1°

Elevator down:10°30' ± 1° 16°30' ± 1°

Flaps: 1st notch:15° \pm 5°

2nd notch: $60^{\circ} \, {}^{+0^{\circ}}_{-5^{\circ}}$

22. (Reserved)

R.IV Operating and Service Instructions

R.V Note:

1. This plane is identical to DR 315 except cabin layout



lssue: 05 - DR 300/180 R Date: 28 July 2023

Section S: <u>DR 300/180 R</u>

S.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 300/180 R

2. Airworthiness Category: Normal Category

3. Type Certificate Holder: Refer to Note 2 Section PP4. Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: July 24, 1970

7. EASA Type Certification date: January28, 2013 (Type Certificate transfer)

8. The EASA Type Certificates replaces DGAC-France Type Certificate no. 45

S.II Certification Basis

1. Reference Date for determining

the applicable requirements: 22 December 1967

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None
 EASA Environmental Standards: None

S.III Technical Characteristics and Operational Limitations

1. Type Design Definition Refer to the CEAPR document 1003347

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

4. Dimensions:

 Span
 8.72 m
 (28.61 ft)

 Height
 2.23 m
 (7.32 ft)

 Length
 7.08 m
 (23.23 ft)

 Wing Area
 13.60 m²
 (146.39 foot²)

5. Engines: Lycoming O-360-A3A

The EASA type certification standard includes that of FAA TC E-286, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS



DR 200, DR300, and DR400 series

Issue: 05 - DR 300/180 R

standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Date: 28 July 2023

5.1 Engine Limits: Maximum Continuous Power: 2700 rpm

6. Propellers:

TCDS No: EASA.A.367

Manufactur	Model	a	Number	Minimum static RPM
er		Ø	of blades	at sea level
	76 EM 8S5-058		2	2450 (2)
Sensenich	76 EM 8S5-064	1.93 m (1)	2	2250 (2)
	76 EM 8S5-054		2	2500 (2)
Hoffmann	HO4-27HM-170-128	1.70 m	4	2240 (2)

Remarks:

- (1) No acceptable diameter reduction for repair.
- (2) Do not continuous operate between 2150 rpm and 2350 rpm.

The EASA type certification standard includes that of FAA TC P4EA and P6NE, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 100/100LL octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming No. 1014.

Ashless dispersant (AD) Mineral Air temperature grades grades All temperature SAE15W50 or SAE20W50 Above 80°F (+25°C) SAE60 SAE60 Above 60°F (+15°C) SAE40 or SAE50 SAE50 30°F to 90°F (O°C à +30°C) SAE40 SAE40 0°F to 70°F (-15°C à +20°C) SAE30, SAE40 or SAE20W40 SAE30 0°F to 90°F (-15°C à +30°C) SAE20W50 or SAE15W50 **SAE20W50** Below 10°F (-10°C) SAE30 or SAE20W30 SAE20

8. Fluid capacities:

8.1 Fuel:

Maiı	n tank	Auxilia	ry tank
(lit	tres)	(optiona	I) (litres)
Capacity	Usable	Capacity	Usable
110	100	50	50



Issue: 05 DR 300/180 R Date: 28 July 2023

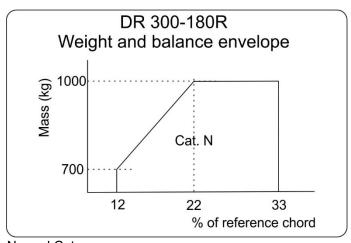
9. Air speeds: V_{NE}305 km/h (165 knots IAS) V_{NO}270 km/h (146 knots IAS) V_C......270 km/h (146 knots IAS) V_A200 km/h (108 knots IAS)

10. Maximum Operating Altitude: Refer to approved aircraft flight manual. 11. Operational Capability: Refer to approved aircraft flight manual.

12. Maximum Masses:

"N" Category		
Take-off Landing		
1000 kg (2205 lb)	950 kg (2094 lb)	

13. Centre of Gravity Range:



V_{FE}170 km/h (92 knots IAS)

Normal Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 700 kg Intermediate limit (22 % ref.):0.376 m aft of datum at 1000

Aft limit (33 % ref.): 0.564 m aft of datum at 1000 kg

14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

19. Baggage / Cargo Compartment

Normal Category: Flaps up n+ 3.8 Flaps up n - 1.9 Flaps down n..... + 2 Flaps down n.....0

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.

Maximum baggage compartment: 60 kg (132 lb) at 1.90m

aft of datum

TCDS No: EASA.A.367

20. Wheels and Tires

Issue: 05

Main gear track2.58 m (8.46 ft) Wheel tire size380 x 150 or 5.00-5 Front gear angular movement left: 27°right: 27°

Tire pressure refer to the maintenance manual Oleo strut pressure refer to the maintenance manual

21. Control surface movements

up 9°30' -30' Elevator:

> down 12° -30' Relative to the trailing edge of the

Ailerons: wings

neutral down up 2°30' 11°10' 16°30 15° 1°45' 9°45'

Date: 28 July 2023

Elevator up:25°30' ± 1°......6° ± 1° Elevator tab:

Elevator down: 10°30' ± 1°16°30' ± 1°

Flaps: 1st notch:15° ± 5°

2nd notch: 60° -5°

22. (Reserved)

S.IV Operating and Service Instructions

Airplane Flight Manual...... Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Manual.................................. Refer to the latest amendment of Service Letter no. 6

S.V Note:

1. This plane is identical to DR 315 except:

- powerplant

- structure

- landing gears

- towing ability

2. Glider and banner towing Refer to approved flight manual

Takeoff maximum mass: 720kg Seaplane maximum mass towed: 550kg

Propeller approved for these operations: Sensenich 76 EM 8S5-058

Hoffmann H04-27HM-170-128

Issue: 05 - DR 300/140 Date: 28 July 2023

Section T: <u>DR 300/140</u>

T.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 300/140

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: February 22, 1971

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

T.II <u>Certification Basis</u>

1. Reference Date for determining

the applicable requirements: 22 December 1967

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None
 EASA Environmental Standards: None

T.III Technical Characteristics and Operational Limitations

1. Type Design Definition Refer to the CEAPR document 1003347

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

4. Dimensions:

 Span
 8.72 m
 (28.61 ft)

 Height
 2.23 m
 (7.32 ft)

 Length
 6.96 m
 (22.83 ft)

 Wing Area
 13.60 m²
 (146.39 foot²)

5. Engines: Lycoming O-320-E2A

The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS



standards certificated by individual EU member state prior

Date: 28 July 2023

to 28 September 2003 are also acceptable.

2700 rpm 5.1 Engine Limits: Maximum continuous power:

6. Propellers:

Manufactur	Model	a	Number	Minimum static RPM
er	iviodei	Ø	of blades	at sea level
	M 74 DMS-2-64	1 02 m (*)	2	- 2200 rpm
Concenials	74 DM 6S5-2-64	1.83 m (*)	2	
Sensenich	M 74 DMS-0-64	1.88 m	2	
	74 DM 6S5-0-64		2	

Remarks: (*) No acceptable diameter reduction for repair.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 80/87 octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral		
7 iii temperatare	grades	grades		
All temperature	SAE15W50 or SAE20W50			
Above 80°F (+25°C)	SAE60	SAE60		
Above 60°F (+15°C)	SAE40 or SAE50	SAE50		
30°F to 90°F (O°C à +30°C)	SAE40	SAE40		
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30		
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50		
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20		

8. Fluid capacities:

8.1 Fuel:

Main tank (litres)		Auxiliary tank (optional) (litres)	
Capacity	Usable	Capacity	Usable
110	100	50	50

8.2 Oil: Oil sump capacity 8 U.S. quarts (7.57 litres)

9. Air speeds:

V_{NE}	295 km/h (159 knots IAS)
V_{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
V_A	200 km/h (108 knots IAS)
V_{FE}	170 km/h (92 knots IAS)

Refer to approved aircraft flight manual. 10. Maximum Operating Altitude:

11. Operational Capability: Refer to approved aircraft flight manual.

12. Maximum Masses:

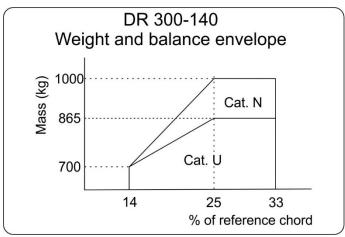
"N" Cat	"U" Category	
Take-off Landing		
1000 kg (2205 lb)	950 kg (2094 lb)	865 kg (1907 lb)



Issue: 05

DR 300/140 Date: 28 July 2023

13. Centre of Gravity Range:



Normal Category

Forward limit (14 % ref.): .. 0.240 m aft of datum at 700 kg Intermediate limit (25 % ref.):0.427 m aft of datum at 1000 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 1000 kg

Utility Category

Forward limit (14 % ref.): .. 0.240 m aft of datum at 700 kg Intermediate limit (25 % ref.):0.427 m aft of datum at 865 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 865 kg

Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

14. Datum:

Normal Category:	Flaps up n	+ 3.8
	Flaps up n	1.9
	Flaps down n	
	Flaps down n	
Utility Category:	Flaps up n	+ 4.4
	Flaps up n	2.2
	Flaps down n	
	Flaps down n	0

- 16. Levelling Means: Horizontal reference upper fuselage spar
- 17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum
- 18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.
- 19. Baggage / Cargo Compartment Maximum baggage compartment: 40 kg (88 lb) at 1.90m aft of datum
- 20. Wheels and Tires

Main gear track	2.58 m (8.46 ft)
	380 x 150 or 5.00-5
Front gear angular	movement left: 27°
	right: 27°
Tire pressure	refer to the maintenance manual
Oleo strut pressure	refer to the maintenance manual

21. Control surface movements

DR 200, DR300, and DR400 series

Issue: 05 - DR 300/140 Date: 28 July 2023

wings

 up
 neutral
 down

 16°30'
 2°30'
 11°10'

 15°
 1°45'
 9°45'

Elevator tab: Elevator up:25°30' \pm 1°......6° \pm 1°

Elevator down: 10°30' ± 1° 16°30' ± 1°

Flaps: 1st notch:15° \pm 5°

2nd notch: 60° -5°

22. (Reserved)

TCDS No: EASA.A.367

T.IV Operating and Service Instructions

T.V Note:

1. This plane is identical to DR 315 except powerplant



Issue: 05 - DR 300/125 Date: 28 July 2023

Section U: <u>DR 300/125</u>

U.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 300/125

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: May 11, 1971

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

U.II Certification Basis

1. Reference Date for determining

the applicable requirements: 22 December 1967

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None
 EASA Environmental Standards: None

U.III Technical Characteristics and Operational Limitations

1. Type Design Definition Refer to the CEAPR document 1003347

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

4. Dimensions:

5. Engines: Lycoming O-235-F2B or O-235-F2A or O-235-J2A

The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS



> standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Date: 28 July 2023

5.1 Engine Limits: Maximum Continuous Power: 2800 rpm

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Mac Cauley	1 A 135 JCM 71-54	1.80 m	2	2300 rpm (*)

Remarks: (*) Do not continuous operate between 2025 rpm and 2325 rpm.

The EASA type certification standard includes that of FAA TC P-842, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 100/100LL octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
All temperature	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:

Main tank		Auxiliary tank	
(litres)		(optiona	I) (litres)
Capacity	Usable	Capacity	Usable
110	100	50	50

Oil sump capacity 6 U.S. quarts (5.68 litres) 8.2 Oil:

9. Air speeds:

V_{NE}	295 km/h (159 knots IAS)
V_{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
V_A	200 km/h (108 knots IAS)
	170 km/h `(92 knots IAS)

10. Maximum Operating Altitude: Refer to approved aircraft flight manual.

11. Operational Capability: Refer to approved aircraft flight manual.

12. Maximum Masses:

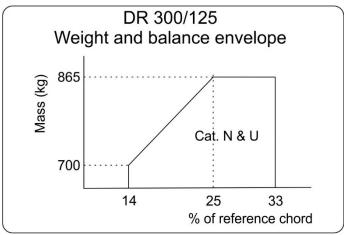
"N" Category		"U" Category
Take-off	Landing	
865 kg (1907 lb)	865 kg (1907 lb)	865 kg (1907 lb)

13. Centre of Gravity Range:



TCDS No: EASA.A.367

lssue: 05 - DR 300/125 Date: 28 July 2023



Normal and Utility Category

Forward limit (14 % ref.): .. 0.240 m aft of datum at 700 kg Intermediate limit (25 % ref.):0.427 m aft of datum at 865 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 865 kg

14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

Normal Category:	Flaps up nFlaps up n	
	Flaps down n	
	Flaps down n	0
Utility Category:	Flaps up n	+ 4.4
	Flaps up n	2.2
	Flaps down n	+ 2
	Flaps down n	0

- 16. Levelling Means: Horizontal reference upper fuselage spar
- 17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum
- 18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 (maximum 120kg (265lb)) at 1.19m aft of datum.
- 19. Baggage / Cargo Compartment Maximum baggage compartment: 40 kg (88 lb) at 1.90m aft of datum

DR 200, DR300, and DR400 series

TCDS No: EASA.A.367 Issue: 05 DR 300/125 Date: 28 July 2023 20. Wheels and Tires Main gear track2.58 m (8.46 ft) Wheel tire size380 x 150 or 5.00-5 Front gear angular movement left: 27°right: 27° Tire pressure refer to the maintenance manual Oleo strut pressure refer to the maintenance manual 21. Control surface movements Elevator: down 12° -30' Ailerons: Relative to the trailing edge of the wings neutral up down 16°30 2°30' 11°10' 15° 1°45' 9°45' Elevator up:25°30' ± 1°......6° ± 1° Elevator tab: Elevator down: $10^{\circ}30' \pm 1^{\circ}16^{\circ}30' \pm 1^{\circ}$ Flaps: 1st notch:15° \pm 5° 22. (Reserved)

U.IV Operating and Service Instructions

Airplane Flight Manual...... Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Manual...... Refer to the latest amendment of Service Letter no. 6

U.V Note:

1. This plane is identical to DR 315 except powerplant



lssue: 05 - DR 300/120 Date: 28 July 2023

Section V: <u>DR 300/120</u>

V.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 300/120

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: February 11, 1975

7. EASA Type Certification date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

V.II Certification Basis

1. Reference Date for determining

the applicable requirements: 3 February 1975

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

Requirements elected to comply: None
 EASA Special Conditions: None
 EASA Exemptions: None
 EASA Equivalent Safety Findings: None
 EASA Environmental Standards: None

V.III Technical Characteristics and Operational Limitations

1. Type Design Definition Refer to the CEAPR document 1003347

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

4. Dimensions:

 Span
 8.72 m
 (28.61 ft)

 Height
 2.23 m
 (7.32 ft)

 Length
 6.96 m
 (22.83 ft)

 Wing Area
 13.60 m²
 (146.39 foot²)

5. Engines: Lycoming O-235-L2A or O-235-K2A or O-235-K2B

The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS



> standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Date: 28 July 2023

5.1 Engine Limits: Maximum Continuous Power: 2800 rpm

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Mac Cauley	1 A 135 JCM 71-47	1.80 m 1.77 m (1)	2	2200 (2)
Hoffmann	HO 14-178/115	1.78 m 1.73 m (1)	2	2250
Sensenich	72 CKS6-0-56	1.83 m (3)	2	2220
Sensemen	72 CKS5-0-56	1.03 111 (3)	2	2220

Remarks:

- (1) Minimum diameter after repair.
- (2) Do not continuous operate between 2025 rpm and 2325 rpm.
- (3) No acceptable diameter reduction for repair.

The EASA type certification standard includes that of FAA TC P-842 and FAA TC P-904, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids: 7.1 Fuel:

100/100LL octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
All temperature	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

- 8. Fluid capacities:
 - 8.1 Fuel:

Main tank		Auxiliary tank	
(litres)		(optional) (litres)	
Capacity	Usable	Capacity	Usable
110	100	50	50

8.2 Oil: Oil sump capacity 6 U.S. quarts (5.68 litres)

9. Air speeds:

V_{NE}	295 km/h (159 knots IAS)
V_{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
V_A	200 km/h (108 knots IAS)
V_{FE}	170 km/h (92 knots IAS)

Refer to approved aircraft flight manual. 10. Maximum Operating Altitude:

11. Operational Capability:

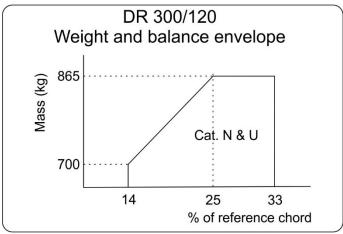
Refer to approved aircraft flight manual.

12. Maximum Masses:

"N" Ca	"U" Category	
Take-off Landing		
865 kg (1907 lb)	865 kg (1907 lb)	865 kg (1907 lb)

Date: 28 July 2023

13. Centre of Gravity Range:



Normal and Utility Category

Forward limit (14 % ref.): .. 0.240 m aft of datum at 700 kg Intermediate limit (25 % ref.):0.427 m aft of datum at 865

Aft limit (33 % ref.): 0.564 m aft of datum at 865 kg

14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

Normal Category: Flaps up n + 3.8 Flaps up n - 1.9 Flaps down n..... + 2 Flaps down n.....0 **Utility Category:** Flaps up n + 4.4

Flaps up n - 2.2 Flaps down n..... + 2 Flaps down n.....0

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 (maximum 120kg (265lb)) at 1.19m aft of datum.

19. Baggage / Cargo Compartment

Maximum baggage compartment: 40 kg (88 lb) at 1.90m

aft of datum

20. Wheels and Tires

Main gear track	2.58 m (8.46 ft)
Wheel tire size	380 x 150 or 500-5
Front gear angular	movement left: 27°
	right: 27°
Tire pressure	refer to the maintenance manual
Oleo strut pressure	e refer to the maintenance manual

21. Control surface movements



DR 200, DR300, and DR400 series

lssue: 05 - DR 300/120 Date: 28 July 2023

down 12° -30'

Ailerons: Relative to the trailing edge of the

wings

 up
 neutral
 down

 16°30'
 2°30'
 11°10'

 15°
 1°45'
 9°45'

Elevator tab: Elevator up:.....25°30' \pm 1°......6° \pm 1°

Elevator down: $10^{\circ}30' \pm 1^{\circ}16^{\circ}30' \pm 1^{\circ}$

Flaps: 1st notch:15° \pm 5°

2nd notch: $60^{\circ} - 5^{\circ}$

Rudder: $25^{\circ}_{-0^{\circ}}^{+3^{\circ}}$

22. (Reserved)

TCDS No: EASA.A.367

V.IV Operating and Service Instructions

V.V Note:

1. This plane is identical to DR 315 except powerplant

lssue: 05 - DR 400/125 Date: 28 July 2023

Section W: <u>DR 400/125</u>

W.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 400/125

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: May 10, 1972

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

W.II Certification Basis

1. Reference Date for determining

the applicable requirements: 21 March 1971

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

6. Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

8. EASA Exemptions: None9. EASA Equivalent Safety Findings: None10. EASA Environmental Standards: None

W.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition: Refer to CEAPR document n°1001131

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed...

4. Dimensions:

5. Engines: Lycoming O-235-F2B or O-235-J2A

The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28



> September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Date: 28 July 2023

Maximum Continuous Power: 5.1 Engine Limits: 2800 rpm

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Mac Cauley	1 A 135 JCM 71-54	1.80 m	2	2300 rpm (*)

Remarks: (*) Do not continuous operate between 2025 rpm and 2325 rpm.

The EASA type certification standard includes that of FAA TC P-842, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

100/100LL octane minimum aviation grade gasoline. 7.1 Fuel:

Refer to latest revision of Service Instruction Lycoming

No. 1070.

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014.

Air temperature	Ashless dispersant (AD) grades	Mineral grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:

Main tank		Auxiliary tank		
(litres)		(optional) (litres)		
Ca	Capacity Usable		Capacity	Usable
	110	100	50	50

Oil sump capacity 6 U.S. quarts (5.68 litres) 8.2 Oil:

9. Air speeds:

V_{NO}260 km/h (140 knots IAS) V_C......260 km/h (140 knots IAS) V_A215 km/h (116 knots IAS) V_{FE}170 km/h (92 knots IAS)

10. Maximum Operating Altitude: Refer to approved aircraft flight manual.

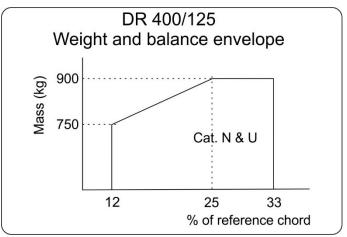
11. Operational Capability: Refer to approved aircraft flight manual.

12. Maximum Masses:

"N" Cat	"U" Category	
Take-off	Landing	
900 kg (1984 lb)	900 kg (1984 lb)	900 kg (1984 lb)



13. Centre of Gravity Range:



Date: 28 July 2023

Normal and Utility Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 900

Aft limit (33 % ref.): 0.564 m aft of datum at 900 kg

14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

Normal Category: Flaps up n + 3.8 Flaps up n - 1.9

Flaps down n..... + 2 Flaps down n.....0

Category Utility: Flaps up n + 4.4

Flaps up n - 2.2 Flaps down n..... + 2 Flaps down n.....0

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.

19. Baggage / Cargo Compartment Maximum baggage compartment: 40 kg (88 lb) at 1.90m aft of datum

20. Wheels and Tires:

Main gear track2.58 m (8.46 ft) Wheel tire size Front gear angular movement left: 27°right: 27° Tire pressure refer to the maintenance manual Oleo strut pressure refer to the maintenance manual

21. Control surface movements

up 9°30' ± 30' Elevator: down 12° ± 30' Relative to the trailing edge of the

Ailerons: wings

> neutral down up 15°± 1° 2° ± 1° 10°± 1°

Date: 28 July 2023

Elevator up:25°30' ± 1°......6° ± 1° Elevator tab:

Elevator down: $10^{\circ}30' \pm 1^{\circ}16^{\circ}30' \pm 1^{\circ}$ 1st notch:15° ± 5° Flaps:

2nd notch: 60° -5°

(1) For planes fitted with brakes controlled with rudder pedals:

16° (-0°, +2°) before operating drum brakes 20° (-0°, +3°) before operating disk brakes

22. (Reserved)

W.IV Operating and Service Instructions

Airplane Flight Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	. Refer to the latest amendment of Service Letter no. 6

W.V Note:

- 1. This plane is identical to DR 300/180R except:
 - powerplant
 - centre and front parts of the fuselage
 - forward sliding canopy
 - fuel circuit

lssue: 05 - DR 400/140 Date: 28 July 2023

Section X: <u>DR 400/140</u>

X.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 400/140

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: December 01, 1972

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

X.II Certification Basis

1. Reference Date for determining

the applicable requirements: 21 March 1971

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

6. Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

8. EASA Exemptions: None9. EASA Equivalent Safety Findings: None10. EASA Environmental Standards: None

X.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition: Refer to CEAPR document n°1001131

Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

4. Dimensions:

 Span
 8.72 m
 (28.61 ft)

 Height
 2.23 m
 (7.32 ft)

 Length
 6.96 m
 (22.83 ft)

 Wing Area
 13.60 m²
 (146.39 foot²)

5. Engines: Lycoming O-320-E2A

TCDS No: EASA.A.367

Issue: 05 DR 400/140 Date: 28 July 2023

> The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2700 rpm

6. Propellers:

Manufacturer	Model	Ø	Number	Minimum static RPM
Manuacturer	Model		of blades	at sea level
	M 74 DMS-2-64	1.83 m (*)	2	2200 rpm
Conconich	74 DM 6S5-2-64		2	
Sensenich	M 74 DMS-0-64	1 00	2	
	74 DM 6S5-0-64	1.88 m	2	

Remarks: (*) No acceptable diameter reduction for repair.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 80/87 octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014.

Air temperature	Ashless dispersant (AD) grades	Mineral grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:

Main tank (litres)		Auxiliary tank (optional) (litres)		
Capacity	Usable	Capacity	Usable	
110	100	50	50	

8.2 Oil: Oil sump capacity 8 U.S. quarts (7.57 litres) Usable...... 6 U.S. quarts (5.68 litres)

9. Air speeds:

V_{NE}	
V_{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
V_A	215 km/h (116 knots IAS)
V_{FE}	170 km/h (92 knots IAS)

10. Maximum Operating Altitude: Refer to approved aircraft flight manual.



11. Operational Capability:

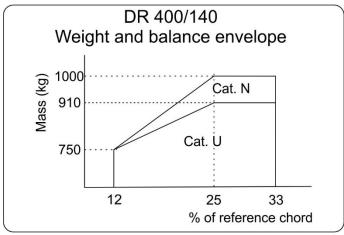
Refer to approved aircraft flight manual.

12. Maximum Masses:

"N" Cat	"U" Category	
Take-off	Landing	
1000 kg (2205 lb)	1000 kg (2205 lb)	910 kg (2006 lb)

Date: 28 July 2023

13. Centre of Gravity Range:



Normal Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 1000

Aft limit (33 % ref.): 0.564 m aft of datum at 1000 kg **Utility Category**

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 910 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 910 kg

14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

Normal Category:	Flaps up n	+ 3.8
	Flaps up n	1.9
	Flaps down n	+ 2
	Flaps down n	
Utility Category:	Flaps up n	+ 4.4
	Flaps up n	2.2
	Flaps down n	+ 2
	Flaps down n	0

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.

19. Baggage / Cargo Compartment Maximum baggage compartment: 40 kg (88 lb) at 1.90m aft of datum

20. Wheels and Tires

Main gear track2.58 m (8.46 ft) Wheel tire size Front gear angular movementleft: 27°



DR 200, DR300, and DR400 series

Issue: 05 - DR 400/140 Date: 28 July 2023

Tire pressure refer to the maintenance manual Oleo strut pressure refer to the maintenance manual

21. Control surface movements

TCDS No: EASA.A.367

Ailerons: Relative to the trailing edge of the

wings

 up
 neutral
 down

 15°± 1°
 2° ± 1°
 10°± 1°

Elevator tab: Elevator up:25°30' \pm 1°......6° \pm 1°

Elevator down: 10°30' ± 1°16°30' ± 1°

Flaps: 1st notch:15° \pm 5°

2nd notch: 60° -5°

Rudder:25° ^{+3°} _{-0°} (1)

(1) For planes fitted with brakes controlled with rudder pedals:

16° (-0°, +2°) before operating drum brakes 20° (-0°, +3°) before operating disk brakes

22. (Reserved)

X.IV Operating and Service Instructions

Airplane Flight Manual	Refer to the latest amendment of Service Letter no.	6
Airplane Maintenance Manual	Refer to the latest amendment of Service Letter no.	6
Airplane Maintenance Schedule	Refer to the latest amendment of Service Letter no.	6

X.V Note:

1. This plane is identical to DR 400/125 except powerplant



lssue: 05 - DR 400/160 Date: 28 July 2023

Section Y: <u>DR 400/160</u>

Y.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 400/160

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: September 06, 1972

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

Y.II Certification Basis

1. Reference Date for determining

the applicable requirements: 21 March 1971

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

6. Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

8. EASA Exemptions: None9. EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 6.

Y.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition: Refer to CEAPR document n°1001131 for DR400

"STANDARD 88" (Refer to note 2)

Refer to CEAPR document n°1001130 for DR400

"STANDARD 92" (Refer to note 2)

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed...

4. Dimensions: Span8.72 m (28.61 ft)

Height2.23 m (7.32 ft)

Wing Area 14.20 m² (152.85 foot²)

5. Engines: Lycoming O-320-D2A

> The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior

Date: 28 July 2023

to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power:

Remarks: Maximum continuous power limited by noise

regulation.

6. Propellers:

Manufactur er	Model	Ø	Number of blades	Minimum static RPM at sea level
Conconich	74 DM 6S5-2-66	1 83 m (*)	2	2150 rpm
Sensenich	74 DM 6S5-2-64	1.83 m (*)	2	2250 rpm

Remarks: (*) No acceptable diameter reduction for repair.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 91/96 or 100/130 octane minimum aviation grade

Refer to latest revision of Service Instruction Lycoming

No. 1070

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014.

110.1011.					
Air temperature	Ashless dispersant (AD)	Mineral			
All temperature	grades	grades			
All temperature	SAE15W50 or SAE20W50				
Above 80°F (+25°C)	SAE60	SAE60			
Above 60°F (+15°C)	SAE40 or SAE50	SAE50			
30°F to 90°F (O°C à +30°C)	SAE40	SAE40			
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30			
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50			
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20			

8. Fluid capacities:

8.1 Fuel:

Maii	n tank	RH tank		LH tank		Auxiliary tank	
(lit	res)	(litr	es)	(litr	es)	(optiona	l) (litres)
Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable
110	100/109 (1)	40	40	40	40	50	50

(1) New standard called "Standard 92" from serial no. 2210, unusable quantity of fuel reduced from 10 litres to 1 litre, (Refer to note 2).

8.2 Oil: Oil sump capacity 8 U.S. quarts (7.57 litres)

Usable...... 6 U.S. quarts (5.68 litres)



9. Air speeds:

V_{NO}260 km/h (140 knots IAS) V_C260 km/h (140 knots IAS) V_A215 km/h (116 knots IAS) V_{FE}170 km/h (92 knots IAS)

Date: 28 July 2023

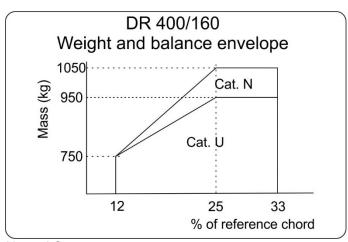
Refer to approved aircraft flight manual. 10. Maximum Operating Altitude:

11. Operational Capability: Refer to approved aircraft flight manual.

12. Maximum Masses:

"N" Ca	"U" Category	
Take-off	Landing	
1050 kg (2315 lb)	1045 kg (2304 lb)	950 kg (2094 lb)

13. Centre of Gravity Range:



Normal Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 1050

Aft limit (33 % ref.): 0.564 m aft of datum at 1050 kg **Utility Category**

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 950

Aft limit (33 % ref.): 0.564 m aft of datum at 950 kg

14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

Normal Category:	Flaps up n Flaps up n Flaps down n Flaps down n	1.9 + 2
Utility Category:	Flaps up n Flaps up n Flaps down n Flaps down n	2.2 + 2

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.

DR 200, DR300, and DR400 series

lssue: 05 - DR 400/160 Date: 28 July 2023

19. Baggage / Cargo Compartment

Maximum baggage compartment: 40 kg (88 lb) at 1.90m aft of datum

20. Wheels and Tires

TCDS No: EASA.A.367

2.58 m (8.46 ft)
380 x 150 or 5.00-5
movement left: 27°
right: 27°
refer to the maintenance manual
e refer to the maintenance manual

21. Control surface movements

Ailerons: Relative to the trailing edge of the

wings

up	neutral	down
15°± 1°	2° ± 1°	10°± 1°

Elevator tab: Elevator up:25°30' \pm 1°......6° \pm 1°

Elevator down: 10°30' ± 1°16°30' ± 1° 1st notch:15° ± 5°

Rudder:25° ^{+3°}_{-0°} (1)

(1) For planes fitted with brakes controlled with rudder pedals:

16° (-0°, +2°) before operating drum brakes 20° (-0°, +3°) before operating disk brakes

22. (Reserved)

Y.IV Operating and Service Instructions

Airplane Flight Manual Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance Manual. Refer to the latest amendment of Service Letter no. 6 Airplane Maintenance ScheduleRefer to the latest amendment of Service Letter no. 6

Y.V Note:

- 1. This plane is identical to DR 400/125 except:
 - powerplant
 - leading edge of centre part of the wings
 - leading edge fuel tanks
 - luggage compartment door
- 2. "Standard 92" models: Since February 1994 (from serial nr. 2220 included)

"Standard 88" models: Before February 1994 (before serial nr.2220 excluded)

Issue: 05 - DR 400/180 Date: 28 July 2023

Section Z: <u>DR 400/180</u>

Z.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 400/180

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: May 10, 1972

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

Z.II Certification Basis

1. Reference Date for determining

the applicable requirements: 21 March 1971

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

6. Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

8. EASA Exemptions: None9. EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 6.

Z.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition: Refer to CEAPR document n°1001131 for DR400

"STANDARD 88" (Refer to note 2)

Refer to CEAPR document n°1001130 for DR400

"STANDARD 92" (Refer to note 2)

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

Height2.23 m (7.32 ft)

Wing Area 14.20 m² (152.85 foot²)

5. Engines:

Lycoming O-360-A3A or O-360-A1A or O-360-A1P (*)

Date: 28 July 2023

(*) from serial nr 2207 included

The EASA type certification standard includes that of FAA TC E-286, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Maximum Continuous Power: 2700 rpm

6. Propellers:

Manufactur	Model	- A	Number	Minimum static RPM
er	iviodei	Ø	of blades	at sea level
Sensenich	76 EM 8S5-0-54		2	2500 rpm (3)
	76 EM 8S5-0-58	1.93 m (1)	2	2500 rpm (3)
	76 EM 8S5-0-64	1.93 111 (1)	2	2200 rpm (3)
	76 EM 8S5-0-68		2	2250 rpm (3)
Hoffmann	HO 27 HM/180/160	1.80 m (2)	2	2350 rpm

Remarks:

- (1) No acceptable diameter reduction for repair.
- (2) When Hoffmann HO 27 installed, major change nr 35 must be applied.
- (3) Do not continuous operate between 2150 rpm and 2350 rpm.

The EASA type certification standard includes that of FAA TC P4EA and FAA TC P3EU, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Number Governor Minimum static Manufacturer Model Ø of RPM at sea level blades HO V 123 Woodward B 210-Constant speed 1.80 m Hoffmann K/180R 689 (4)

Remarks:

(4) Modification of engine from O-360-A3A to O-360-A1A The EASA type certification standard includes that of FAA TC P5EU, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids: 7.1 Fuel:

Lycoming O-360-A3A or Lycoming O-360-A1A:

100/100LL octane minimum aviation grade gasoline. Lycoming O-360-A1P:

91/96 octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070.



TCDS No: EASA.A.367 DR 400/180

7.2 Engine Oil:

Issue: 05

Refer to latest revision of Service Instruction Lycoming No. 1014

Date: 28 July 2023

110. 1014.				
Air temperature	Ashless dispersant (AD)	Mineral		
All temperature	grades	grades		
All temperature	SAE15W50 or SAE20W50			
Above 80°F (+25°C)	SAE60	SAE60		
Above 60°F (+15°C)	SAE40 or SAE50	SAE50		
30°F to 90°F (O°C à +30°C)	SAE40	SAE40		
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30		
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50		
Below 10°F (-10°C)	SAE30 or SAE20W30	SAF20		

8. Fluid capacities:

8.1 Fuel:

	n tank	RH tank					Auxiliary tank	
(lit	res)	(litr	es)	(litr	es)	(optiona	I) (litres)	
Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable	
110	100/109 (1)	40	40	40	40	50	50	

(1) New standard called "Standard 92" from serial no. 2210, unusable quantity of fuel reduced from 10 litres to 1 liter, (refer to note 2).

Oil sump capacity 8 U.S. quarts (7.57 litres) 8.2 Oil:

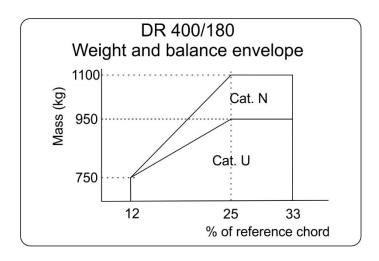
9. Air speeds:

V_{NE}	308 km/h (166 knots IAS)
V_{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
V_{A}	215 km/h (116 knots IAS)
V_{FE}	170 km/h (92 knots IAS)

- 10. Maximum Operating Altitude: Refer to approved aircraft flight manual.
- 11. Operational Capability:
- Refer to approved aircraft flight manual.
- 12. Maximum Masses:

"N" Ca	"U" Category	
Take-off	Landing	
1100 kg (2425 lb)	1045 kg (2304 lb)	950 kg (2094 lb)

13. Centre of Gravity Range:



TCDS No: EASA.A.367

Issue: 05 DR 400/180 Date: 28 July 2023

Normal Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 1100 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 1100 kg

Utility Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 950

Aft limit (33 % ref.): 0.564 m aft of datum at 950 kg

14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

Normal Category: Flaps up n + 3.8 Flaps up n - 1.9

Flaps down n..... + 2 Flaps down n.....0

Utility Category: Flaps up n + 4.4

Flaps up n - 2.2 Flaps down n..... + 2 Flaps down n.....0

16. Levelling means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.

19. Baggage / Cargo Compartment Maximum baggage compartment: 60 kg (132 lb) at 1.90m

aft of datum

.....2.58 m (8.46 ft) 20. Wheels and Tires: Main gear track

Wheel tire size Front gear angular movement left: 27°right: 27° Tire pressure refer to the maintenance manual

Oleo strut pressure refer to the maintenance manual

21. Control surface movements:

Elevator: up 9°30' ± 30'

down 12° ± 30'

Relative to the trailing edge of the Ailerons:

wings

neutral down 2° ± 1° 15°± 1° 10°± 1°

Elevator up:25°30' ± 1°......6° ± 1° Elevator tab:

Elevator down: 10°30' ± 1°16°30' ± 1°

1st notch:15° ± 5° Flaps:

2nd notch: $60^{\circ} - 5^{\circ}$

(1) For planes fitted with brakes controlled with rudder pedals:

> 16° (-0°, +2°) before operating drum brakes 20° (-0°, +3°) before operating disk brakes

22. (Reserved)



Issue: 05 - DR 400/180 Date: 28 July 2023

Z.IV Operating and Service Instructions

Airplane Flight Manual	Refer	to the	latest	amendme	nt of	Service	Letter	no.	6
Airplane Maintenance Manual	Refer	to the	latest	amendme	nt of	Service	Letter	no.	6
Airplane Maintenance Schedule	Refer	to the	latest	amendme	nt of	Service	Letter	no.	6

Z.V Note:

TCDS No: EASA.A.367

- 1. This plane is identical to DR 400/160 except:
 - Powerplant
 - towing ability (if equipped with towing hook)
- 2. "Standard 92" models: Since October 1993 (serial nr. 2207 and from serial nr. 2216 included)

"Standard 88" models: Before October 1993 (serial nr.2207 excluded and before serial nr.2216 excluded)

TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

lssue: 05 - DR 400/180 R Date: 28 July 2023

Section AA: <u>DR 400/180 R</u>

AA.I General

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 400/180 R

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: November 28, 1972

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

AA.II Certification Basis

1. Reference Date for determining

the applicable requirements: 3 August 1972

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

6. Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

Airplane and towed sailplane maximum masses are limited considering the minimum climb performances required.

8. EASA Exemptions: None9. EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 6.

AA.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition: Refer to CEAPR document n°1001131 for DR400

"STANDARD 88"

Refer to CEAPR document n°1001130 for DR400

"STANDARD 92"

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.



Issue: 05 DR 400/180 R Date: 28 July 2023

4. Dimensions: Span8.72 m (28.61 ft)(7.32 ft)Height2.23 m

(22.83 ft) - Round spinner Length......6.96 m (23.29 ft) - Sharp spinner 7.10 m

Wing Area 13.60 m² (146.39 foot²)

Lycoming O-360-A3A or Lycoming O-360-A1P (*) 5. Engines:

(*) from serial nr 2207 included

The EASA type certification standard includes that of FAA TC E-286, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior

to 28 September 2003 are also acceptable.

Maximum Continuous Power: 5.1 Engine Limits: 2600 rpm

Remark: Maximum continuous power limited by noise

regulation.

6. Propellers:

TCDS No: EASA.A.367

Manufacturar	nufacturer Model	Ø	Number	Minimum static RPM
Manufacturer		Ø	of blades	at sea level
	76 EM 8S5-0-54		2	2500
Sensenich	76 EM 8S5-0-58	1.93 m (*)	2	2400
	76 EM 8S5-0-64		2	2300
Hoffmann	HO 27 HM/180/138	1 00 m	2	2400
Evra	TR5-180-102-140	1.80 m	3	

The EASA type certification standard includes that of FAA TC P4EA and FAA TC P3EU, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: Lycoming O-360-A3A:100/100LL octane minimum

aviation

grade gasoline.

Lycoming O-360-A1P: 91/96 octane minimum aviation

grade

Refer to latest revision of Service Instruction Lycoming

No. 1070.

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014.

Air temperature	Ashless dispersant (AD) grades	Mineral grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

Issue: 05 - DR 400/180 R

- 8. Fluid capacities:
 - 8.1 Fuel:

TCDS No: EASA.A.367

	n tank res)	Auxilia (optiona	ry tank l) (litres)
Capacity	Usable	Capacity	Usable
110	100/109 (1)	50	50

Date: 28 July 2023

(1) New standard called "Standard 92" from serial no. 2210, unusable quantity of fuel reduced from 10 litres to 1liter, (refer to note 2).

9. Air speeds:

V_{NE}	308 km/h (166 knots IAS)
V_{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
V_A	215 km/h (116 knots IAS)
V_{FE}	170 km/h (92 knots IAS)

10. Maximum Operating Altitude:

Refer to approved aircraft flight manual.

11. Operational Capability:

Refer to approved aircraft flight manual.

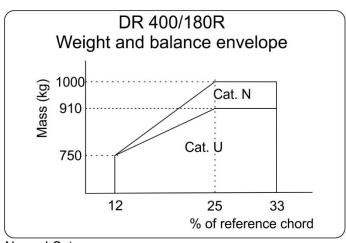
12. Maximum Masses:

	"N" Ca	"U" Category	
	Take-off	Landing	
100	00 kg (2205 lb)	1000 kg (2205 lb)	910 kg (2006 lb)

12.1 Towing mass limitations:

Each maximum mass of the tug and of the towed glider is limited by the minimum climb performance.

13. Centre of Gravity Range:



Normal Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 1000 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 1000 kg Utility Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 910 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 910 kg

DR 400/180 R Issue: 05 Date: 28 July 2023

14. Datum:		e of the rectangular part of the wings. erence section: 1.71 m (5.61 ft)
15. Load factor at maximum weight:	Normal Category:	Flaps up n + 3.8 Flaps up n - 1.9 Flaps down n + 2 Flaps down n 0
	Category Utility:	Flaps up n + 4.4 Flaps up n - 2.2 Flaps down n + 2 Flaps down n 0
16. Levelling Means:	Horizontal referen	ce upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.41±0.	05m aft of datum
18. Maximum Passenger Seating Capa	city: 1 at 0.41±0.05 datum.	5m aft of datum and 2 at 1.19m aft of
19. Baggage / Cargo Compartment	Maximum baggage aft of datum	e compartment: 60 kg (132 lb) at 1.90m
20. Wheels and Tires:		
	Wheel tire size	2.58 m (8.46 ft) 380 x 150 or 5.00-5 r movement left: 27° right: 27°
	Tire pressure Oleo strut pressure	refer to the maintenance manua e refer to the maintenance manua
21. Control surface movements:		
	Elevator:	up
	Ailerons: wings	Relative to the trailing edge of the
		up neutral down 15°± 1° 2°± 1° 10°± 1°
	Elevator tab:	Elevator up:25°30' ± 1°6° ± 1° Elevator down: 10°30' ± 1°16°30' ± 1°
	Flaps:	1st notch:15° \pm 5° \pm 40°
		2nd notch: 60° -5°
	Rudder:	25° ^{+3°} (1)
	(1) For planes fitt	ted with brakes controlled with rudder

pedals:

16° (-0°, +2°) before operating drum brakes 20° (-0°, +3°) before operating disk brakes

22. (Reserved)



Issue: 05 - DR 400/180 R Date: 28 July 2023

AA.IV Operating and Service Instructions

AA.V Note:

TCDS No: EASA.A.367

- 1. This plane is identical to DR 400/125 except:
 - powerplant
 - towing ability
 - landing gears
 - rearview mirror and rear panoramic windows
- 2. "Standard 92" models: Since October 1993 (serial nr.2207 and from serial nr. 2216 included)

"Standard 88" models: Before October 1993 (serial nr. 2207 excluded and before serial nr.2216 excluded)

3. Glider and banner towing: Refer to approved flight manual.



Issue: 05 - DR 400/2+2 Date: 28 July 2023

Section BB: <u>DR 400/2+2</u>

BB.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 400/2+2

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: December 19, 1972

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

BB.II Certification Basis

1. Reference Date for determining

the applicable requirements: 3 August 1972

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

8. EASA Exemptions: None9. EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 6.

BB.III Technical Characteristics and Operational Limitations

1. Type Design Definition: Refer to CEAPR document n°1001131

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification.

Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed..

4. Dimensions:

5. Engines: Lycoming O-235-H2C or O-235-C2C

The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28



TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

Issue: 05 DR 400/2+2 Date: 28 July 2023

> September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2600 rpm

Remark: Maximum continuous power limited by noise

regulation.

6. Propellers:

Manufacturer Model		Ø	Number	Minimum static RPM
Manufacturer	iviodei		of blades	at sea level
	1 A 105 BCM 70-56	1.78 m	2	
Mac Cauley	1 A 105 BCM 70-60	1.70 m	2	2250
	1 A 90 ECM 72-50	1.83 m	2	2230
Evra	88-75-34 F	1.76 m	2	

The EASA type certification standard includes that of FAA TC P-918 and FAA TC P-842, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 80/87 octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014

10. 1014.				
Air temperature	Ashless dispersant (AD)	Mineral		
All temperature	grades	grades		
All temperature	SAE15W50 or SAE20W50			
Above 80°F (+25°C)	SAE60	SAE60		
Above 60°F (+15°C)	SAE40 or SAE50	SAE50		
30°F to 90°F (O°C à +30°C)	SAE40	SAE40		
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30		
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50		
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20		

8. Fluid capacities:

8.1 Fuel:

Main tank		Auxiliary tank		
(litres)		(optional) (litres)		
Capacity	Usable	Capacity	Usable	
110	100	50	50	

8.2 Oil: Oil sump capacity 6 U.S. quarts (5.68 litres)

TCDS No: EASA.A.367 Issue: 05 DR 400/2+2

9. Air speeds:

V_{NO}260 km/h (140 knots IAS) V_C......260 km/h (140 knots IAS) V_A215 km/h (116 knots IAS) V_{FE}170 km/h (92 knots IAS)

Date: 28 July 2023

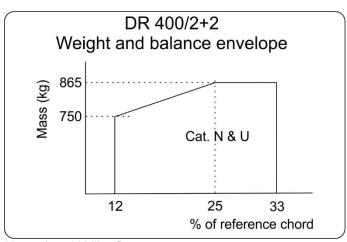
Refer to approved aircraft flight manual. 10. Maximum Operating Altitude:

11. Operational Capability: Refer to approved aircraft flight manual.

12. Maximum Masses:

"N" Cate	"U" Category	
Take-off	Landing	
865 kg (1907 lb)	865 kg (1907 lb)	865 kg (1907 lb)

13. Centre of Gravity Range:



Normal and Utility Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 865

Aft limit (33 % ref.): 0.564 m aft of datum at 865 kg

14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

Normal Category: Flaps up n+ 3.8

Flaps up n - 1.9 Flaps down n..... + 2 Flaps down n.....0

Utility Category: Flaps up n + 4.4

Flaps up n - 2.2 Flaps down n..... + 2 Flaps down n.....0

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.

19. Baggage / Cargo Compartment None

Issue: 05 - DR 400/2+2 Date: 28 July 2023

21. Control surface movements:

TCDS No: EASA.A.367

Ailerons: Relative to the trailing edge of the

wings

 up
 neutral
 down

 15°± 1°
 2° ± 1°
 10°± 1°

Elevator tab: Elevator up:25°30' \pm 1°......6° \pm 1°

Elevator down: 10°30′ ± 1°16°30′ ± 1° 1st notch: 15° + 5°

Flaps: 1st notch:15° ± 5°

2nd notch: $\begin{array}{c} +0^{\circ} \\ 60^{\circ} -5^{\circ} \end{array}$ Rudder: $\begin{array}{c} 25^{\circ} -0^{\circ} \\ \end{array}$ (1)

(1) For planes fitted with brakes controlled with rudder

(1) For planes fitted with brakes controlled with rudder pedals:

16° (-0°, +2°) before operating drum brakes 20° (-0°, +3°) before operating disk brakes

22. (Reserved)

BB.IV Operating and Service Instructions

Airplane Flight Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	. Refer to the latest amendment of Service Letter no. 6

BB.V Note:

- 1. This plane is identical to DR 400/125 except:
 - powerplant
 - luggage compartment removed
 - rear seat



lssue: 05 - DR 400/120 Date: 28 July 2023

Section CC: <u>DR 400/120</u>

CC.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 400/120

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: February 11, 1975

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

CC.II Certification Basis

1. Reference Date for determining

the applicable requirements: 18 September 1974

(Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

8. EASA Exemptions: None9. EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 6.

CC.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition: Refer to CEAPR document n°1001131 for DR400

"STANDARD 88" (Refer to note 2)

Refer to CEAPR document n°1001130 for DR400

"STANDARD 92" (Refer to note 2)

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed..

4. Dimensions: Span8.72 m (28.61 ft)

Height2.23 m (7.32 ft)

Wing Area 13.60 m² (146.39 foot²)

5. Engines: Lycoming O-235-L2A or O-235-K2A or O-235-K2B



Issue: 05 - DR 400/120

The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Date: 28 July 2023

5.1 Engine Limits: Maximum continuous Power: 2800 rpm

6. Propellers:

TCDS No: EASA.A.367

Manufacturer	Model	Ø	Number	Minimum static RPM
Manuaciurei	iviodei	Ø	of blades	at sea level
Mac Cauley	1 A 135 JCM 71-47	1.80 m	2	2220
Hoffmann	HO-14-178/115	1.78 m	2	2250
	72CK-S6-0-56	1.83 m (*)	2	2220
Sensenich	72CK-S5-0-56	1.03 111 ()	2	2220
	72CK-S6-0-54	1.83 m	2	2300

Remark: (*) No acceptable diameter reduction for repair.

The EASA type certification standard includes that of FAA TC P-842 and FAA TC P-904, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 100/100LL octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070.

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming No. 1014.

Ashless dispersant (AD) Mineral Air temperature grades grades All temperature SAE15W50 or SAE20W50 Above 80°F (+25°C) SAE60 SAE60 Above 60°F (+15°C) SAE40 or SAE50 SAE50 30°F to 90°F (O°C à +30°C) SAE40 SAE40 0°F to 70°F (-15°C à +20°C) SAE30, SAE40 or SAE20W40 SAE30 0°F to 90°F (-15°C à +30°C) SAE20W50 or SAE15W50 SAE20W50 Below 10°F (-10°C) SAE30 or SAE20W30 SAE20

8. Fluid capacities:

8.1 Fuel:

Mai	n tank	Auxiliary tank	
(lit	res)	(optiona	I) (litres)
Capacity Usable		Capacity	Usable
110	100/109 (1)	50	50

(1) New standard called "Standard 92" from serial number 2210, unusable quantity of fuel reduced from 10 litres to 1liter. (Refer to note 2)

9. Air speeds:



TCDS No: EASA.A.367 DR 2

Date: 28 July 2023

10. Maximum Operating Altitude: Refer to approved aircraft flight manual.

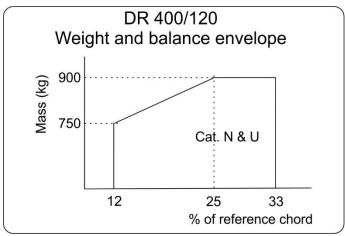
11. Operational Capability: Refer to approved aircraft flight manual.

12. Maximum Masses:

Issue: 05

"N" Cat	"U" Category	
Take-off	Landing	
900 kg (1984 lb)	900 kg (1984 lb)	900 kg (1984 lb)

13. Centre of Gravity Range:



Normal and Utility Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 900 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 900 kg

- 14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)
- 15. Load factor at maximum weight:

Normal Category:	Flaps up n Flaps up n Flaps down n Flaps down n	1.9 + 2
Utility Category:	Flaps up nFlaps up n	2.2

- 16. Levelling Means: Horizontal reference upper fuselage spar.
- 17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum
- 18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.
- 19. Baggage / Cargo Compartment Maximum baggage compartment: 40 kg (88 lb) at 1.90m aft of datum
- 20. Wheels and Tires:

Main gear track		2.58 m (8.46 ft)
Wheel tire size		380 x 150 or 5.00-5
Front gear angular	movement	left: 27°
		right: 27°



lssue: 05 - DR 400/120 Date: 28 July 2023

Tire pressure refer to the maintenance manual Oleo strut pressure refer to the maintenance manual

21. Control surface movements:

TCDS No: EASA.A.367

Ailerons: Refer to following table

 up
 neutral
 down

 15°± 1°
 2° ± 1°
 10°± 1°

Elevator tab: Elevator up:25°30' \pm 1°......6° \pm 1°

Elevator down: 10°30' ± 1°16°30' ± 1°

Flaps: 1st notch:15° \pm 5°

2nd notch: 60° -5°

Rudder:25° -0° (1)

(1) For planes fitted with brakes controlled with rudder pedals:

16° (-0°, +2°) before operating drum brakes 20° (-0°, +3°) before operating disk brakes

22. (Reserved)

CC.IV Operating and Service Instructions

CC.V Note:

- 1. This plane is identical to DR 400/125 except powerplant
- 2. "Standard 92" models: Since July 1993 (from serial nr. 2212 included)

"Standard 88" models: Before July 1993 (before serial nr.2212 excluded)

Issue: 05 - DR 400/125i Date: 28 July 2023

Section DD: <u>DR 400/125i</u>

DD.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 400/125i

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: September 25, 1975

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

DD.II Certification Basis

1. Reference Date for determining

the applicable requirements: 18 September 1974

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

6. Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

8. EASA Exemptions: None9. EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 6.

DD.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition: Refer to CEAPR document n°1001131

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

4. Dimensions:

5. Engines: Continental IO-240 A, B

The EASA type certification standard includes that of FAA TC E-7SO, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS



TCDS No: EASA.A.367

Issue: 05 DR 400/125i Date: 28 July 2023

> standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum continuous power: 2800 rpm

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Governor	Minimum static RPM at sea level
MT Propeller	MTV-7-D/170-09	1.70 m	3	Electrical variable pitch	Constant speed

The EASA type certification standard includes that of FAA TC P20BO, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 91/96 or 100/130 octane minimum aviation grade gasoline

7.2 Engine Oil: Teledyne Continental engine IO-240-B

(Refer to Continental specifications MHS24 or MHS-25

and SB M87-12R1)

Oil	Ashless dispersant (AD)	Straight mineral
All temperatures	SAE15W50 or 20W50	
Above +4°C (40°F)	SAE15W50 or 20W60	SAE50
Below +4°C (40°F)	10W30, 15W30, 20W50	SAE30

7.3 Coolant: Not Applicable

8. Fluid capacities:

8.1 Fuel:

Maiı	n tank	Auxiliary tank	
(litres)		(optional) (litres)	
Capacity	Usable	Capacity	Usable
110	109	50	50

8.2 Oil:

maximum	minimum
5.7 litres	2.9 litres

9. Air speeds:

V 000 long/lb /440 longet	(SAI s
V _{NO} 260 km/h (140 knot	,
Vc260 km/h (140 knot	s IAS)
V _A 215 km/h (116 knot	s IAS)
V _{FE} 170 km/h (92 knot	s IAS)

10. Maximum Operating Altitude:

Refer to approved aircraft flight manual.

11. Operational Capability:

Refer to approved aircraft flight manual.



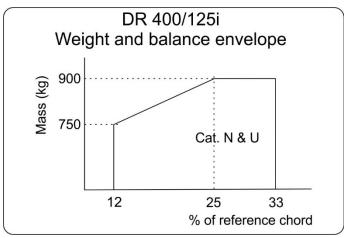
TCDS No: EASA.A.367 Issue: 05 DR 400/125i

12. Maximum Masses:

"N" Cat	"U" Category	
Take-off Landing		
900 kg (1984 lb)	900 kg (1984 lb)	900 kg (1984 lb)

Date: 28 July 2023

13. Centre of Gravity Range:



Normal and Utility Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 900

Aft limit (33 % ref.): 0.564 m aft of datum at 900 kg

14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

Normal Category:	Flaps up n	1.9
	Flaps down n	
	Flaps down n	
Utility Category:	Flaps up n	+ 4.4
	Flaps up n	2.2
	Flaps down n	+ 2
	Flaps down n	0

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.

19. Baggage / Cargo Compartment

Maximum baggage compartment: 40 kg (88 lb) at 1.90m aft of datum

20. Wheels and Tires

Main gear track		2.58 m (8.46 ft)
		380 x 150 or 5.00-5
Front gear angular	movement	left: 27°
		right: 27°
Tire pressure	refer to the	maintenance manual
Oleo strut pressure	refer to the	maintenance manual

lssue: 05 - DR 400/125i Date: 28 July 2023

21. Control surface movements

TCDS No: EASA.A.367

Ailerons: Relative to the trailing edge of the

wings

 up
 neutral
 down

 15°± 1°
 2° ± 1°
 10°± 1°

Elevator tab: Elevator up:.....25°30' \pm 1°......6° \pm 1°

Elevator down: 10°30′ ± 1°16°30′ ± 1° 15° + 5°

Flaps: 1st notch:15° ± 5°

2nd notch: $60^{\circ} - 5^{\circ}$

Rudder:25° ^{+3°}_{-0°} (1)

(1) For planes fitted with brakes controlled with rudder pedals:

16° (-0°, +2°) before operating drum brakes

20° (-0°, +3°) before operating disk brakes

22. (Reserved)

DD.IV Operating and Service Instructions

Airplane Flight Manual	Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	Refer to the latest amendment of Service Letter no. 6

DD.V Note:

1. This plane is identical to DR 400/120 except powerplant



lssue: 05 - DR 400/140 B Date: 28 July 2023

Section EE: <u>DR 400/140 B</u>

EE.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 400/140 B

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: November 09, 1975

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

EE.II Certification Basis

1. Reference Date for determining

the applicable requirements: 29 August 1975

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

6. Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

8. EASA Exemptions: None9. EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 6.

EE.III<u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition: Refer to CEAPR document n°1001131 for DR400

"STANDARD 88" (Refer to note 2)

Refer to CEAPR document n°1001130 for DR400

"STANDARD 92" (Refer to note 2)

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

Height2.23 m (7.32 ft)

Wing Area 13.60 m² (146.39 foot²)



Issue: 05 - DR 400/140 B Date: 28 July 2023

Engines:

TCDS No: EASA.A.367

Lycoming O-320-D2A

The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

Propeller Manufacturer	Model	Maximum Continuous Power RPM
Sensenich	74 DM 6S5-2-64	2700 rpm (1)
Sensenion	74 DM 6S5-2-60	2500 rpm (1)

Remarks: (1) Maximum continuous power limited by noise regulation.

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Sensenich	74 DM 6S5-2-64	1.83 m (1)	2	2200 rpm
Sensemen	74 DM 6S5-2-60	1.03 111 (1)	2	2300 rpm

Remarks: (1) No acceptable diameter reduction for repair.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 91/96 or 100/130 octane minimum aviation grade

gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
7 ii temperatare	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

- DR 400/140 B

8. Fluid capacities:

8.1 Fuel:

TCDS No: EASA.A.367

Issue: 05

Main tank (litres)		Auxiliary tank (optional) (litres)	
Capacity	Úsable	Capacity Usabl	
110	100/109 (1)	50	50

Date: 28 July 2023

(1) New standard called "Standard 92" from serial number 2210, unusable quantity of fuel reduced from 10 litres to 1 liter, (refer to note 2).

9. Air speeds:

V_{NE}	308 km/h (166 knots IAS)
V_{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
V_A	215 km/h (116 knots IAS)
V_{FE}	170 km/h (92 knots IAS)

10. Maximum Operating Altitude:

Refer to approved aircraft flight manual.

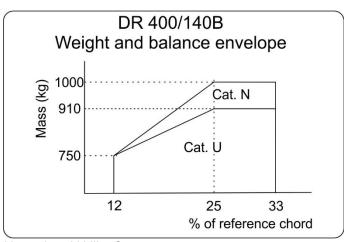
11. Operational Capability:

Refer to approved aircraft flight manual.

12. Maximum Masses:

"N" Category		"U" Category
Take-off Landing		
1000 kg (2205 lb)	1000 kg (2205 lb)	910 kg (2006 lb)

13. Centre of Gravity Range:



Normal and Utility Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 1000 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 1000 kg Normal and Utility Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 910 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 910 kg

14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:



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TCDS No: EASA.A.367

DR 400/140 B Date: 28 July 2023 Issue: 05

c. 95	211 100, 210 2	24.6. 2014., 2020
	Normal Category	7: Flaps up n
	Utility Category:	Flaps up n + 4.4 Flaps up n - 2.2 Flaps down n + 2 Flaps down n 0
16. Levelling Means:	Horizontal refere	nce upper fuselage spar
17. Minimum Flight Crew:	1 (pilot) at 0.41±	0.05m aft of datum
18. Maximum Passenger Seating 0	Capacity: 1 at 0.41±0. datum.	05m aft of datum and 2 at 1.19m aft of
19. Baggage / Cargo Compartmen	t Maximum bagga aft of datum	ge compartment: 40 kg (88 lb) at 1.90m
20. Wheels and Tires:	Wheel tire size Front gear angul Tire pressure	ar movementleft: 27° right: 27°
21. Control surface movements:	Oldo du di proces	
	Elevator:	up
	Ailerons: wings	down
		up neutral down
	Elevator tab:	15°± 1° 2° ± 1° 10°± 1° Elevator up:25°30' ± 1°6° ± 1° Elevator down: 10°30' ± 1°16°30' ± 1°
	Flaps:	1st notch:15° ± 5°
		2nd notch: $60^{\circ} - 5^{\circ}$
		25° ^{+3°} _{-0°} (1)
	pedals: 16° (-0°, +2°	itted with brakes controlled with rudder) before operating drum brakes) before operating disk brakes
22. (Reserved)		
V Operating and Service Instru	ctions	
Airplane Flight Manual Airplane Maintenance Manual.	Refer to the	latest amendment of Service Letter no. 6 latest amendment of Service Letter no. 6

EE.I

Airplane Flight Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	. Refer to the latest amendment of Service Letter no. 6



Issue: 05 - DR 400/140 B Date: 28 July 2023

EE.V Note:

TCDS No: EASA.A.367

1. This plane is identical to DR 400/140 except powerplant

2. "Standard 92" models: Since June 1993 (from serial nr 2211 included)

"Standard 88" models: Before June 1993 (before serial nr.2211 excluded)



Issue: 05 - DR 400/120A Date: 28 July 2023

Section FF: DR 400/120A

FF.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 400/120 A

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: November 15, 1976

7. EASA Type Certification date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

FF.II Certification Basis

1. Reference Date for determining

the applicable requirements: 28 June 1976

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

6. Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

8. EASA Exemptions: None9. EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 6.

FF.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition: Refer to CEAPR document n°1001131

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed...

4. Dimensions:

 Span
 8.72 m
 (28.61 ft)

 Height
 2.23 m
 (7.32 ft)

 Length
 6.96 m
 (22.83 ft)

 Wing Area
 13.60 m²
 (146.39 foot²)

5. Engines: Lycoming O-235-L2A or O-235-K2A or O-235-K2B

The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS



TCDS No: EASA.A.367 Issue: 05 DR 400/120A

> standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Date: 28 July 2023

5.1 Engine Limits: Maximum continuous power: 2800 rpm

6. Propellers:

<u>. </u>				
Manufactur	Model	Ø	Number	Minimum static RPM
er	iviodei	Ø	of blades	at sea level
Mac Cauley	1 A 135 JCM 71-50	1.80 m	2	2200 rpm (2)
		1.77 m (1)	2	
	1 A 135 JCM 71-47	1.80 m	2	
		1.77 m (1)	2	
Hoffmann	HO-14-178/115	1.78 m	2	2250 rnm
		1.73 m (1)	2	2250 rpm

Remarks:

- (1) Minimum diameter after repair.
- (2) Do not continuous operate between 2025 rpm and 2325 rpm.

The EASA type certification standard includes that of FAA TC P-842, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 100/100LL octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014.

110.1011.				
Air temperature	Ashless dispersant (AD)	Mineral		
All temperature	grades	grades		
All temperature	SAE15W50 or SAE20W50			
Above 80°F (+25°C)	SAE60	SAE60		
Above 60°F (+15°C)	SAE40 or SAE50	SAE50		
30°F to 90°F (O°C à +30°C)	SAE40	SAE40		
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30		
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50		
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20		

8. Fluid capacities:

8.1 Fuel:

Mai	n tank	Auxiliary tank		
(litres)		(optional) (litres)		
Capacity	Usable	Capacity	Usable	
110	100	50	50	

8.2 Oil: Oil sump capacity 6 U.S. quarts (5.68 litres)



TCDS No: EASA.A.367 Issue: 05 DR 400/120A

9. Air speeds:

V_{NE}	
V_{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
V_{A}	215 km/h (116 knots IAS)
V_{FE}	

Date: 28 July 2023

10. Maximum Operating Altitude:

Refer to approved aircraft flight manual.

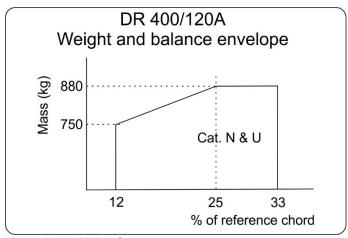
11. Operational Capability:

Refer to approved aircraft flight manual.

12. Maximum Masses:

"N" Ca	"U" Category	
Take-off Landing		
880 kg (1940 lb)	880 kg (1940 lb)	880 kg (1940 lb)

13. Centre of Gravity Range:



Normal and Utility Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 880

Aft limit (33 % ref.): 0.564 m aft of datum at 880 kg

14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

Normal Category:	Flaps up n	+ 3.8
	Flaps up n	1.9
	Flaps down n	
	Flaps down n	
Utility Category:	Flaps up n	+ 4.4
- , - <u> </u>	Flaps up n	
	Flaps down n	
	Flaps down n	0

- 16. Levelling Means: Horizontal reference upper fuselage spar
- 17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum
- 18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of
- 19. Baggage / Cargo Compartment Maximum baggage compartment: 40 kg (88 lb) at 1.90m aft of datum

Issue: 05 - DR 400/120A Date: 28 July 2023

20. Wheels and Tires:

TCDS No: EASA.A.367

21. Control surface movements

Relative to the trailing edge of the

Ailerons: wings:

 up
 neutral
 down

 15°± 1°
 2° ± 1°
 10°± 1°

Elevator tab: Elevator up:25°30' \pm 1°......6° \pm 1°

Elevator down: $10^{\circ}30' \pm 1^{\circ}16^{\circ}30' \pm 1^{\circ}$

Flaps: 1st notch:15° ± 5°

2nd notch: $60^{\circ} - 5^{\circ}$

Rudder:25 $^{\circ}_{-0^{\circ}}^{+3^{\circ}}(1)$

(1) For planes fitted with brakes controlled with rudder pedals:

16° (-0°, +2°) before operating drum brakes 20° (-0°, +3°) before operating disk brakes

22. (Reserved)

FF.IV Operating and Service Instructions

Airplane Flight Manual	Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	Refer to the latest amendment of Service Letter no. 6

FF.V Note:

- 1. This plane is identical to DR 400/120 except:
 - propeller
 - maximum mass



lssue: 05 - DR 400/160D Date: 28 July 2023

Section GG: DR 400/160D

GG.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 400/160 D

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: April 27, 1981

7. EASA Type Certification date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

GG.II Certification Basis

1. Reference Date for determining

the applicable requirements: 21 March 1971

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

EASA Exemptions: None
 EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 6.

GG.III Technical Characteristics and Operational Limitations

1. Type Design Definition: Refer to CEAPR document n°1001131

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed...

4. Dimensions: Span8.72 m (28.61 ft)

Height2.23 m (7.32 ft)

7.10 m (23.29 ft) - Sharp spinner

Wing Area 14.2 m² (152.85 foot²)

5. Engines: Lycoming O-320-D2A

The EASA type certification standard includes that of FAA TC E-274, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS



TCDS No: EASA.A.367 Issue: 05 DR 400/160D

standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Date: 28 July 2023

5.1 Engine Limits: Maximum continuous Power: 2600 rpm

Remark: Maximum continuous power limited by noise

regulation.

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Sensenich	M74 DMS-2-66	1 02 m (1)	2	2150 rpm
Sensenion	74 DM6S5-2-64	1.83 m (1)	2	2250 rpm

Remark: (1) No acceptable diameter reduction for repair.

The EASA type certification standard includes that of FAA TC P-886, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 91/96 or 100/130 octane minimum aviation grade

gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070.

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014.

140: 1011:				
Air temperature	Ashless dispersant (AD)	Mineral		
All temperature	grades	grades		
All temperature	SAE15W50 or SAE20W50			
Above 80°F (+25°C)	SAE60	SAE60		
Above 60°F (+15°C)	SAE40 or SAE50	SAE50		
30°F to 90°F (O°C à +30°C)	SAE40	SAE40		
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30		
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50		
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20		
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W		

Fluid capacities:

8.1 Fuel:

Main tank		RH	tank	LH tank		Auxiliary tank	
(lit	res)	(litr	es)	(litr	es)	(optiona	l) (litres)
Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable
110	100/109 (1)	40	40	40	40	50	50

(1) New standard called "Standard 92" from serial number 2210, unusable quantity of fuel reduced from 10 litres to 1 liter (refer to note 2).

Oil sump capacity 8 U.S. quarts (7.57 litres) 8.2 Oil:

9. Air speeds:

V_{NE}308 km/h (166 knots IAS) V_{NO}260 km/h (140 knots IAS) V_C......260 km/h (140 knots IAS)



10. Maximum Operating Altitude:

Refer to approved aircraft flight manual.

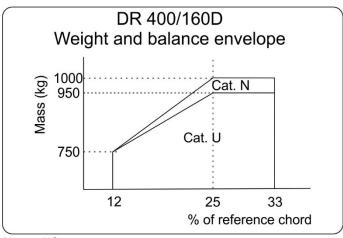
11. Operational Capability:

Refer to approved aircraft flight manual.

12. Maximum Masses:

"N" Ca	"U" Category		
Take-off	Landing		
1000 kg (2205 lb)	1000 kg (2205 lb)	950 kg (2094 lb)	

13. Centre of Gravity Range:



Normal Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 1000 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 1000 kg Utility Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 950 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 950 kg

14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

 Normal Category:
 Flaps up n
 + 3.8

 Flaps up n
 - 1.9

 Flaps down n
 + 2

 Flaps down n
 0

 Utility Category:
 Flaps up n
 + 4.4

 Flaps up n
 - 2.2

 Flaps down n
 + 2

 Flaps down n
 0

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.

19. Baggage / Cargo Compartment Maximum baggage compartment: 40 kg (88 lb) at 1.90m aft of datum

20. Wheels and Tires:



Issue: 05 - DR 400/160D Date: 28 July 2023

21. Control surface movements:

TCDS No: EASA.A.367

down 12° ± 30'

Ailerons: Refer to following table

 up
 neutral
 down

 15°± 1°
 2° ± 1°
 10°± 1°

Elevator tab: Elevator up:25°30' \pm 1°......6° \pm 1°

Elevator down: $10^{\circ}30' \pm 1^{\circ}16^{\circ}30' \pm 1^{\circ}$

Flaps: 1st notch:15° \pm 5°

2nd notch: $60^{\circ} - 5^{\circ}$

(1) For planes fitted with brakes controlled with rudder pedals:

16° (-0°, +2°) before operating drum brakes 20° (-0°, +3°) before operating disk brakes

22. (Reserved)

GG.IV Operating and Service Instructions

Airplane Flight Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	. Refer to the latest amendment of Service Letter no. 6



Issue: 05 - DR 400/160D Date: 28 July 2023

GG.VNote:

TCDS No: EASA.A.367

- 1. This plane is identical to DR 400/160 except:
 - maximum continuous power rpm
 - maximum mass
- 2. "Standard 92" model: since November 1993



Issue: 05 - DR 400/120 D Date: 28 July 2023

Section HH: <u>DR 400/120 D</u>

HH.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 400/120 D

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

DGAC Type Certification date: April 28, 1981

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

HH.II Certification Basis

1. Reference Date for determining

the applicable requirements: 28 June 1976

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

6. Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

8. EASA Exemptions: None9. EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 6.

HH.III Technical Characteristics and Operational Limitations

1. Type Design Definition: Refer to CEAPR document n°1001131

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed..

4. Dimensions:

5. Engines: Lycoming O-235-L2A or O-235-K2A or O-235-K2B

The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28



Issue: 05 - DR 400/120 D

September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Date: 28 July 2023

5.1 Engine Limits:

TCDS No: EASA.A.367

Maximum Continuous Power:

Propeller Manufacturer	Propeller model	Maximum Continuous Power RPM
Sensenich	72 CKS6-0-56	2600 (1)
Hoffmann	HO-14-178/115	2583 (1)

Remarks: (1) Maximum continuous power limited by noise regulation.

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Sensenich	72 CK-S6-0-56	1.83 m (1)	2	2220
Hoffmann	HO-14-178/115	1.78 m 2	2220	
пошпапп	HO-14-176/113	1.73 m (2)	2	2250

Remarks:

- (1) No acceptable diameter reduction for repair.
- (2) Minimum diameter after repair.

The EASA type certification standard includes that of FAA TC P-904, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

100/100LL octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070.

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
All temperature	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

Issue: 05 - DR 400/120 D

8. Fluid capacities:

8.1 Fuel:

TCDS No: EASA.A.367

Main tank		Auxiliary tank	
(litres)		(optional) (litres)	
Capacity	Usable	Capacity	Usable
110	100	50	50

Date: 28 July 2023

9. Air speeds:

V_{NE}	308 km/h (166 knots IAS)
V_{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
V_A	215 km/h (116 knots IAS)
	170 km/h `(92 knots IAS)

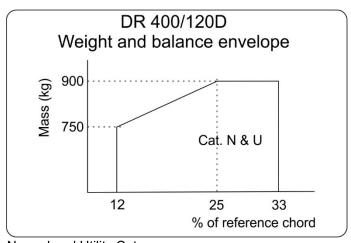
10. Maximum Operating Altitude: Refer to approved aircraft flight manual.

11. Operational Capability: Refer to approved aircraft flight manual.

12. Maximum Masses:

Ī	"N" Category		"U" Category
	Take-off	Landing	
	900 kg (1984 lb)	900 kg (1984 lb)	900 kg (1984 lb)

13. Centre of Gravity Range:



Normal and Utility Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 900 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 900 kg

14. Datum: Wing leading edge of the rectangular part of the wings.

Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

Normal Category:	Flaps up n	1.9 + 2
Utility Category:	Flaps up nFlaps up nFlaps down nFlaps down n	2.2 + 2



DR 200, DR300, and DR400 series TCDS No: EASA.A.367

Issue: 05 DR 400/120 D Date: 28 July 2023

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of

datum.

19. Baggage / Cargo Compartment Maximum baggage compartment: 40 kg (88 lb) at 1.90m

aft of datum

20. Wheels and Tires

Main gear track	2.58 m (8.46 ft)
Wheel tire size	380 x 150 or 5.00-5
Front gear angular	movementleft: 27°
	right: 27°
Tire pressure	refer to the maintenance manual
Oleo strut pressure	refer to the maintenance manual

21. Control surface movements

Elevator:	.up				9°3	0' ±	30'
	down						
Ailerons:	.Relative	to	the	trailing	edge	of	the
wings							

wings			
	up	neutral	down
	15°± 1°	2° ± 1°	10°± 1°
Elevator tab:	Elevator up:	25°30' ± 1	1°6° ± 1°
	Elevator dow	n: 10°30' ± 1	1° 16°30' ± 1°
Flaps:	1st notch:		5° ± 5°
			+0°
	2nd notch:		0° -5°
Rudder:			25° ^{+3°} _{-0°} (1)

1) For planes fitted with brakes controlled with rudder

16° (-0°, +2°) before operating drum brakes

20° (-0°, +3°) before operating disk brakes

22. (Reserved)

HH.IV Operating and Service Instructions

Airplane Flight Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	Refer to the latest amendment of Service Letter no. 6

HH.V Note:

1. This plane is identical to DR 400/120 except maximum continuous power rpm



Issue: 05 - DR 400/180 S Date: 28 July 2023

Section II: <u>DR 400/180 S</u>

II.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 400/180S

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: February 11, 1985

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

II.II Certification Basis

1. Reference Date for determining

the applicable requirements: 31 January 1985

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

8. EASA Exemptions: None9. EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 6.

II.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition: Refer to CEAPR document n°1001131

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

4. Dimensions: Span8.72 m (28.61 ft)

Height2.23 m (7.32 ft)

Wing Area 14.2 m² (152.85 foot²)

5. Engines: Lycoming O-360-A3A

The EASA type certification standard includes that of FAA TC E-286, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS



TCDS No: EASA.A.367 Issue: 05 DR 400/180 S

> standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Date: 28 July 2023

5.1 Engine Limits: Maximum continuous power: 2600 rpm

Remark: Maximum continuous power limited by noise

regulation.

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Sensenich	76 EM8S5-0-64	1.93 m (1)	2	2250 (2)

Remarks:

- (1) No acceptable diameter reduction for repair.
- (2) Do not continuous operate between 2150 rpm and 2350 rpm.

The EASA type certification standard includes that of FAA TC P4EA, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 100/100LL octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070.

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral
All temperature	grades	grades
All temperature	SAE15W50 or SAE20W50	
Above 80°F (+25°C)	SAE60	SAE60
Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:

Main tank		RH tank		LH tank		Auxiliary tank	
(litres) (litres)		es)	(litres)		(optional) (litres)		
Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable
110	100/109 (1)	40	40	40	40	50	50

(1) New standard called "Standard 92" from serial number 2210, unusable quantity of fuel reduced from 10 litres to 1 liter, (refer to note 2).

Oil sump capacity 8 U.S. quarts (7.57 litres) 8.2 Oil: Usable...... 6 U.S. quarts (5.68 litres)

9. Air speeds:

V_{NE}	
	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
V_{A}	215 km/h (116 knots IAS)
$V_{\text{FE}} \\$	



TCDS No: EASA.A.367 Issue: 05 DR 400/180 S

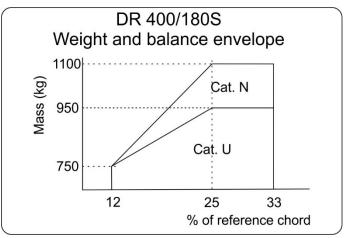
10. Maximum Operating Altitude: Refer to approved aircraft flight manual. 11. Operational Capability: Refer to approved aircraft flight manual.

12. Maximum Masses:

"N" Ca	"U" Category	
Take-off	Landing	
1100 kg (2425 lb)	1045 kg (2304 lb)	950 kg (2094 lb)

Date: 28 July 2023

13. Centre of Gravity Range:



Normal Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 1100 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 1100 kg **Utility Category**

Forward limit (12 % ref.): .. 0.205 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 950

Aft limit (33 % ref.): 0.564 m aft of datum at 950 kg

14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

Normal Category: Flaps up n+ 3.8 Flaps up n - 1.9

Flaps down n..... + 2 Flaps down n.....0

Utility Category: Flaps up n + 4.4

Flaps up n - 2.2 Flaps down n..... + 2 Flaps down n.....0

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of

19. Baggage / Cargo Compartment Maximum baggage compartment: 60 kg (132 lb) at 1.90m aft of datum)

DR 200, DR300, and DR400 series

lssue: 05 - DR 400/180 S Date: 28 July 2023

20. Wheels and Tires:

TCDS No: EASA.A.367

21. Control surface movements

Ailerons: wings

 up
 neutral
 down

 15°± 1°
 2° ± 1°
 10°± 1°

Elevator tab: Elevator up:25°30' \pm 1°......6° \pm 1°

Elevator down: 10°30′ ± 1°16°30′ ± 1° 1st notch:15° ± 5°

(1) For planes fitted with brakes controlled with rudder pedals:

16° (-0°, +2°) before operating drum brakes 20° (-0°, +3°) before operating disk brakes

22. (Reserved)

II.IV Operating and Service Instructions

Airplane Flight Manual	Refer to	the latest	amendment	of Service	Letter no	o. 6
Airplane Maintenance Manual	Refer to	the latest	amendment	of Service	Letter no	o. 6
Airplane Maintenance Schedule	Refer to	the latest	amendment	of Service	Letter no	o. 6

II.V Note:

- 1. This plane is identical to DR 400/180 except:
 - maximum continuous power rpm
 - Sensenich 76 EM8S5-0-64 propeller only
- 2. "Standard 92" model



Issue: 05 - DR 400/100 Date: 28 July 2023

Section JJ: DR 400/100

JJ.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 400/100

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: November 06, 1987

7. EASA Type Certification date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

JJ.II <u>Certification Basis</u>

1. Reference Date for determining

the applicable requirements: 13 April 1987

(Reserved)
 (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

6. Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

8. EASA Exemptions: None9. EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 6.

JJ.III Technical Characteristics and Operational Limitations

1. Type Design Definition: Refer to CEAPR document n°1001131 for DR400

"STANDARD 88" (Refer to note 2)

Refer to CEAPR document n°1001130 for DR400

"STANDARD 92" (Refer to note 2)

2. Description: Single-engine, two-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

4. Dimensions: Span8.72 m (28.61 ft)

Height2.3 m (7.32 ft)

Wing Area 13.60 m² (146.39 foot²)

5. Engines: Lycoming O-235-L2A



TCDS No: EASA.A.367 Issue: 05 DR 400/100

> The EASA type certification standard includes that of FAA TC E-223, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits:

2600 rpm

Maximum continuous power:

Remark: Maximum continuous power limited by noise

Date: 28 July 2023

regulation.

6. Propellers:

Manufacturer	Model	Ø	Number	Minimum static RPM
iviariulacturei	Model	Ø	of blades	at sea level
Sensenich	72 CKS6-0-56	1.83 m (*)	2	2220 rpm

Remarks: (*) No acceptable diameter reduction for repair.

The EASA type certification standard includes that of FAA TC P-904, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel:

100/100LL octane minimum aviation grade gasoline. Refer to latest revision of Service Instruction Lycoming No. 1070.

7.2 Engine Oil:

Refer to latest revision of Service Instruction Lycoming No. 1014.

Ashless dispersant (AD) Mineral Air temperature grades grades SAE15W50 or SAE20W50 All temperature Above 80°F (+25°C) SAE60 SAE60

Above 60°F (+15°C)	SAE40 or SAE50	SAE50
30°F to 90°F (O°C à +30°C)	SAE40	SAE40
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20

8. Fluid capacities:

8.1 Fuel:

Main tank		Auxiliary tank	
(litres)		(optional) (litres)	
Capacity	Usable	Capacity Usabl	
110	100/109 (1)	50	50

(1) New standard called "Standard 92" from serial number 2210, unusable quantity of fuel reduced from 10 litres to 1 liter, (refer to note 2).

Oil sump capacity 6 U.S. quarts (5.68 litres) 8.2 Oil:

9. Air speeds:

V_{NE}	308 km/h (166 knots IAS)
V_{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
V_{A}	215 km/h (116 knots IAS)



Issue: 05 DR 400/100 Date: 28 July 2023

V_{FE}170 km/h (92 knots IAS)

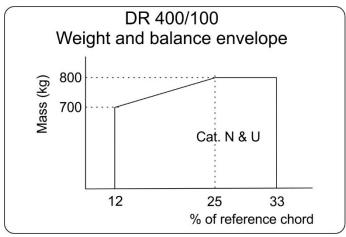
Refer to approved aircraft flight manual. 10. Maximum Operating Altitude:

11. Operational Capability: Refer to approved aircraft flight manual.

12. Maximum Masses:

"N" Ca	"U" Category	
Take-off	Landing	
800 kg (1764 lb)	800 kg (1764 lb)	800 kg (1764 lb)

13. Centre of Gravity Range:



Normal and Utility Category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 700 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 800 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 800 kg

14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

Normal Category:	Flaps up n	+ 3.8
	Flaps up n	1.9
	Flaps down n	+ 2
	Flaps down n	0
Utility Category:	Flaps up n	+ 4.4
	Flaps up n	2.2
	Flaps down n	+ 2
	Flaps down n	0

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum

19. Baggage / Cargo Compartment Maximum baggage compartment: 40 kg (88 lb) at 1.10m

aft of datum

TCDS No: EASA.A.367 Issue: 05 DR 400/100

20. Wheels and Tires:

Main gear track	2.58 m (8.46 ft)
	380 x 150 or 5.00-5
Front gear angular	movementleft: 27°
	right: 27°
Tire pressure	refer to the maintenance manual
Oleo strut pressure	e refer to the maintenance manual

Date: 28 July 2023

21. Control surface movements

Elevator:	.up				9°3	0' ±	: 30'
	down						
Ailerons:	.Relative	to	the	trailing	edge	of	the
wings							

	up	neutral	down	
	15°± 1°	2° ± 1°	10°± 1°	
Elevator tab:	Elevator up:	25°30' ± 1	I°6° ± 1°	

Elevator down: 10°30' ± 1°16°30' ± 1° Flaps:1st notch:15 $^{\circ} \pm 5^{\circ}$

2nd notch: $60^{\circ} - 5^{\circ}$

(1) For planes fitted with brakes controlled with rudder pedals:

16° (-0°, +2°) before operating drum brakes 20° (-0°, +3°) before operating disk brakes

22. (Reserved)

JJ.IV Operating and Service Instructions

Airplane Flight Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	. Refer to the latest amendment of Service Letter no. 6

JJ.V Note:

- 1. This plane is identical to DR 400/120 D except:
 - rear seats removed
 - luggage compartment layout
 - maximum weight
 - brakes
 - new instrument panel
- 2. "Standard 92" models: Since April 2017 (from serial nr 2703 included)

"Standard 88" models: Before April 2017 (before serial nr.2703 excluded)

Issue: 05 - DR 400 RP Date: 28 July 2023

Section KK: DR 400 RP

KK.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR400 RP

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: August 11, 1988

7. EASA Type Certification date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

KK.II Certification Basis

1. Reference Date for determining

the applicable requirements: January 1986

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 32

6. Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

8. EASA Exemptions: None9. EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 6.

KK.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition: Refer to CEAPR document n°1001131

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

4. Dimensions:

5. Engines: Porsche PFM 3200 N01

6. Propellers:



DR 200, DR300, and DR400 series

Issue: 05 - DR 400 RP Date: 28 July 2023

Manufacturer	Model	Ø	Number of blades	Governor	Minimum static RPM at sea level
Hoffmann	HO V 123 F1/200 CQ	2.00 m	3	Woodward B 2109-681	Constant speed

The EASA type certification standard includes that of FAA TC P5EU, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

TCDS No: EASA.A.367

7.1 Fuel: 91/96 or 100/130 octane minimum aviation grade gasoline

7.2 Engine Oil: Only automotive type SAE 5W 30

SAE 5W 50

SAE 10W 30

SAE 15W 50 (*)

SAE 20W 50 (*)

(*) Do not use below -5°C (25°F) external on ground temperature

8. Fluid capacities:

8.1 Fuel:

Main tank		Auxiliary tank	
(litres)		(optional) (litres)	
Capacity	Usable	Capacity	Usable
115	108	50	50

8.2 Oil: Refer to approved flight manual

9. Air speeds:

V_{NE}	
V_{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
V_A	215 km/h (116 knots IAS)
V_{FE}	

10. Maximum Operating Altitude: Refer to approved aircraft flight manual.

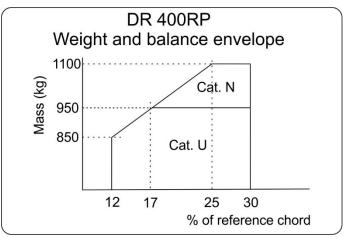
11. Operational Capability: Refer to approved aircraft flight manual.

12. Maximum Masses:

"N" Ca	"U" Category	
Take-off	Landing	
1100 kg (2425 lb)	1100 kg (2425 lb)	950 kg (2094 lb)

Issue: 05 DR 400 RP Date: 28 July 2023

13. Centre of Gravity Range:



Normal category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 850 kg Intermediate limit (25 % ref.):0.428 m aft of datum at 1100

Aft limit (30 % ref.): 0.513 m aft of datum at 1100 kg Utility category

Forward limit (12 % ref.): .. 0.205 m aft of datum at 850 kg Intermediate limit (17 % ref.):0.294 m aft of datum at 950

Aft limit (30 % ref.): 0.513 m aft of datum at 950 kg

Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

14. Datum:

Normal Category:	Flaps up n Flaps up n Flaps down n Flaps down n	1.9 + 2
Utility Category:	Flaps up nFlaps up nFlaps down nFlaps down n	2.2 + 2

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.

19. Baggage / Cargo Compartment Maximum baggage compartment: 60 kg (132 lb) at 1.90m aft of datum

Issue: 05 - DR 400 RP Date: 28 July 2023

20. Wheels and Tires

TCDS No: EASA.A.367

21. Control surface movements

up	neutral	down
15°± 1°	2° ± 1°	10°± 1°
Elevator un:	25°30' ± 1	° 6° + 1°

Elevator tab:........6° \pm 1°6° \pm 1° Elevator down: 10°30′ \pm 1°16°30′ \pm 1°

22. (Reserved)

KK.IV Operating and Service Instructions

KK.V Note:

- 1. This plane is identical to DR 400/180 R except:
 - powerplant
 - maximum weight

Issue: 05 - DR 400 NGL Date: 28 July 2023

Section LL: DR 400 NGL

LL.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 400 NGL

2. Airworthiness Category: Normal Category

Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

DGAC Type Certification date: February 19, 1991

7. EASA Type Certification date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

LL.II Certification Basis

1. Reference Date for determining

the applicable requirements: 21 March 1971

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

6. Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

8. EASA Exemptions: None9. EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 10.

LL.III Technical Characteristics and Operational Limitations

1. Type Design Definition: Refer to CEAPR document n°1002197

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

4. Dimensions:

5. Engines: Lycoming O-360-A3A

The EASA type certification standard includes that of FAA TC E-286, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS



TCDS No: EASA.A.367 Issue: 05 **DR 400 NGL**

standards certificated by individual EU member state prior

Date: 28 July 2023

to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2600 rpm

Remark: Maximum continuous power limited by noise

regulation.

Propellers:

Manufacturer	Model	Ø	Number of blades	Minimum static RPM at sea level
Sensenich	76 EM8S5-0-64	1.93 m (1)	2	2180 rpm (2)

Remarks:

- (1) No acceptable diameter reduction for repair.
- (2) Do not continuous operate between 2150 rpm and 2350 rpm.

The EASA type certification standard includes that of FAA TC P4EA, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 100/100LL octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070.

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming No. 1014.

140. 1014.				
Air temperature	Ashless dispersant (AD)	Mineral		
'	grades	grades		
All temperature	SAE15W50 or SAE20W50			
Above 80°F (+25°C)	SAE60	SAE60		
Above 60°F (+15°C)	SAE40 or SAE50	SAE50		
30°F to 90°F (O°C à +30°C)	SAE40	SAE40		
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30		
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50		
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20		

8. Fluid capacities:

8.1 Fuel:

Mair	n tank	RH tank		LH tank		Auxiliary tank	
(lit	res)	(litr	es)	(litres)		(optional) (litres)	
Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable
110	100	40	40	40	40	50	50

8.2 Oil: Oil sump capacity 8 U.S. quarts (7.57 litres)



TCDS No: EASA.A.367 Issue: 05 **DR 400 NGL**

9. Air speeds:

V_{NE}	308 km/h (166 knots IAS)
V_{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
V_A	215 km/h (116 knots IAS)
V_{FE}	170 km/h (92 knots IAS)

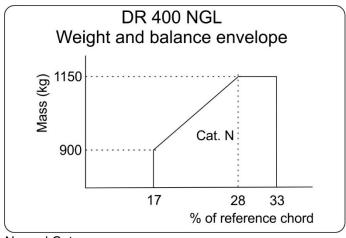
- 10. Maximum Operating Altitude: Refer to approved aircraft flight manual.
- 11. Operational Capability: Refer to approved aircraft flight manual.

12. Maximum Masses:

"N" Category				
Take-off Landing				
1150 kg (2535 lb)	1150 kg (2535 lb)			

Date: 28 July 2023

13. Centre of Gravity Range:



Normal Category

Forward limit (17 % ref.): .. 0.294 m aft of datum at 900 kg Intermediate limit (28 % ref.):0.478 m aft of datum at 1150

Aft limit (33 % ref.): 0.564 m aft of datum at 1150 kg

14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

Normal Category: Flaps up n+ 3.8 Flaps up n - 1.9 Flaps down n..... + 2 Flaps down n.....0

- 16. Levelling Means: Horizontal reference upper fuselage spar
- 17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum
- 18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.
- 19. Baggage / Cargo Compartment Maximum baggage compartment: 60 kg (132 lb) at 1.90m aft of datum

TCDS No: EASA.A.367 Issue: 05 DR 400 NGL

20. Wheels and Tires:

Main gear track		2.58 m (8.46 ft)
Front gear angular	movement	left: 27°
		right: 27°
Tire pressure	refer to the m	aintenance manual

Date: 28 July 2023

Oleo strut pressure refer to the maintenance manual

21. Control surface movements

Elevator:	up				9°3	0' ±	: 30'
	down						
Ailerons:	Relative	to	the	trailing	edge	of	the
wings				_	_		

wings			
	up	neutral	down
	15°± 1°	2° ± 1°	10°± 1°
Elevator tab:	Elevator up:	25°30' ± 1	1°6° ± 1°
	Elevator dow	n: 10°30' ± 1	1° 16°30' ± 1°
Flaps:	1st notch:		15° ± 5°
			+0°
	2nd notch:		60° -5°
Rudder:			25° ^{+3°} -0°

22. (Reserved)

LL.IV Operating and Service Instructions

Airplane Flight Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	. Refer to the latest amendment of Service Letter no. 6

LL.V Note:

- 1. This plane is identical to DR 400/180 except:
 - larger cabin
 - maximum weight
 - Sensenich 76 EM8S5-0-64 propeller only

TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

Issue: 05 - DR 400/200R Date: 28 July 2023

Section MM: DR 400/200R

MM.I General

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 400/200 R

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: December 11, 1992

7. EASA Type Certification Date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

MM.IICertification Basis

1. Reference Date for determining

the applicable requirements: 3 August 1972

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

6. Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

Airplane and towed sailplane maximum masses are limited considering the minimum climb performances required.

8. EASA Exemptions: None9. EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 10.

MM.III Technical Characteristics and Operational Limitations

1. Type Design Definition: Refer to CEAPR document n°1001131 for DR400

"STANDARD 88"

Refer to CEAPR document n°1001130 for DR400

"STANDARD 92"

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.



Issue: 05 - DR 400/200R Date: 28 July 2023

4. Dimensions:

TCDS No: EASA.A.367

5. Engines: Lycoming IO-360-A1 B6

The EASA type certification standard includes that of FAA TC 1E10, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

5.1 Engine Limits: Maximum Continuous Power: 2700 rpm

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Governor	Minimum static RPM at sea level
Hartzell	HC-C2YK-1BF/F7666A-2	1.88 m	2	Woodward B 2109-681	Constant speed (*)

Remarks: (*) Variable pitch from 14° to 29.2°

The EASA type certification standard includes that of FAA TC P-920, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

7.1 Fuel: 100/130 octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070.

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014.

Air temperature	Ashless dispersant (AD)	Mineral		
All temperature	grades	grades		
All temperature	SAE15W50 or SAE20W50			
Above 80°F (+25°C)	SAE60	SAE60		
Above 60°F (+15°C)	SAE40 or SAE50	SAE50		
30°F to 90°F (O°C à +30°C)	SAE40	SAE40		
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30		
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50		
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20		

8. Fluid capacities:

8.1 Fuel:

Main tank		Auxiliary tank		
(lit	res)	(optiona	l) (litres)	
Capacity	Usable	Capacity	Usable	
110	109	50	50	

8.2 Oil: Oil sump capacity 8 U.S. quarts (7.57 litres)

Usable...... 6 U.S. quarts (5.68 litres)



DR 400/200R Date: 28 July 2023

9. Air speeds:

Issue: 05

V_{NE}	
V_{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
V_A	215 km/h (116 knots IAS)
V_{FE}	

- 10. Maximum Operating Altitude:
- Refer to approved aircraft flight manual.
- 11. Operational Capability:
- Refer to approved aircraft flight manual.

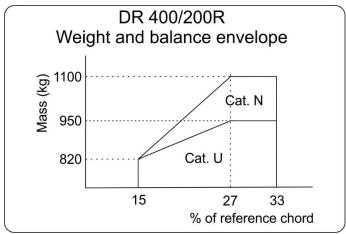
12. Maximum Masses:

"N" Ca	"U" Category	
Take-off	Landing	
1100 kg (2425 lb)	1100 kg (2425 lb)	950 kg (2094 lb)

12.1 Towing mass limitations:

Each maximum mass of the tug and of the towed glider is limited by the minimum climb performance.

13. Centre of Gravity Range:



Normal Category

Forward limit (15 % ref.): .. 0.257 m aft of datum at 820 kg Intermediate limit (27 % ref.):0.462 m aft of datum at 1100 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 1100 kg Utility Category

Forward limit (15 % ref.): .. 0.257 m aft of datum at 820 kg Intermediate limit (27 % ref.):0.462 m aft of datum at 950 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 950 kg

14. Datum: Wing leading edge of the rectangular part of the wings Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

Normal Category:	Flaps up nFlaps up n	
	Flaps down n	
	Flaps down n	
Utility Category:	Flaps up n	+ 4.4
, , ,	Flaps up n	
	Flaps down n	
	Flaps down n	

16. Levelling Means:

Horizontal reference upper fuselage spar



DR 200, DR300, and DR400 series

Issue: 05 DR 400/200R Date: 28 July 2023

17. Minimum Flight Crew:	1 (pilot) at 0.41 ± 0.05 m att of datum	
18 Maximum Passenger Seating (Capacity: 1 at 0.41+0.05m aft of datum and 2 at 1.19m aft	

18. Maximum Passenger Seating Capacity: 1 at 0.41 ± 0.05 m aft of datum and 2 at 1.19m aft of datum.

19. Baggage / Cargo Compartment Maximum baggage compartment: 60 kg (132 lb) at 1.90m

aft of datum

20. Wheels and Tires

TCDS No: EASA.A.367

Main gear track	2	.58 m (8.46 ft)
Wheel tire size	380 x	150 or 5.00-5
Front gear angular	movement	left: 27°
		right: 27°
Tire pressure	refer to the mainte	nance manual
Oleo strut pressure	e refer to the mainte	nance manual

21. Control surface movements

Elevator:	up				9°3	0' ±	: 30'
	down						
Ailerons:	Relative	to	the	trailing	edge	of	the

wings			
	up	neutral	down
	15°± 1°	2° ± 1°	10°± 1°
Elevator tab:	Elevator up:	25°30' ± 1	1°6° ± 1°
	Elevator dow	n: 10°30' ± 1	1° 16°30' ± 1°
Flaps:	1st notch:		15° ± 5°
			+0°
	2nd notch:		60° – 5°
Rudder:			25° ^{+3°} _{-0°}

22. (Reserved)

MM.IV Operating and Service Instructions

Airplane Flight Manual	Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	Refer to the latest amendment of Service Letter no. 6

MM.V Note:

- 1. This plane is identical to DR 400/180 R except:
 - Powerplant
 - maximum weight
- 2. Glider and Banner towing: Refer to approved flight manual.

lssue: 05 - DR 400/500 Date: 28 July 2023

Section NN: DR 400/500

NN.I General

TCDS No: EASA.A.367

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR 400/500

2. Airworthiness Category: Normal Category

Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. (Reserved)

6. DGAC Type Certification date: March 10, 1998

7. EASA Type Certification date: January28, 2013 (Type Certificate transfer)

8. The EASA type Certificates replaces DGAC-France Type Certificate no. 45

NN.II Certification Basis

1. Reference Date for determining

the applicable requirements: 21 March 1971

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

6. Requirements elected to comply: None

7. EASA Special Conditions: Canopy emergency release system

EASA Exemptions: None
 EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 10.

NN.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition: Refer to CEAPR document n°1002197

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.

4. Dimensions:

5. Engines: Lycoming IO-360-A1 B6

The EASA type certification standard includes that of FAA TC 1E10, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS



TCDS No: EASA.A.367 Issue: 05 DR 400/500

> standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

Date: 28 July 2023

5.1 Engine Limits: Maximum Continuous Power: 2700 rpm

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Governor	Minimum static RPM at sea level
Hartzell	HC-C2YK-1BF/F7666A-2	1.88 m	2	Woodward B 2109-681	Constant speed (*)

Remarks: (*) variable pitch from 14° to 29.2°

The EASA type certification standard includes that of FAA TC P-920, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

7. Fluids:

100/130 octane minimum aviation grade gasoline. 7.1 Fuel:

Refer to latest revision of Service Instruction Lycoming

No. 1070.

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014.

140: 1014:				
Air temperature	Ashless dispersant (AD)	Mineral		
All temperature	grades	grades		
All temperature	SAE15W50 or SAE20W50			
Above 80°F (+25°C)	SAE60	SAE60		
Above 60°F (+15°C)	SAE40 or SAE50	SAE50		
30°F to 90°F (O°C à +30°C)	SAE40	SAE40		
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30		
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50		
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20		

8. Fluid capacities:

8.1 Fuel:

Main tank		RH	tank	LH tank		Auxiliary tank		
(lit	res)	(litres)		(litres)		(optional) (litres)		
Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable	
105	104	40	40	40	40	50	50	

Oil sump capacity 8 U.S. quarts (7.57 litres) 8.2 Oil: Usable...... 6 U.S. quarts (5.68 litres)

9. Air speeds:

V_{NE}	308 km/h (166 knots IAS)
V_{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
V_A	215 km/h (116 knots IAS)
V_{FE}	

10. Maximum Operating Altitude: Refer to approved aircraft flight manual.

11. Operational Capability: Refer to approved aircraft flight manual.

12. Maximum Masses:

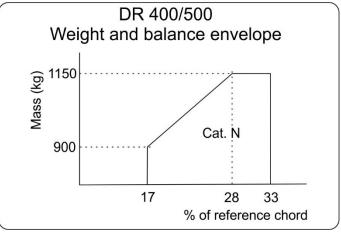
"N" Category				
Take-off	Landing			



TCDS No: EASA.A.367

lssue: 05 - DR 400/500 Date: 28 July 2023

13. Centre of Gravity Range:



Normal Category

Forward limit (17 % ref.): .. 0.294 m aft of datum at 900 kg Intermediate limit (28 % ref.):0.478 m aft of datum at 1150 kg

Aft limit (33 % ref.): 0.564 m aft of datum at 1150 kg

14. Datum: Wing leading edge of the rectangular part of the wings. Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

 Normal Category:
 Flaps up n
 + 3.8

 Flaps up n
 - 1.9

 Flaps down n
 + 2

 Flaps down n
 0

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum

18. Maximum Passenger Seating Capacity: 1 at 0.41±0.05m aft of datum and 2 at 1.19m aft of datum.

19. Baggage / Cargo Compartment Maximum baggage compartment: 60 kg (132 lb) at 1.90m aft of datum

DR 200, DR300, and DR400 series

lssue: 05 - DR 400/500 Date: 28 July 2023

20. Wheels and Tires:

TCDS No: EASA.A.367

21. Control surface movements

Elevator: up $9^\circ 30' \pm 30'$ down $12^\circ \pm 30'$ Ailerons: Relative to the trailing edge of the wings

 up
 neutral
 down

 15°± 1°
 2°± 1°
 10°± 1°

 Elevator tab:........Elevator up:
25°30′± 1°......6°± 1°

 Elevator down:
 10°30′± 1°16°30′± 1°

22. (Reserved)

NN.IV Operating and Service Instructions

Airplane Flight Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Manual	. Refer to the latest amendment of Service Letter no. 6
Airplane Maintenance Schedule	. Refer to the latest amendment of Service Letter no. 6

NN.V Note:

1. This plane is identical to DR 400 NGL except powerplant



TCDS No: EASA.A.367 DR 200, DR300, and DR400 series

lssue: 05 - DR400 / 200 I Date: 28 July 2023

Section OO: <u>DR400 / 200 I</u>

OO.I General

1. a) Type: DR 200, DR 300, and DR 400 series

b) Model: DR400 / 200 I

Airworthiness Category: Normal and Utility Category
 Type Certificate Holder: Refer to Note 2 Section PP
 Manufacturer: Refer to Note 3 Section PP

5. EASA Type Certification Application Date: 26 April 2016

6. (Reserved)

7. (Reserved)

8. EASA Type Certification Date: 25 September 2017

OO.II EASA Certification Basis

1. Reference date for determining

the applicable requirements: 3 August 1972

2. (Reserved)

3. (Reserved)

4. Certification Basis: France AIR2052

5. Airworthiness Requirements: France AIR2052 amendment June 6th, 1966

FAR part 23 as amended by amendment 7

6. Requirement elected to comply: None

7. Special Conditions: Canopy emergency release system

8. EASA Exemptions: None

9. EASA Equivalent Safety Findings: None

10. EASA Environmental Standards: ICAO Annex 16, Vol.1. Chap 10

OO.III <u>Technical Characteristics and Operational Limitations</u>

1. Type Design Definition: Refer to C.E.A.P.R. document 1001130

2. Description: Single-engine, four-seat, low-wing airplane, wood

construction, fixed tricycle landing gear.

3. Equipment: The basic required equipment as prescribed in the

applicable airworthiness regulations (see Certification Basis) must be installed in the aircraft for certification. Stall warning system "Safe Flight" n°164 or APR 79.88.00 or approved equivalent must be installed.



Issue: 05 - DR400 / 200 I

4. Dimensions

TCDS No: EASA.A.367

Span8.72 m (28.61 ft) Height2.23 m (7.32 ft) Length.......7.10 m (23.29 ft) Wing Area14.20 m² (152.85 foot²)

5. Engine: Lycoming IO-360-A1 B6

The EASA type certification standard includes that of FAA TC 1E10, based on individual EU member state acceptance or certification of this standard prior to 28 September 2003. Other standards confirming to TC/TCDS standards certificated by individual EU member state prior to 28 September 2003 are also acceptable.

2700 rpm

Date: 28 July 2023

5.1 Engine Limits: Maximum Continuous Power:

6. Propellers:

Manufacturer	Model	Ø	Number of blades	Type Certificate	Sense of rotation
MT Propeller	MTV-12B/188-59b	1.88 m	3	EASA TC P 013	Clockwise (viewed in flight direction)

7. Fluids:

7.1 Fuel: 100/130 octane minimum aviation grade gasoline.

Refer to latest revision of Service Instruction Lycoming

No. 1070.

7.2 Engine Oil: Refer to latest revision of Service Instruction Lycoming

No. 1014.

140: 1014:						
Air temperature	Ashless dispersant (AD)	Mineral				
	grades	grades				
All temperature	SAE15W50 or SAE20W50					
Above 80°F (+25°C)	SAE60	SAE60				
Above 60°F (+15°C)	SAE40 or SAE50	SAE50				
30°F to 90°F (O°C à +30°C)	SAE40	SAE40				
0°F to 70°F (-15°C à +20°C)	SAE30, SAE40 or SAE20W40	SAE30				
0°F to 90°F (-15°C à +30°C)	SAE20W50 or SAE15W50	SAE20W50				
Below 10°F (-10°C)	SAE30 or SAE20W30	SAE20				

8. Fluid capacities:

8.1 Fuel:

Main tank		RH	tank	LH tank		Auxiliary tank		
(litres)		(litr	es)	(litres)		(optional) (litres)		
Capacity	Usable	Capacity	Usable	Capacity	Usable	Capacity	Usable	
110	109	40	40	40	40	50	50	



TCDS No: EASA.A.367 Issue: 05 DR400 / 200 I

9. Air speeds:

V_{NE}	308 km/h (166 knots IAS)
V_{NO}	260 km/h (140 knots IAS)
Vc	260 km/h (140 knots IAS)
V_A	215 km/h (116 knots IAS)
V_{FE}	

Date: 28 July 2023

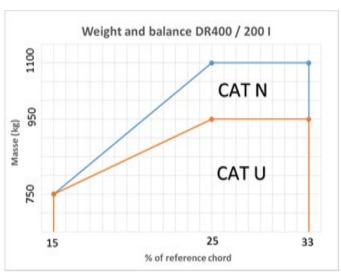
Refer to approved aircraft flight manual. 10. Maximum Operating Altitude:

11. Operational Capability: Refer to approved aircraft flight manual.

12. Maximum Masses:

"N" Ca	"U" Category	
Take-off	Landing	
1100 kg (2425 lb)	1100 kg (2425 lb)	950 kg (2094 lb)

13. Centre of Gravity Range:



Normal Category

Forward limit (15 % ref.): 0.257 m aft of datum at 750 kg Intermediate limit (25 % ref.):0.427 m aft of datum at 1100 kg Aft limit (33 % ref.): 0.564 m aft of datum at 1100 kg **Utility Category**

Forward limit (15 % ref.): 0.257 m aft of datum at 750 kg Intermediate limit (25 % ref.):. 0.427 m aft of datum at 950 kg Aft limit (33 % ref.): 0.564 m aft of datum at 950 kg

14. Datum: Wing leading edge of the rectangular part of the wings Cord length at reference section: 1.71 m (5.61 ft)

15. Load factor at maximum weight:

Normal Category:	Flaps up n	+ 3.8
	Flaps up n	1.9
	Flaps down n	
	Flaps down n	
Utility Category:	Flaps up n	+ 4.4
- , - 5 ,	Flaps up n	
	Flaps down n	
	Flaps down n	0

16. Levelling Means: Horizontal reference upper fuselage spar

17. Minimum Flight Crew: 1 (pilot) at 0.41±0.05m aft of datum DR 200, DR300, and DR400 series

Issue: 05 - DR400 / 200 I Date: 28 July 2023

22. (Reserved)

TCDS No: EASA.A.367

OO.IV Operating and Service Instructions

Airplane Flight Manual	Refer to	the la	atest a	amendm	ent of	Service	Letter i	าด. 6
Airplane Maintenance Manual	Refer to	the la	atest a	amendm	ent of	Service	Letter i	no. 6
Airplane Maintenance Schedule	Refer to	the la	atest a	amendm	ent of	Service	Letter i	าด. 6
Airplane Structural Repair Manual	Refer to	the la	atest a	amendm	ent of	Service	Letter i	no. 6
Illustrated Parts Catalogue	Refer to	the la	atest a	amendm	ent of	Service	Letter i	no. 6

OO.VNotes

1. This plane is identical to DR 400/180 except for:

Powerplant installation which is identical to DR400/200R except for the propeller MT Propeller MTV-12B/188-59b

2. First model is serial number 2695.

Date: 28 July 2023

Issue: 05 - Common Notes

Section PP: <u>Common Notes</u>

1. Type transformation:

Type transformation are only possible by the manufacturer.

DR 400 RP:

TCDS No: EASA.A.367

It is not authorized to transform a DR400/180R to a DR400 RP.

DR 400/200 R and DR 400/200 I:

It is not authorized to transform a DR 400/180R to a DR 400/200 R or a DR 400/200 I.

2. Type Certificate Holder:

C.E.A.P.R. (Centre Est Aéronautique Pierre Robin)
1 route de Troyes
21121 DAROIS
FRANCE

3. Manufacturer:

From October 1957 to August 1996

Centre Est Aéronautique Boite Postale 40 21 DIJON FRANCE

Avions P. Robin 21121 FONTAINE LES DIJON FRANCE

From September 1996 to December 2003

C.A.B (Construction Aéronautique de Bourgogne)

1 route de Troyes 21121 DAROIS FRANCE

From January 2004 to August 2008

APEX Industries 1 route de Troyes 21121 DAROIS FRANCE

2011 : DR400-140B serial number 2650 only

Finch Aircraft 1 route de Troyes 21121 DAROIS FRANCE

Since May 2011

Robin Aircraft 1b route de Troyes 21121 DAROIS FRANCE



DR 200, DR300, and DR400 series

TCDS No: EASA.A.367 Issue: 05 **Common Notes** Date: 28 July 2023

ADMINISTRATIVE SECTION

I. Acronyms & Abbreviations

II. Type Certificate Holder Record

Société Avions Pierre Robin Société Avions Robin **ROBIN** Aviation **APEX Aircraft** C.E.A.P.R.

III. Change Record

Issue 1	January 28, 2013	Initial issue on transfer of this Type Certificate to CEAPR	
Issue 2	September 25 ,2017	New model DR400 / 200 I (refer to section AB)	
Issue 3	December 2018	Merger with EASA.A.510 (DR 200 series)	
Issue 4	November 2 nd , 2020	TC holder and manufacturer transferred to Section PP Note 2 and 3	
		Type Definition Design is added for each aircraft	
		DR300 & DR400 series : wheel dimension 5.00-5 added	
		DR400 series : Standard 92 and Standard 88 models definition added	
		Section JJ : DR400/100 - Update	
		Section NN : DR400/500 N.III.15 – Load Factor correction	
		Section OO : DR400/200I - editorial change	
		Section PP : Note 1;2 & 3 - update	
Issue 5	July 28 th , 2023	DR200, DR300 & DR400 series : Oleo strut and tire pressure suppression	
		DR253 series : Tire dimension 5.00-5 added + note addition.	

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